NASA TM X- 73957-4

NASA TECHNICAL MEMORANDUM

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LaRC DESIGN ANALYSIS REPORT

FOR

NATIONAL TRANSONIC FACILITY

FOR

304 STAINLESS STEEL TUNNEL SHELL

THERMAL ANALYSIS

VOL. 4S

BY

JAMES W. RAMSEY, JR., JOHN T. TAYLOR, JOHN F. WILSON,
CARL E. GRAY, JR., ANNE D. LEATHERMAN, JAMES R. ROCKER,
AND JOHNNY W. ALLRED

National Aeronautics and Space Administration

Langley Research Center
Hampton, Virginia 23665

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7. Author(s) J. W. Ramsey, Jr., J. T. Taylor, J. F. Wilson, C. E. Gray, Jr., A. D. Leatherman, J. R. Rooker, and J. W. Allred

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Formal Documentation of Design Analyses to Obtain Code Approval of Fabricated National Transonic Facility

16. Abstract This report contains the results of extensive computer (finite element, finite difference and numerical integration), thermal, fatigue, and special analyses of critical portions of a large pressurized, cryogenic wind tunnel (National Transonic Facility). The computer models, loading and boundary conditions are described. Graphic capability was used to display model geometry, section properties, and stress results. A stress criteria is presented for evaluation of the results of the analyses. Thermal analyses were performed for major critical and typical areas. Fatigue analyses of the entire tunnel circuit is presented.

The major computer codes utilized are: SPAR - developed by Engineering Information Systems, Inc. under NASA Contracts NAS8-30536 and NAS1-13977; SALORS - developed by Langley Research Center and described in NASA TN D-7179; and SRA - developed by Structures Research Associates under NASA Contract NAS1-10091; "A General Transient Heat-Transfer Computer Program for Thermally Thick Walls" developed by Langley Research Center and described in NASA TM X-2058.

17. Key Words (Suggested by Author(s))

Pressure Vessel
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NATIONAL TRANSONIC FACILITY

TUNNEL SHELL

NASA - LaRC

THERMAL ANALYSIS

304 STAINLESS STEEL

SEPTEMBER 1976

VOLUME 4S

LaRC CALCULATIONS

FOR THE

NATIONAL TRANSONIC FACILITY

TUNNEL SHELL

DATE: SEPTEMBER, 1976

APPROVED: James W. Ramsey Jr.

DR. JAMES W. RAMSEY, JR., HEAD
STRUCTURAL ENGINEERING SECTION

ANALYSTS:

John T. Taylor

JOHN T. TAYLOR
HEAD SHELL ANALYST

John F. Wilson

JOHN F. WILSON, SHELL WORK PACKAGE & CONSTRUCTION MANAGER

Carl E. Gray, Jr.

CARL E. GRAY, JR.
SHELL ANALYST

Anne D. Leatherman

ANNE D. LEATHERMAN
SHELL PROGRAMMER

James R. Rooker

JAMES R. ROOKER
SHELL/THERMAL ANALYST

Johnny W. Allred

JOHNNY W. ALLRED
SHELL/THERMAL ANALYST

This report is one volume of a Design Analysis Report prepared by LaRC on portions of the pressure shell for the National Transonic Facility. This report is to be used in conjunction with reports prepared under NASA Contract NAS1-13535(c) by the Ralph M. Parsons Company (Job Number 5409-3 dated September 1976) and Fluidyne Engineering Corporation (Job Number 1060 dated September 1976). The volumes prepared by LaRC are listed below:

1. Finite Difference Analysis of Cone/Cylinder Junction (304 S.S.) Vol. 1, NASA TM X-73957-1.
2. Finite Element Analysis of Corners #3 and #4 (304 S.S.), Vol. 2S, NASA TM X-73957-2.
3. Finite Element Analysis of Plenum Region Including Side Access Reinforcement, Side Access Door and Angle of Attack Penetration (304 S.S.), Vol. 3S, NASA TM X73957-3.
4. Thermal Analysis (304 S.S.) Vol. 4S, NASA TM X73957-4.
5. Finite Element and Numerical Integration Analyses of the Bulkhead Region (304 S.S.), Vol. 5S, NASA TM X73957-5.
6. Fatigue Analysis (304 S.S.), Vol. 6S, NASA TM X73957-6.
7. Special Studies (304 S.S.), Vol. 7S, NASA TM X73957-7.

NTF DESIGN CRITERIA
FOR 304 STAINLESS STEEL

GENERAL

THE DESIGN OF THE PRESSURE SHELL REFLECTED IN THIS REPORT SATISFIES THE DESIGN REQUIREMENTS OF THE ASME BOILER AND PRESSURE VESSEL CODE, SECTION VIII, DIVISION 1. SINCE DIVISION 1 DOES NOT CONTAIN RULES TO COVER ALL DETAILS OF DESIGN, ADDITIONAL ANALYSES WERE PERFORMED IN AREAS HAVING COMPLEX CONFIGURATIONS SUCH AS THE CONE CYLINDER JUNCTIONS, THE GATE VALVE BULKHEADS, THE BULKHEAD-SHELL ATTACHMENTS, THE PLENUM ACCESS DOORS AND REINFORCEMENT AREAS, THE ELLIPTICAL CORNER SECTIONS, AND THE FIXED REGION (RING S8) OF THE TUNNEL. THE DIVISION 1 DESIGN CALCULATIONS, THE ADDITIONAL ANALYSES AND THE CRITERIA FOR EVALUATION OF THE RESULTS OF THE ADDITIONAL ANALYSES TO ENSURE COMPLIANCE WITH THE INTENT OF DIVISION 1 REQUIREMENTS ARE CONTAINED IN THE TEXT OF THIS REPORT. THE DESIGN ANALYSES AND ASSOCIATED CRITERIA CONSIDERED BOTH THE OPERATING AND HYDROSTATIC TEST CONDITIONS.

IN CONJUNCTION WITH THE DESIGN, A DETAILED FATIGUE ANALYSIS OF THE PRESSURE SHELL WAS ALSO PERFORMED UTILIZING THE METHODS OF THE ASME CODE, SECTION VIII, DIVISION 2.

MATERIAL

THE PRESSURE SHELL MATERIAL SHALL BE ASME, SA-240, GRADE 304 FOR PLATE AND SA-182, GRADE F304 FOR FORGINGS. THE MATERIAL PROPERTIES AT TEMPERATURES EQUAL TO OR BELOW 150°F ARE AS FOLLOWS:

(A) PLATE

YIELD = 30.0 KSI
ULTIMATE = 75.0 KSI

(B) WELDS (AUTOMATIC, SEMIAUTOMATIC, OR "STICK")

YIELD = 30.0 KSI
ULTIMATE = 75.0 KSI

OPERATING, DESIGN AND TEST CONDITIONS

THE OPERATING, DESIGN AND TEST CONDITIONS FOR THE TUNNEL PRESSURE SHELL AND ASSOCIATED SYSTEMS AND ELEMENTS ARE SUMMARIZED BELOW:

1. OPERATING MEDIUM

ANY MIXTURE OF AIR AND NITROGEN

2. DESIGN TEMPERATURE RANGE

MINUS 320 DEGREES FAHRENHEIT TO PLUS 150 DEGREES FAHRENHEIT, EXCEPT IN THE REGION OF THE PLENUM BULKHEADS AND GATE VALVES INSIDE A 23-FOOT, 4-INCH DIAMETER, FOR WHICH THE TEMPERATURE RANGE IS MINUS 320 DEGREES FAHRENHEIT TO PLUS 200 DEGREES FAHRENHEIT.

3. PRESSURE RANGE

	TUNNEL CONFIGURATION	OPERATING PRESSURE RANGE, PSIA	DESIGN PRESSURES PSID
A.	CONDITION I - PLENUM ISOLATION GATES OPEN AND TUNNEL OPERATING:		
	TUNNEL CIRCUIT EXCEPT PLENUM	8.3 to 130	A. 8 EXTERNAL B. 119 INTERNAL
	PLENUM (PLENUM PRESSURE IS LIMITED TO .4 TO 1 TIMES THE REMAINDER OF THE TUNNEL CIRCUIT	3.3 to 130	A. 15 EXTERNAL B. 119 INTERNAL
	BULKHEAD		56 (EXTERNAL TO PLENUM)
B.	CONDITION II - PLENUM ISOLATION GATES OPEN AND TUNNEL SHUTDOWN:		
	ENTIRE TUNNEL CIRCUIT	8.3 to 130	A. 8 EXTERNAL B. 119 INTERNAL
	BULKHEAD		0
C.	CONDITION III - PLENUM ISOLATION GATES AND ACCESS DOORS CLOSED:		
	TUNNEL CIRCUIT EXCEPT PLENUM	8.3 to 130	A. 8 EXTERNAL B. 119 INTERNAL

PLENUM (PLENUM OPERATING PRESSURE CAN EXCEED THE PRESSURE IN THE REMAINDER OF THE TUNNEL CIRCUIT BY 24 PSI, BUT DOES NOT EXCEED THE 130 PSIA MAXIMUM OPERATING PRESSURE)	0 to 130	A. 15 EXTERNAL B. 119 INTERNAL
BULKHEAD		A. 25 (INTERNAL TO PLENUM) B. 119 (EXTERNAL TO PLENUM) FOR MINUS 320 DEGREES FAHRENHEIT TO PLUS 150 DEGREES FAHRENHEIT *C. 115.7 (EXTERNAL TO PLENUM) FOR PLUS 151 DEGREES FAHRENHEIT TO PLUS 200 DEGREES FAHRENHEIT

*OPERATING PROCEDURES LIMIT PRESSURES TO THAT SHOWN.

D. CONDITION IV - PLENUM ISOLATION GATES CLOSED AND ACCESS DOORS OPEN:

TUNNEL CIRCUIT EXCEPT PLENUM	8.3 to 130	A. 8 EXTERNAL B. 119 INTERNAL
PLENUM	14.7	0
BULKHEAD		A. 119 (EXTERNAL TO PLENUM) FOR MINUS 320 DEGREES FAHRENHEIT TO PLUS 150 DEGREES FAHRENHEIT *B. 115.7 (EXTERNAL TO PLENUM) FOR PLUS 151 DEGREES FAHRENHEIT TO PLUS 200 DEGREES FAHRENHEIT

*OPERATING PROCEDURES LIMIT PRESSURES TO THAT SHOWN.

4. HYDROSTATIC TEST DESIGN CONDITIONS

THE PRESSURE SHELL WAS DESIGNED FOR HYDROSTATIC TEST IN ACCORDANCE WITH THE REQUIREMENTS OF THE ASME CODE, SECTION VIII, DIVISION 1. THE TEST PRESSURES SHALL BE AS FOLLOWS. PRESSURE SHELL TEMPERATURE SHALL BE EQUAL TO OR BELOW 100°F DURING HYDROSTATIC TESTS.

CONDITION (1) - MAXIMUM INTERNAL PRESSURE CONDITION FOR THE ENTIRE TUNNEL CIRCUIT

$$PH_1 = 1.5\ (119)\ \left(\frac{18.7}{18.2}\right) + \text{HYDROSTATIC HEAD}$$

$$= 183.4\ \text{PSI} + \text{HYDROSTATIC HEAD}$$

CONDITION (2) - MAXIMUM DIFFERENTIAL PRESSURE CONDITION ACROSS THE PLENUM BULKHEADS

$$PH_2 = 1.5\ \left(\frac{18.7}{18.2}\right)\ (119) + \text{HYDROSTATIC HEAD}$$

$$= 183.4 + \text{HYDROSTATIC HEAD}$$

$$PH_2{}^* = 1.5\ (115.7)\ \left(\frac{18.7}{17.7}\right) + \text{HYDROSTATIC HEAD}$$

$$= 183.4 + \text{HYDROSTATIC HEAD}$$

*TUNNEL OPERATION LIMITATIONS PRECLUDE PRESSURE DIFFERENTIALS ACROSS BULKHEADS IN EXCESS OF 115.7 PSI FOR BULKHEAD AND GATE TEMPERATURES IN EXCESS OF 150°F.

CONDITION (3) - MAXIMUM REVERSE DIFFERENTIAL PRESSURE CONDITION ACROSS THE PLENUM BULKHEADS

$$PH_3 = 1.5\ \left(\frac{18.7}{18.2}\right)\ (25) = 38.5\ \text{PSI}$$

THE PRESSURE SHELL EXCEPT FOR THE PLENUM SHALL BE PRESSURIZED TO 144.9 PSIG. THE PLENUM SHALL BE PRESSURIZED TO 183.4 PSIG.

PRESSURE SHELL STRESS EVALUATION CRITERIA

THIS CRITERIA ESTABLISHES THE BASIS FOR ANALYSIS AND DESIGN OF THE PRESSURE SHELL SO IT WILL MEET OR EXCEED ALL OF THE REQUIREMENTS OF SECTION VIII, DIVISION 1 OF THE ASME BOILER AND PRESSURE VESSEL CODE AND CAN BE STAMPED WITH A DIVISION 1 "U" STAMP.

1. SECTION VIII, DIVISION 1, DIRECT APPLICATION

(A) THE MAXIMUM ALLOWABLE STRESS (S)

$S = 18.2$ KSI (-320°F TO +150°F)

$S = 17.7$ KSI (-320°F TO +200°F)

(B) PRIMARY BENDING PLUS PRIMARY MEMBRANE STRESSES

THE LOCAL MEMBRANE STRESSES ARE NOT GENERALLY CONSIDERED IN SECTION VIII, DIVISION 1 DESIGNS. HOWEVER, FOR THE PURPOSE OF DESIGNING LOCAL REINFORCEMENT AT BRACKETS, RINGS OR PENETRATIONS NOT COVERED BY DESIGN BASED ON STRESS ANALYSIS, THE LOCAL SHELL MEMBRANE STRESS SHALL BE:

$$P_b + P_m \leq 1.5\ SE$$

NOTE: E IS JOINT EFFICIENCY

2. IN REGIONS OF THE PRESSURE SHELL WHERE DIVISION 1 DOES NOT CONTAIN RULES TO COVER ALL DETAILS OF DESIGN (REF. U-2(g)), ADDITIONAL ANALYSES WERE PERFORMED UTILIZING THE GUIDELINES OF THE ASME CODE, SECTION VIII, DIVISION 2, APPENDIX 4, "DESIGN BASED ON STRESS ANALYSIS." THE BASIC STRESS CRITERIA FOR DIVISION 2 IS REPRESENTED IN FIGURE 4-130.1 AND RESTATED BELOW INDICATING ANY MODIFICATIONS OR EXCESS REQUIREMENTS APPLIED TO IT TO REMAIN WITHIN THE INTENT OF DIVISION 1 AND TO OBTAIN A DIVISION 1 STAMP.

A. GENERAL PRINCIPAL MEMBRANE STRESS

MAXIMUM ALLOWABLE STRESS

$S = 18.2$ KSI (-320°F TO +150°F)

$S = 17.7$ KSI (-320°F TO +200°F)

MAXIMUM ALLOWABLE STRESS INTENSITY

$S_m = 20.0$ KSI (-320°F TO +300°F)

B. PRIMARY GENERAL MEMBRANE STRESS INTENSITY

$$P_m \leq S_m$$

AND IN ORDER TO COMPLY WITH DIVISION 1, THE MAXIMUM PRINCIPAL MEMBRANE STRESS MUST BE:

$$P_m^* \leq S$$

NOTE: THE * IS USED TO DENOTE THAT MAXIMUM PRINCIPAL STRESSES ARE TO BE COMPUTED FOR THE GIVEN LOADING CONDITION. THE INTENT IS TO DETERMINE THE STRESSES WHICH REPRESENT THE HOOP STRESSES AND MERIDIONAL STRESSES WHICH ARE THE STRESSES USED IN DIVISION 1 COMPUTATIONS.

C. DESIGN LOADS, PRIMARY LOCAL MEMBRANE STRESS INTENSITY

$$P_L \leq 1.5\ S_m$$

NOTE: LOCAL MEMBRANE STRESS INTENSITY IS DEFINED IN ACCORDANCE WITH DIVISION 2, APPENDIX 4-112(i). THE TOTAL MERIDIONAL LENGTH IS CONSIDERED TO BE 1.0 $\sqrt{RT}$.

D. DESIGN LOADS, PRIMARY LOCAL MEMBRANE PLUS PRIMARY BENDING STRESS INTENSITY

$$P_L + P_b \leq 1.5\ S_m$$

E. OPERATING LOADS, PRIMARY PLUS SECONDARY STRESS INTENSITY

$$P_L + P_b + Q \leq 3\ S_m$$

3. A FATIGUE ANALYSIS WAS CONDUCTED IN ACCORDANCE WITH SECTION VIII, DIVISION 2 WITHOUT MODIFICATION.

4. HYDROSTATIC TEST CONDITION DESIGN CONSIDERATIONS

A. PRESSURE SHELL

IN ACCORDANCE WITH DIVISION 1 OF THE ASME CODE, DESIGN ANALYSIS OF THE PRESSURE SHELL FOR THE HYDROSTATIC TEST CONDITION IS NOT REQUIRED. HOWEVER, IN ORDER TO PROVIDE A SATISFACTORY ENGINEERING DESIGN FOR THE PRESSURE SHELL SPECIAL EMPHASIS WAS GIVEN, AS PROMPTED BY NOTE (1) OF SECTION VIII, DIVISION 1 OF THE ASME CODE, TO FLANGES OF GASKETED JOINTS OR OTHER APPLICATIONS WHERE SLIGHT AMOUNTS OF DISTORTION CAN CAUSE LEAKAGE OR MALFUNCTION. EXAMPLES OF THESE AREAS ARE THE PLENUM, PLENUM ACCESS DOORS, PLENUM ACCESS DOOR REINFORCEMENT, THE BULKHEADS, AND BULKHEAD FLANGES.

B. SUPPORT RINGS

DESIGN OF THE PRESSURE SHELL SUPPORT RINGS, INCLUDING

THE CORNER RINGS, FOR THE HYDROSTATIC TEST CONDITION, COMPLIES WITH THE FOLLOWING:

(A) THE COMBINED VALUE OF THE SHELL CIRCUMFERENTIAL PRESSURE STRESS, S_1 AND SHELL BENDING STRESS S_2, RESULTING FROM ACTION OF A PORTION OF THE SHELL AS AN INNER FLANGE OF THE RING, SHALL NOT EXCEED 0.8 WELD YIELD STRESS:

$$S_1 + S_2 \leq 0.8 \text{ WELD YIELD STRESS,}$$

WHERE, FOR SUPPORT RINGS NOT ANALYZED BY FINITE ELEMENT TECHNIQUES,

$S_1 = P_H (\frac{R}{T}) + .6 P_H$; P_H INCLUDES HYDROSTATIC HEAD CORRECTION, AND

S_2 = RING BENDING STRESS AT INNER FLANGE, BASED ON AN EFFECTIVE WIDTH OF THE PRESSURE SHELL ACTING AS AN INNER FLANGE OF THE RING OF 1.1 MULTIPLIED BY THE SQUARE ROOT OF $D_o T$.

(B) THE BENDING STRESS, S_{2F} ON THE OUTSIDE FLANGE SHALL NOT EXCEED .9 WELD YIELD STRESS. (IN THE COMPUTER ANALYSIS ALL LOADING CONDITIONS ARE LIMITED TO .9 S_Y ON THE OUTER FLANGE.)

(C) BRACKETS AND SUPPORT PAD WELDMENTS

THE DESIGN FOR ALL LOADING CONDITIONS INCLUDING THE HYDROSTATIC TEST CONDITION OF THOSE PORTIONS OF BRACKETS AND SUPPORT PAD WELDMENTS WHICH ARE ATTACHED TO THE PRESSURE SHELL BUT NOT ON THE SURFACE OF THE SHELL SHALL COMPLY WITH THE REQUIREMENTS OF THE AISC CODE, I.E. MAXIMUM STRESS IN TENSION EQUALS .6 S_Y, ETC.

The enclosed analyses is for 9% Ni with a 6" Temp-Mat Insulation with internal circumferential "T" rings. The new baseline insulation is a closed cell material "Rohacell", with internal tabs. The "Rohacell" insulation reduces the stresses contained herein by a factor of 7.

THERMAL ANALYSIS REPORT

I STEADY STATE ANALYSIS

OF BULKHEAD REGION.

BY ______ DATE ______ SUBJECT ______ SHEET NO. 2 OF ______
CHKD. BY ______ DATE ______ JOB NO. ______

COMPUTER PROGRAMS

1- TEMPERATURES WERE CALCULATED WITH "A GENERAL TRANSIENT HEAT-TRANSFER COMPUTER PROGRAM FOR THERMALLY THICK WALLS". NASA TECHNICAL MEMORANDUM NO. [TM X-2058]

16. Abstract

This program is a general heat-transfer program which employs a finite-difference method for the solution of temperature histories of one-dimensional, two-dimensional, or spherical systems. Options are available for heat input given in tabular form, computed from a trajectory, or computed from a temperature history given for a specific location. The types of heat exchange are: (1) conduction; (2) convection – with (a) given heat input, (b) heating due to skin friction with Van Driest equations, (c) stagnation heating with Sibulkin, Detra-Kemp-Riddell, and Cohen equations; (3) radiation-out; (4) air-conduction; and (5) joint conduction. The system configuration is specified by an arbitrary number of discrete elements and their interrelationships.

2- STRESSES WERE CALCULATED WITH "SPAR" WHICH IS A SYSTEM OF COMPUTER PROGRAMS USED PRIMARILY TO PERFORM STRESS, BUCKLING, AND VIBRATIONAL ANALYSES OF LINEAR FINITE ELEMENT SYSTEMS.

MANUAL NO. EISI/A2200 BY

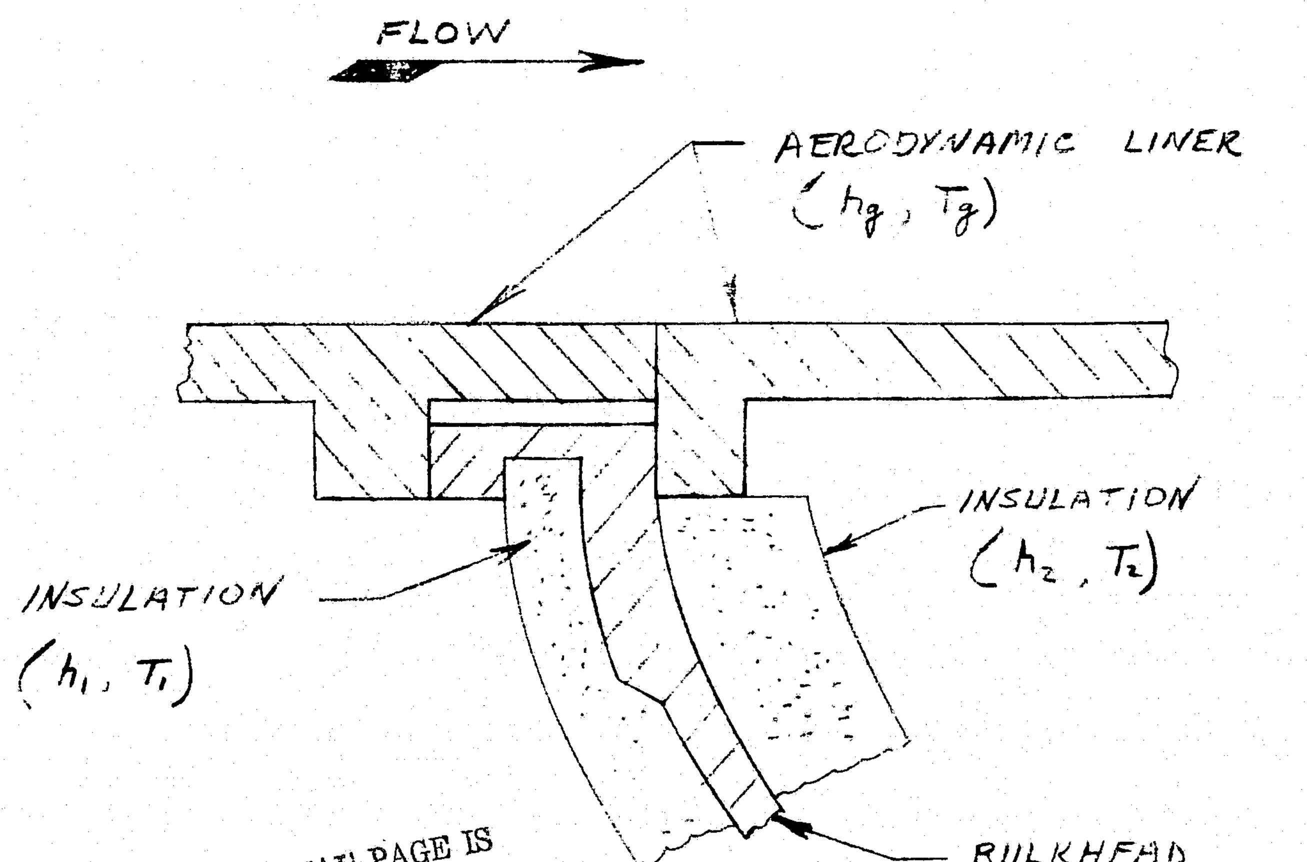
ENGINEERING INFORMATION SYSTEM, INC.
5120 CAMPBELL AVENUE, SUITE 240
SAN JOSE, CALIFORNIA 95130
(408) 379-0730

I- STEADY STATE ANALYSIS OF BULKHEAD

THE STEADY STATE THERMAL ANALYSIS OF THE BULKHEAD (DRAWING NO. HAS BEEN CONDUCTED FOR GATE VALVES OPENED AND CLOSED

A- GATE VALVE OPENED WITH FLOW:

THIS STEADY STATE CASE EXISTS WHEN THE TUNNEL IS IN OPERATION WITH THE AERO-DYNAMIC LINERS CONNECTED TO THE BUCKHEAD AS SHOWN BELOW



WHERE :

h = HEAT TRANSFER COEFFICIENT IN REGIONS SHOWN

T = TEMPERATURE OF GAS

ASSUMPTIONS:

1- ASSUME LINER TEMPERATURE TO EQUAL TO GAS STREAM TEMPERATURE SINCE FLOW IS NEAR MACH 1 AT LINER AND HEAT TRANSFER COEFFICIENT WILL BE LARGE.

2- ASSUME h_1 & h_2 ARE LARGE. THE RESISTANCE OF HEAT FLOW THRU SURFACE FILM WILL BE SMALL COMPARED TO RESISTANCE OF HEAT THRU INSULATION. THEREFORE OUTER SURFACE OF INSULATION WILL BE SAME AS GAS TEMPERATURE.

BOUNDARY CONDITIONS

BASED ON ABOVE ASSUMPTIONS, THE BOUNDARY CONDITIONS ARE SAME AS A/E BOUNDARY CONDITIONS AND SHOWN IN TABLE 1

HEAT TRANSFER COEFFICIENT WILL EXIST ONLY IN BLOCKS 1 THRU 6. AN EFFECTIVE COEFFICENT IS CALCULATED FOR THE OTHER ELEMENT.

EFFECTIVE THERMAL BOUNDARY CONDITION IS DETERMINED BY DIVIDING THE THERMAL CONDUCTIVITY BY THE INSULATION THICKNESS.

FOR EXAMPLE:

$$k = 1.47 \ \frac{\text{Btu-IN}}{\text{FT}^2\text{-HR-}^\circ\text{F}}$$

$$t = 6 \text{ INCHES}$$

$$\therefore h_e = \frac{1.47 \ \frac{\text{Btu-IN}}{\text{FT}^2\text{-HR-}^\circ\text{F}}}{6 \text{ IN}} = .245 \ \frac{\text{Btu}}{\text{FT}^2\text{-HR-}^\circ\text{F}}$$

$$h_e = 4.726 \times 10^{-7} \ \frac{\text{Btu}}{\text{IN}^2\text{-Sec-}^\circ\text{F}} \quad \checkmark$$

GEOMETRY

THE DIMENSIONS OF THE FINITE ELEMENT MODEL IS SHOWN IN FIGURE 1

DETERMINATION OF HEAT TRANSFER COEFFICIENT AND GAS TEMPERATURE FOR COMPUTER PROGRAM.

THE COMPUTER PROGRAM WILL ALLOW ONLY ONE GAS HEAT TRANSFER COEFFICIENT AND ONE GAS TEMPERATURE FOR EACH ELEMENT. THEREFORE, THESE VALUES ARE DEFINED AS FOLLOWS:

$$h_{eff} = \frac{h_1 A_1 + h_2 A_2}{A_1 + A_2}$$

$$T_{eff} = \frac{h_1 A_1 T_1 + h_2 A_2 T_2}{h_1 A_1 + h_2 A_2}$$

WHERE,

$h_1 A_1 T_1$ ARE CONDITIONS ON ONE SIDE OF ELEMENTS

AND

$h_2 A_2 T_2$ ARE CONDITIONS ON OTHER SIDE OF ELEMENTS

THESE VALUES ARE LISTED IN TABLE 1

BY ______ DATE ______ SUBJECT ______ SHEET NO. 1 OF ______
CHKD. BY ______ DATE ______ JOB NO. ______

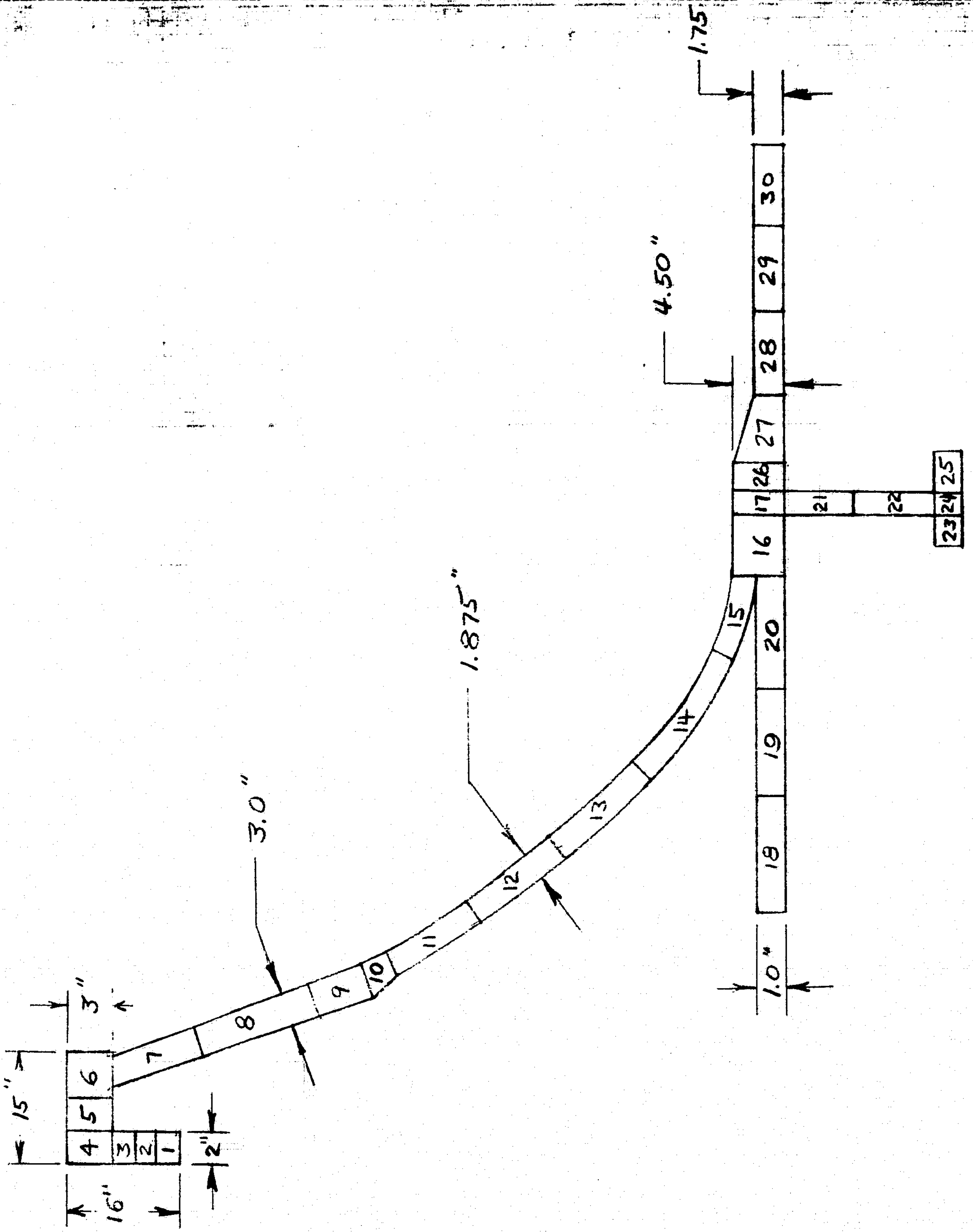


FIGURE - 1

DIMESIONS

ELEMENT NO.	LENGTH	WIDTH
1	6.5"	4.0"
2	4.0	4.0"
3	2.5	4.0"
4	5.0	4.0"
5	5.0	5.0
6	5.0	5.0
7	11.657	3.0
8	20.001	3.0
9	5.309	3.0
10	2.721	2.438
11	16.985	1.875
12	11.284	1.875
13	13.019	1.875
14	14.228	1.875
15	4.708	1.875
16	14.330	4.50
17	1.240	4.50
18	24.00	1.0
19	18.0	1.0
20	18.0	1.0
21	1.24	7.25
22	1.24	12.08
23	5.38	1.24
24	1.24	1.24
25	5.38	1.24
26	2.88	4.50
27	8.50	3.125
28	12.00	1.75
29	21.00	1.75
30	27.00	1.75

TABLE 1

(FLOW BOUNDARY CONDITIONS)

ELEMENT NO.	HEAT TRANSFER COEFFICIENT (Btu/IN²-SEC-°F)	GAS TEMPERATURE (°R)
1	1.066×10^{-5}	160
2	7.566×10^{-6}	
3	7.566×10^{-6}	
4	1.10×10^{-5}	
5	5.401×10^{-6}	
6	8.524×10^{-6}	
7	4.726×10^{-7}	
8		
9		
10		
11		
12		
13		
14		
15	4.726×10^{-7}	160
16	1.711×10^{-6}	506
17	4.726×10^{-7}	160
18	1.698×10^{-6}	505
19	1.698×10^{-6}	505
20	1.698×10^{-6}	505
21	2.894×10^{-6}	560
22		
23		
24		
25	2.894×10^{-6}	560
26	1.711×10^{-6}	506
27	1.706×10^{-6}	506
28	1.7×10^{-6}	505
29	1.7×10^{-6}	505
30	1.7×10^{-6}	505

RESULTS

THE TEMPERATURE DISTRIBUTION WAS CALCULATED FOR THE MODEL SHOWN IN FIGURE 1. THE UPDATED MODEL, SHOWN IN FIGURE 2, SHOWS THE FINAL DIMENSIONS OF THE BULKHEAD. A COMPARSION WILL BE SHOWN IN THE TRANSIENT ANALYSIS THAT THIS CHANGE IN DIMESIONS DOES NOT EFFECT THE TEMPERATURES OF THE BULKHEAD SINCE THE HEAT TRANSFER COEFFICIENT IS LARGE "ENOUGH" TO GIVE UNIFORM TEMPERATURE IN THE FLANGE AREA.

THE TEMPERATURE DISTRIBUTION OF THE BUCKHEAD IS SHOWN IN FIGURE 3. THIS AGREES WITHIN 3° OF FLUIDYNE'S CALCULATED RESULTS SHOWN IN FIGURE 4.

THE STRESSES FOR THIS CASE WILL NOT BE CALCULATED SINCE THE TEMPERATURE GRADIENTS ARE NOT AS SEVERE AS IN TRANSIENT CASE SHOWN ON FIGURE 11. THE STRESSES ARE SHOWN ON FIGURES 12, 13, AND 14.

THE UPDATED CONFIGURATION OF THE TUNING-FOLK IS SHOWN IN FIGURE 5. THE TEMPERATURE WILL BE SIMILAR TO THAT SHOWN IN FIGURE 5 SINCE THE TEMPERATURE GRADIENTS IN THIS AREA ARE SMALL COMPARED TO THE INNER FLANGE. THE STRESSES IN THIS AREA ARE ALSO SMALL AS SHOWN IN FIGURES 12, 13, AND 14.

BY ________ DATE ________ SUBJECT ________ SHEET NO. 11 OF ________
CHKD. BY ________ DATE ________ JOB NO. ________

UPDATE OF THERMAL MODEL OF BULKHEAD

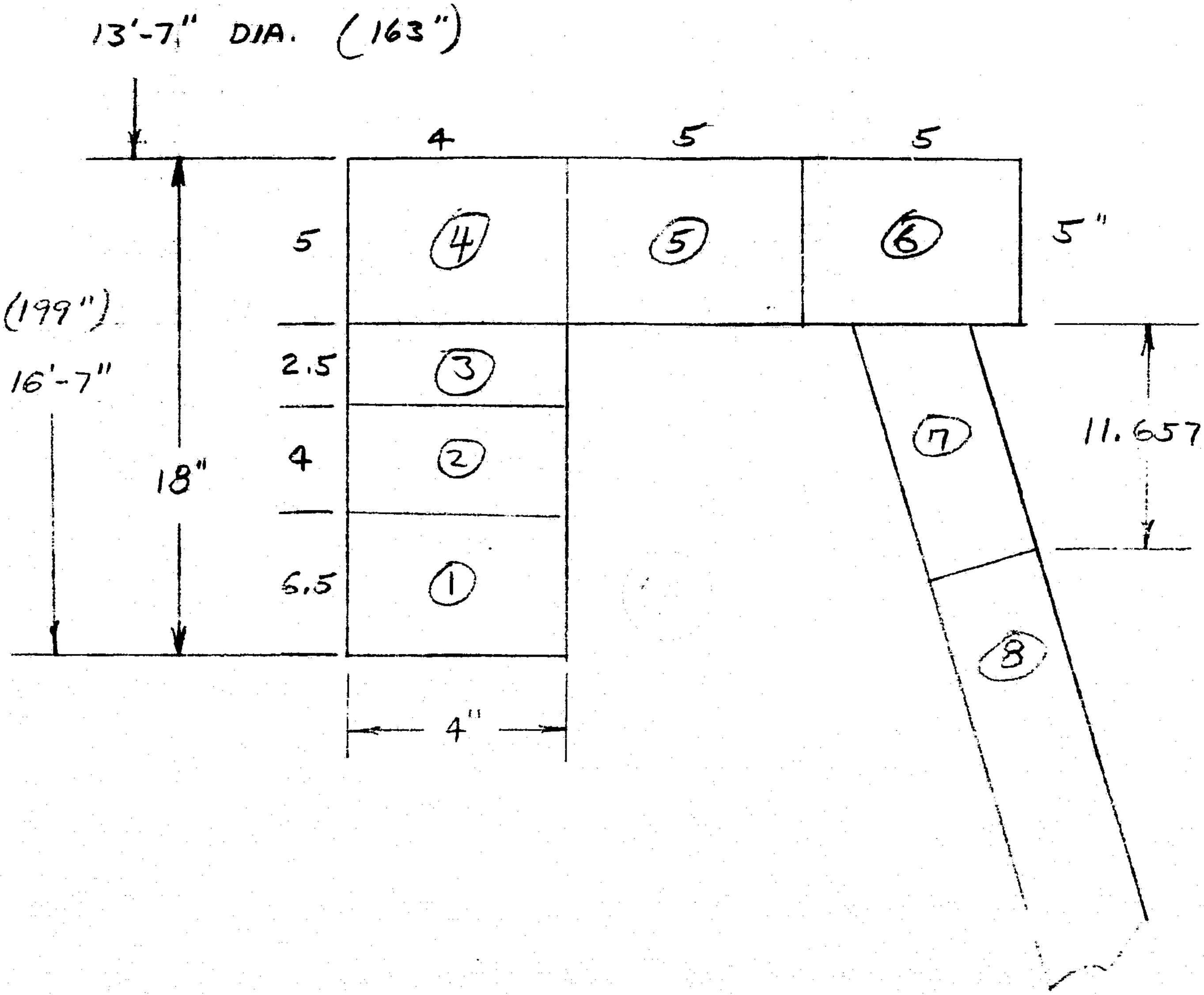


FIGURE 2

BY________ DATE________ SUBJECT________________ SHEET NO.____ OF____

CHKD. BY______ DATE________ JOB NO.____________

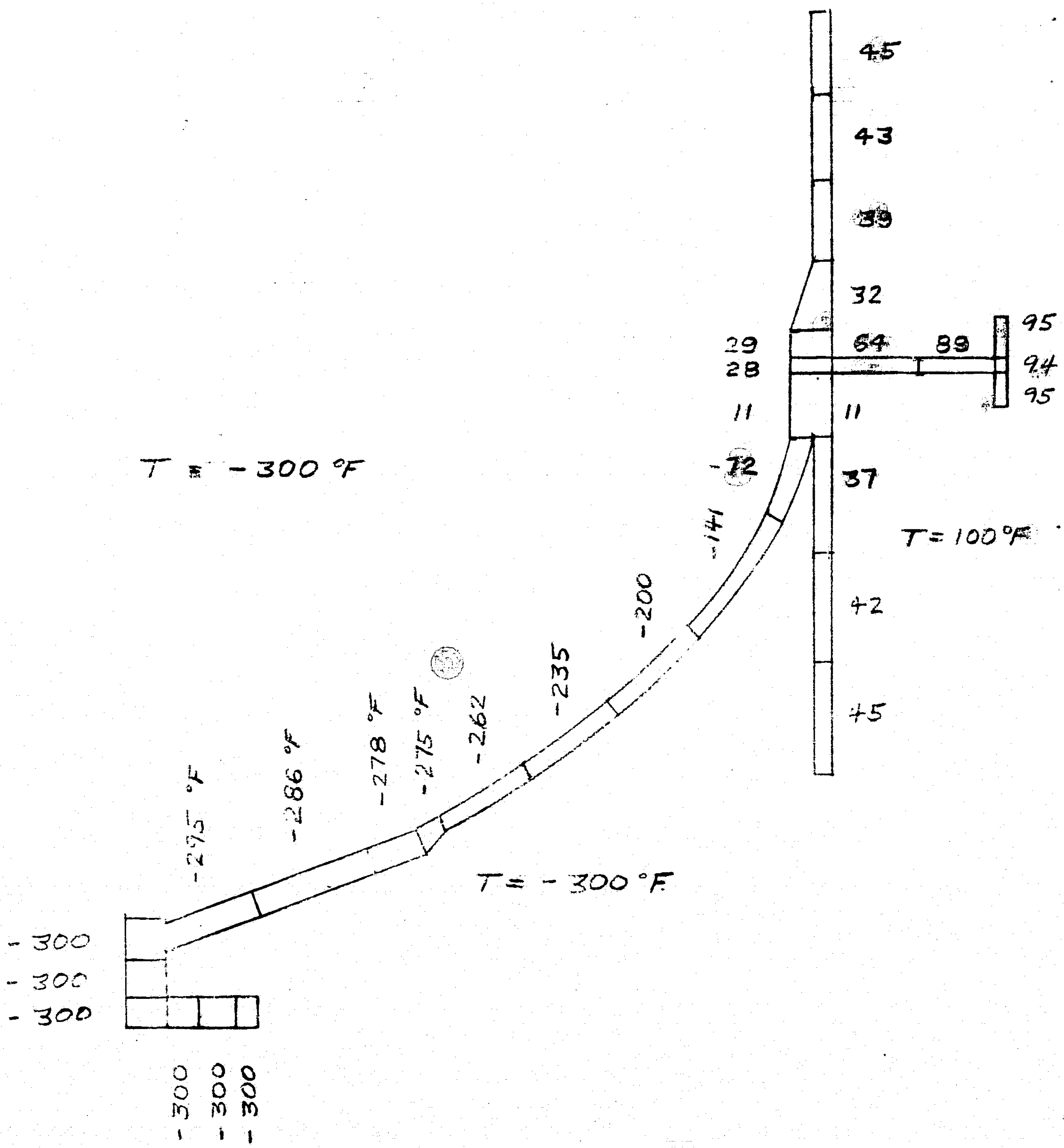


FIGURE 3

BY ______ DATE ______ SUBJECT ______ SHEET NO. ______ OF ______
CHKD. BY ______ DATE ______ JOB NO. ______

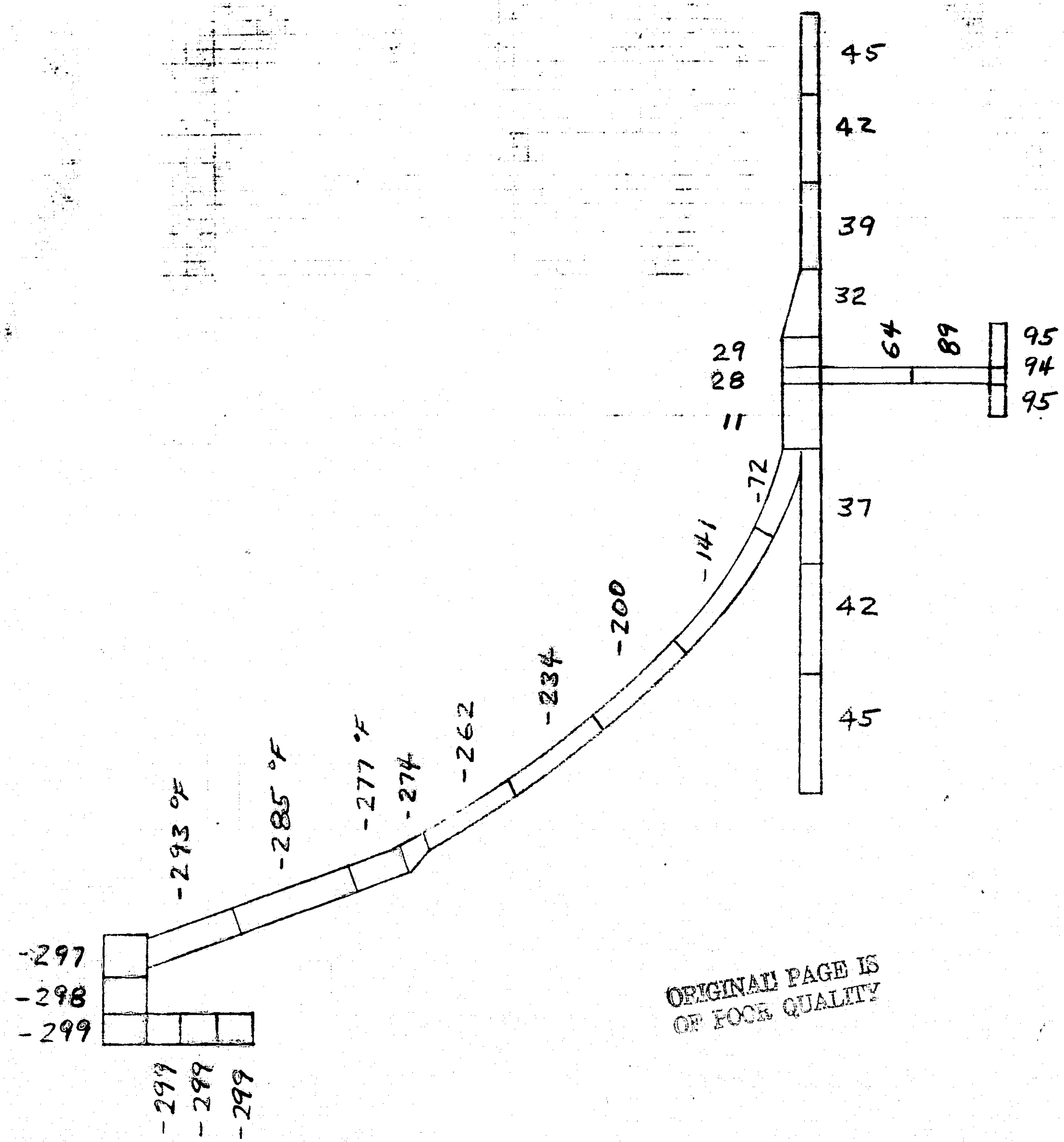


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FIGURE 4

(FLUIDYNE RESULTS)

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CHKD. BY ________ DATE ________ JOB NO. ________________

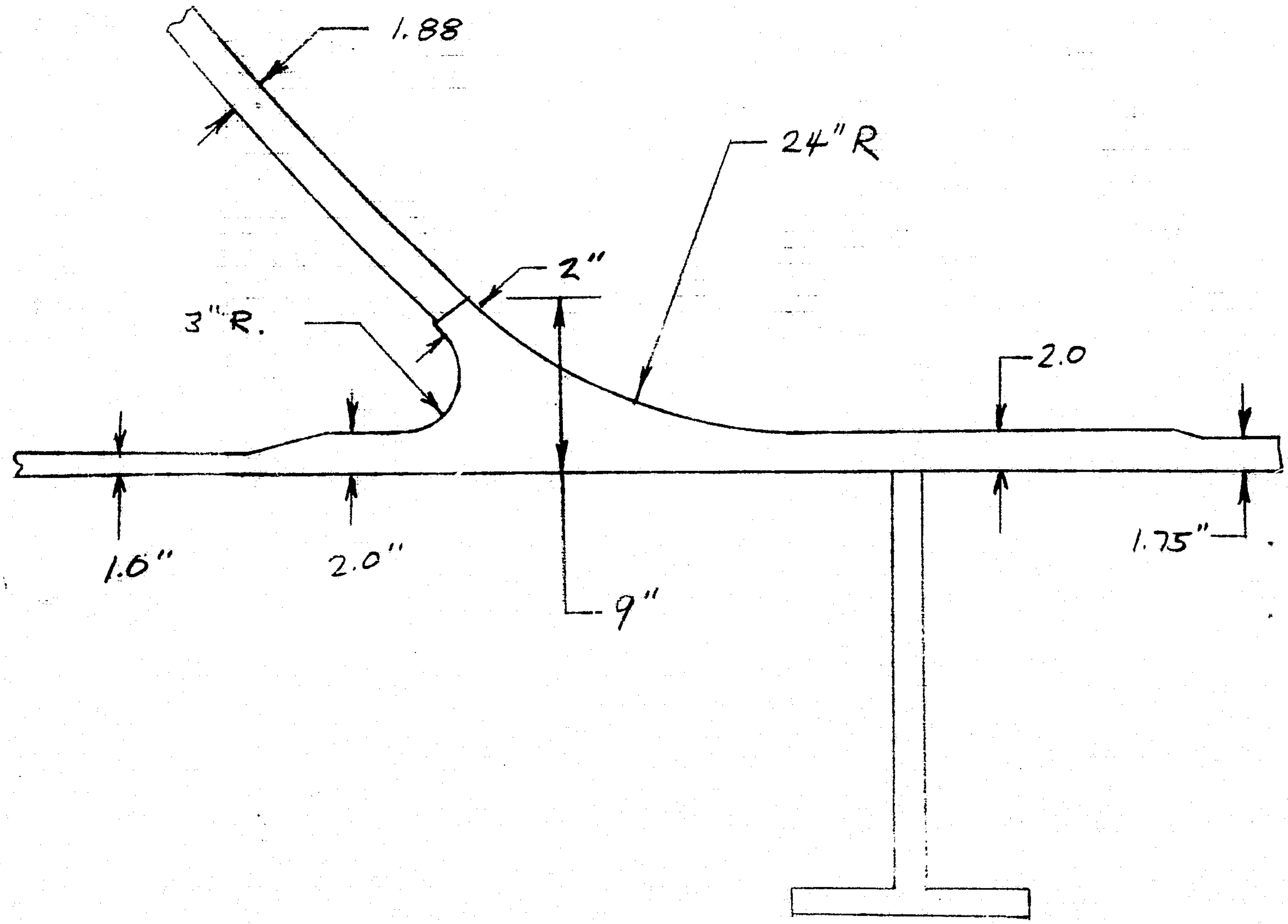
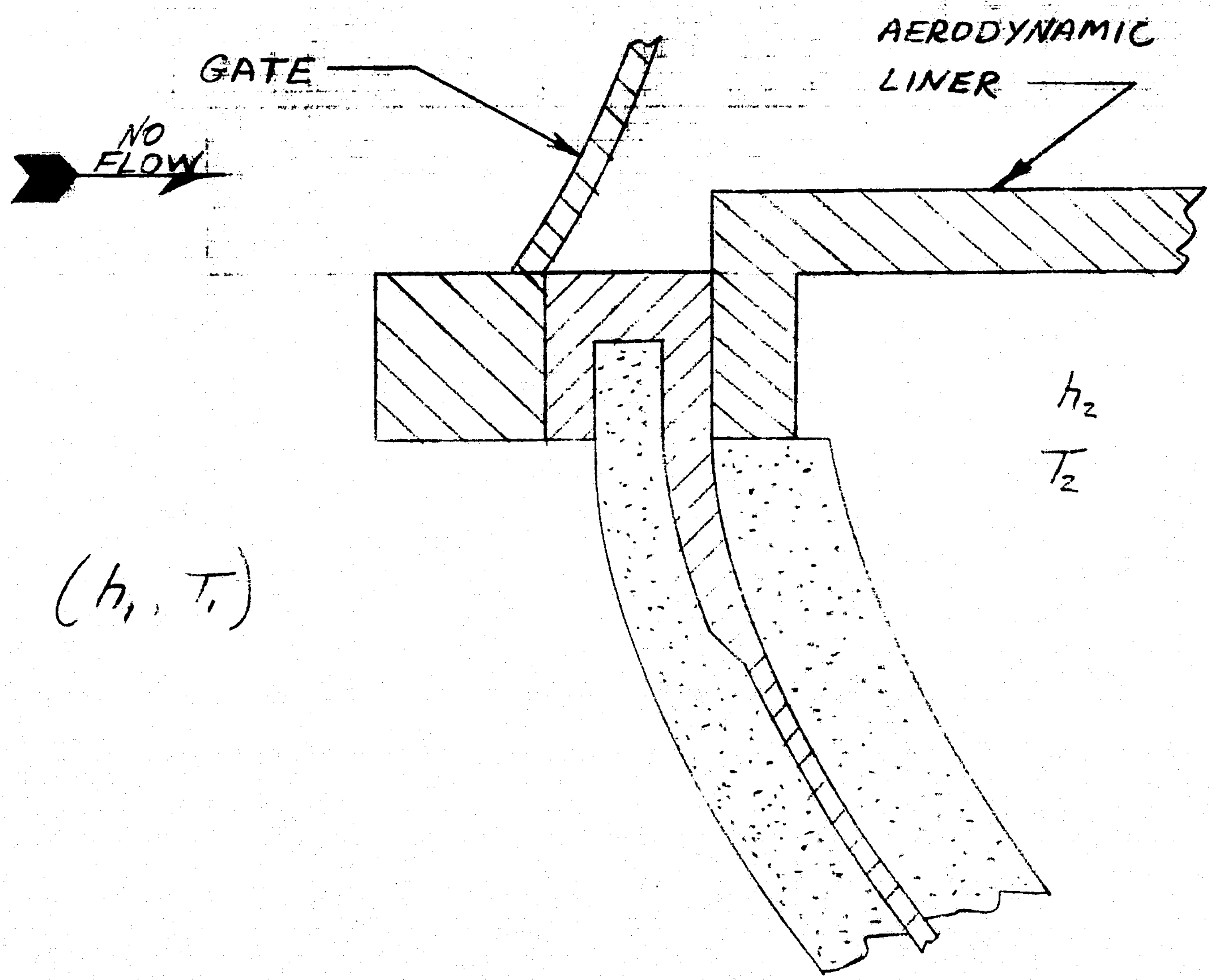


FIGURE 5

(FINAL DIMENSIONS OF TUNING FOLK)

B- GATE VALVE CLOSED - NO FLOW

THIS STEADY STATE CASE EXISTS WHEN THE GATE VALVE IS CLOSED WITH THE F LLOWING BOUNDARY CONDITIONS:



ASSUMPTIONS:

1- ASSUME h_1 & h_2 ARE LARGE, THEREFORE THE SURFACES EXPOSED TO THE GAS ARE ASSUMED TO BE THE SAME AS THE GAS TEMPERATURE.

2- ASSUME TEMPERATURE OF GATE IS -100°F (THIS ASSUMPTION IS CHECKED IN TRANSIENT ANALYSIS) SEE RESULTS FOR CHECK ON THIS ASSUMPTION.

BOUNDARY CONDITIONS:

THE STEADY STATE BOUNDARY CONDITIONS ARE AS FOLLOWS:

$$\begin{cases} T_1 = -300\,°F \\ T_2 = 100\,°F \end{cases}$$

HEAT TRANSFER COEFFICIENTS FOR LINER IN CONTACT WITH GATE AND AERODYNAMIC LINER ARE LISTED IN TABLE 2.

RESULTS:

THE TEMPERATURE DISTRIBUTION IS SHOWN IN FIGURE 6. THIS GRADIENT IS LESS THAN THE FLOW DISTRIBUTION SHOWN IN FIGURE 3. THE TEMPERATURE GRADIENT THRU THE WALL THICKNESS IN NEGLIGBLE. THEREFORE THE THICKNESS THERMAL STRESS WILL BE SMALL. THE LOCAL GRADIENT AT THE GATE VALVE IS LARGER THAN THE FLOW CONDITION BUT

IS LESS THAN GRADIENTS SHOWN LATER FOR THE TRANSIENT HEATING OF THE PLEMUM.

THE ASSUMED GATE TEMPERATURE OF -100°F WAS INCORRECT. THE FINAL GATE TEMPERATURE CALCULATED FROM THE THERMAL ANALYSIS IS -260°F. THE TRANSIENT ANALYSIS WILL GIVE A MORE SEVERE TEMPERATURE AS SHOWN IN NEXT SECTION.

BY________ DATE________ SUBJECT________________ SHEET NO.____ OF____
CHKD. BY________ DATE________ JOB NO.________

TABLE 2

(NONFLOW THERMAL BOUNDARY CONDITIONS)

ELEMENT NO.	HEAT TRANSFER COEFFICIENT ($Btu/IN^2\ sec\text{-}^\circ F$)	GAS TEMPERATURE ($^\circ R$)
1	1.0×10^{-3}	360
2		360
3		360
4		439
5		560
6	1.0×10^{-3}	560
7	4.723×10^{-7}	360
8		
9		
10		
11		
12		
13		
14		
15	4.723×10^{-7}	360
16	1.711×10^{-6}	560
17	4.723×10^{-7}	560
18	1.698×10^{-6}	505
19		505
20	1.698×10^{-6}	505
21	2.894×10^{-6}	560
22		
23		
24		
25	2.894×10^{-6}	
26	1.683×10^{-6}	
27	1.711×10^{-6}	
28	1.70×10^{-6}	
29	1.70×10^{-6}	
30	1.70×10^{-6}	560

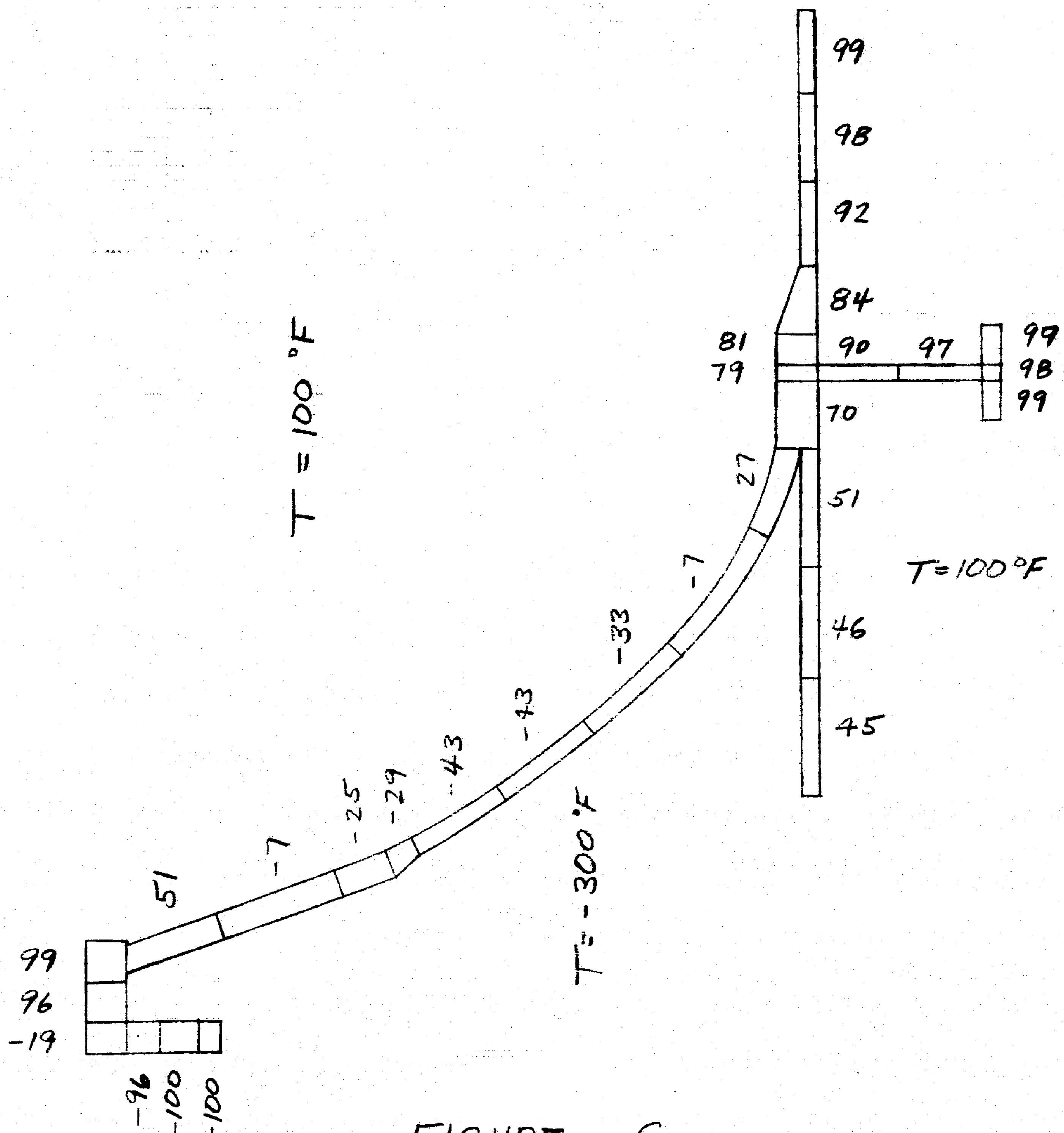

T = 100 °F
99
98
92
84
81
79
90
97
99
98
99
70
27
51
T=100°F
46
45
-7
-33
-43
-43
-29
-25
-7
51
T = -300 °F
99
96
-19
-96
-100
-100

FIGURE - 6

BY	DATE	SUBJECT	SHEET NO. OF
CHKD. BY	DATE		JOB NO.

II. TRANSIENT ANALYSIS OF BULKHEAD

IN ORDER TO CONSERVATIVELY BOUND THE TRANSIENT THERMAL STRESSES IN THE BULKHEAD, TWO CASES WILL BE INVESTIGATED

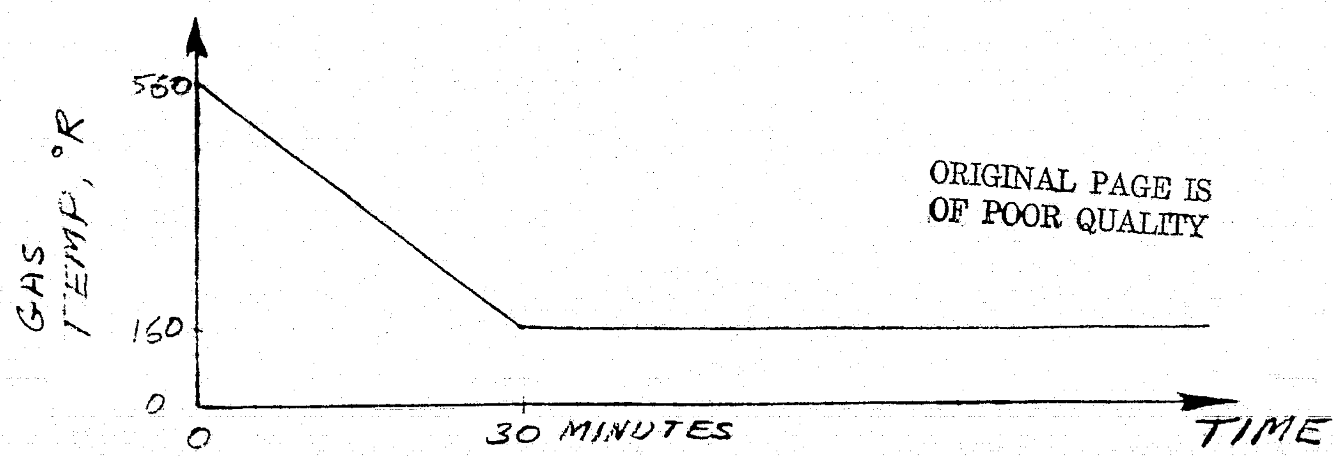
A- THE FLOW MODEL WILL BE SUBJECTED TO A THERMAL SHOCK FROM 560 °R DOWN TO 160 °R IN 30 MINUTES.

B- THE NON FLOW MODEL WILL BE SUBJECTED TO A THERMAL SHOCK FROM STEADY STATE TEMPERATURES (FIGURE 3) UP TO 560 °R IN 30 MINUTES.

A. THERMAL SHOCK TO COOL BULKHEAD

THE MODEL + ASSUMPTIONS ARE SAME AS FLOW CASE IN STEADY STATE CASE. THE GEOMETRY IS SAME ALSO AS SHOWN IN FIGURE 1.

TEMPERATURE DECREASE PLOT



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RESULTS

THE TEMPERATURE DISTRIBUTION CALCULATED IN THE TRANSIENT HEAT TRANSFER PROGRAM IS SHOWN IN FIGURE 7. THIS WORST CASE TO BRING PLEMUM DOWN TO 160°R OCURRED AFTER 30 MINUTES FROM START OF COOL DOWN. THE MAXIMUM TEMPERATURE DIFFERENCE IS 346 °F BETWEEN ELEMENTS 6 AND 7. THIS LARGE GRADIENT TEMPERATURE DISTRIBUTION AT TIME EQUAL TO 30 MINUTES WAS INPUT INTO THE "SPAR" PROGRAM TO CALCULATE THE RESULTANT STRESSES. THESE STRESSES ARE SHOWN IN FIGURE 8.

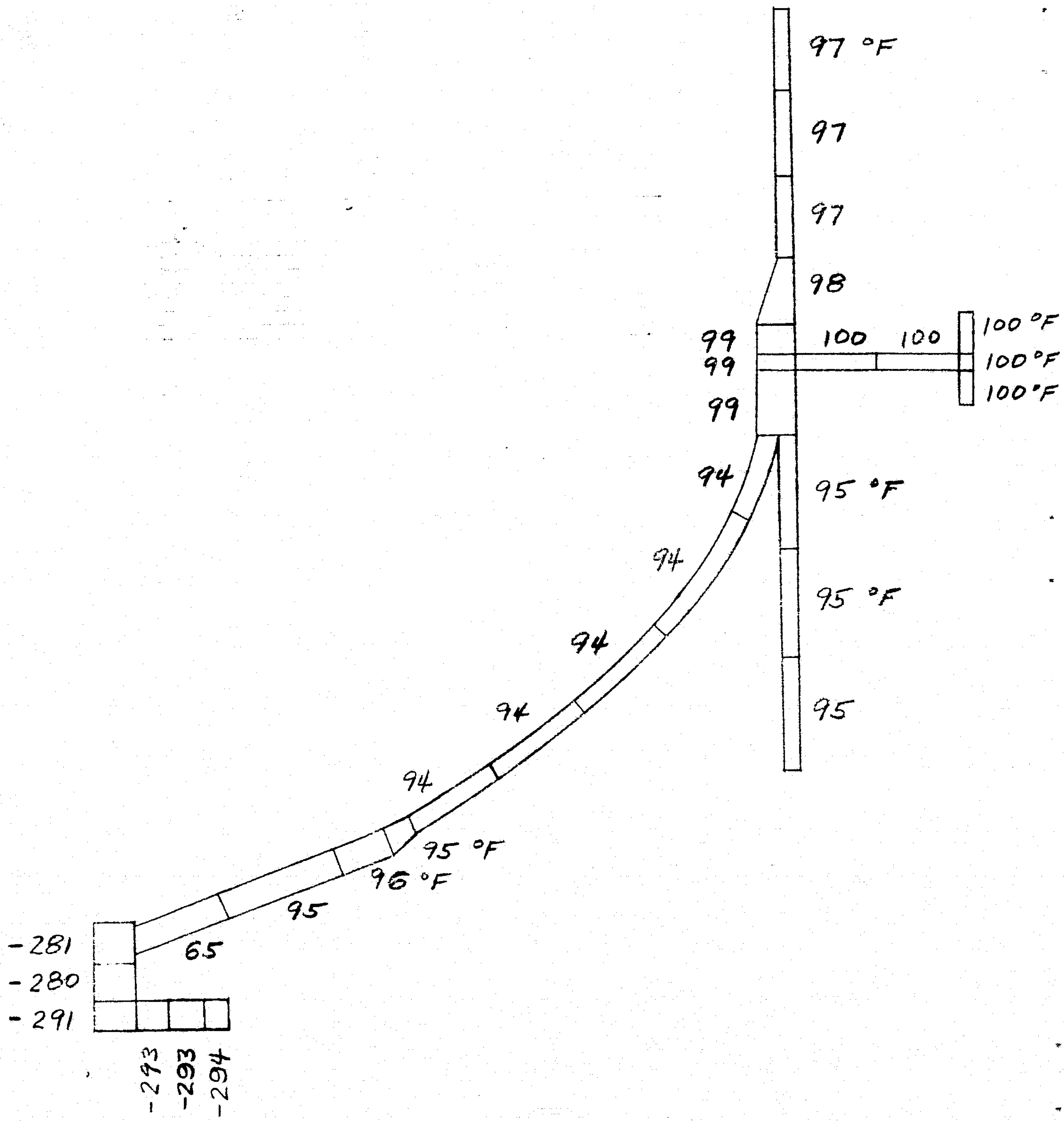
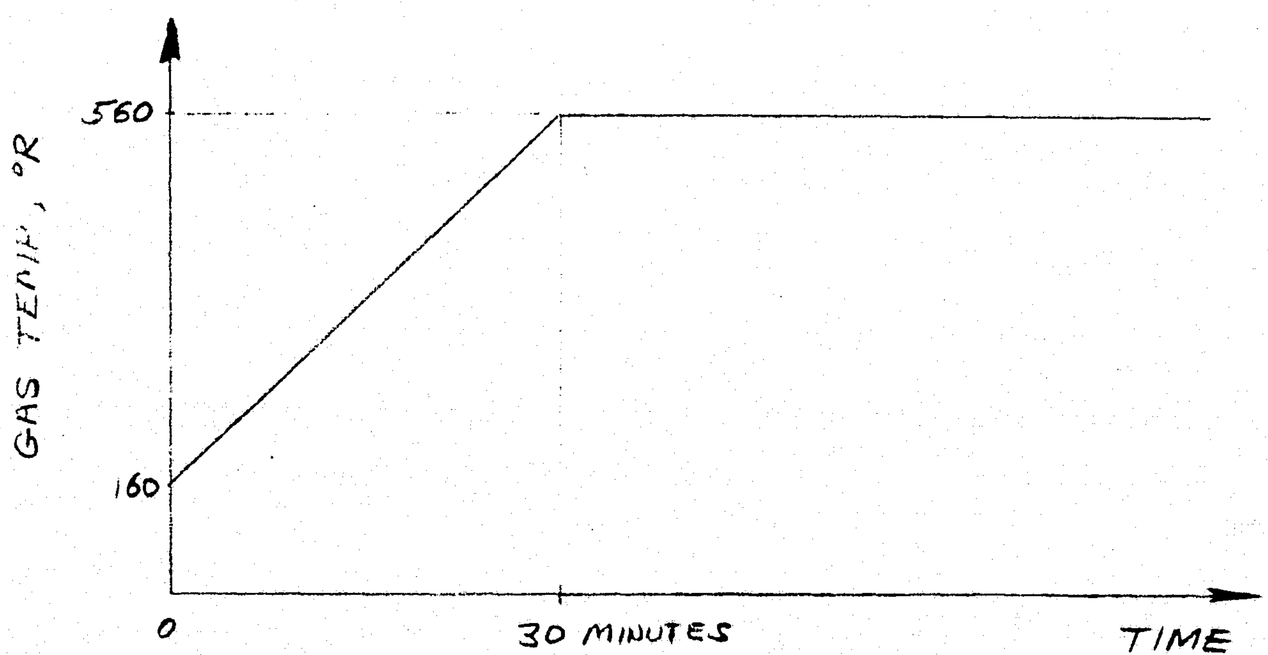

97 °F
97
97
98
99
99
99
100
100
100 °F
100 °F
100 °F
94
95 °F
94
95 °F
94
94
95
94
95 °F
96 °F
95
65
-281
-280
-291
-293
-293
-294

FIGURE - 7

B- THERMAL SHOCK TO HEAT BULKHEAD

THE MODEL + ASSUMPTIONS ARE SAME AS NONFLOW CASE IN STEADY STATE CASE. THE GEOMETRY IS SAME AS SHOWN IN FIG. 1. THE INITIAL TEMPERATURE OF BULKHEAD BEFORE HEAT UP IS SAME AS STEADY STATE DISTRIBUTION WITH FLOW. THIS WAS SHOWN IN FIGURE 3. THE ASSUMPTION IS MADE THAT THE HEAT UP STARTS AS SOON AS THE GATES ARE CLOSED.

TEMPERATURE INCREASE PLOT



RESULTS

THE TEMPERATURE DISTRIBUTION FOR THE 30 MINUTE HEAT UP TIME IS SHOWN IN FIGURE 9. THIS MAXIMUM TEMPERATURE OCURRS AT 30 MINUTES AFTER THE START OF HEAT UP. THE TEMPERATURE DIFFERENCE IS LARGEST BETWEEN ELEMENTS 6 AND 7. (ΔT = 323 °F). THIS TEMPERATURE DISTRIBUTION WAS INPUT INTO THE SPAR PROGRAM TO CALCULATE MAXIMUM STRESSES (THERMAL AND PRESSURE). THE STRESSES ARE SHOWN IN FIGURE 10.

THE MAX. STRESS IS -51 KSI WHICH IS BELOW THE ALLOWABE OF 52.5 KSI. NOW, RERUN THE TEMPERATURE PROGRAM FOR A HEAT UP TIME OF 4 HOURS. THIS TEMPERATURE DISTRIBUTION WHICH GIVES MAXIMUM GRADIENT IS SHOWN IN FIGURE 11. THE MAXIMUM GRADIENT FOR THIS CASE OCCURS BETWEEN ELEMENTS 4 AND 5. THIS CASE WAS INPUT INTO THE SPAR PROGRAM ALSO GIVING AN ACCEPTABLE STRESS VALUE OF -44 KSI. THE STRESS DISTRIBUTION FOR THE THIS THERMAL CASE AND 119 PSIG PRESSURE IS SHOWN IN FIGURES 12, 13 AND 14.

THE EFFECTS OF THE CHANGE IN THICKNESSES OF THE BUCKHEAD WERE CHECKED BY RE-RUNNING THE TRANSIENT HEAT TRANSFER PROGRAM. THESE THICKNESSES ARE SHOWN IN FIGURE 2. THE TEMPERATURES SHOWN IN FIGURE 15 ARE ALMOST EQUAL TO THOSE SHOWN IN FIGURE 11.

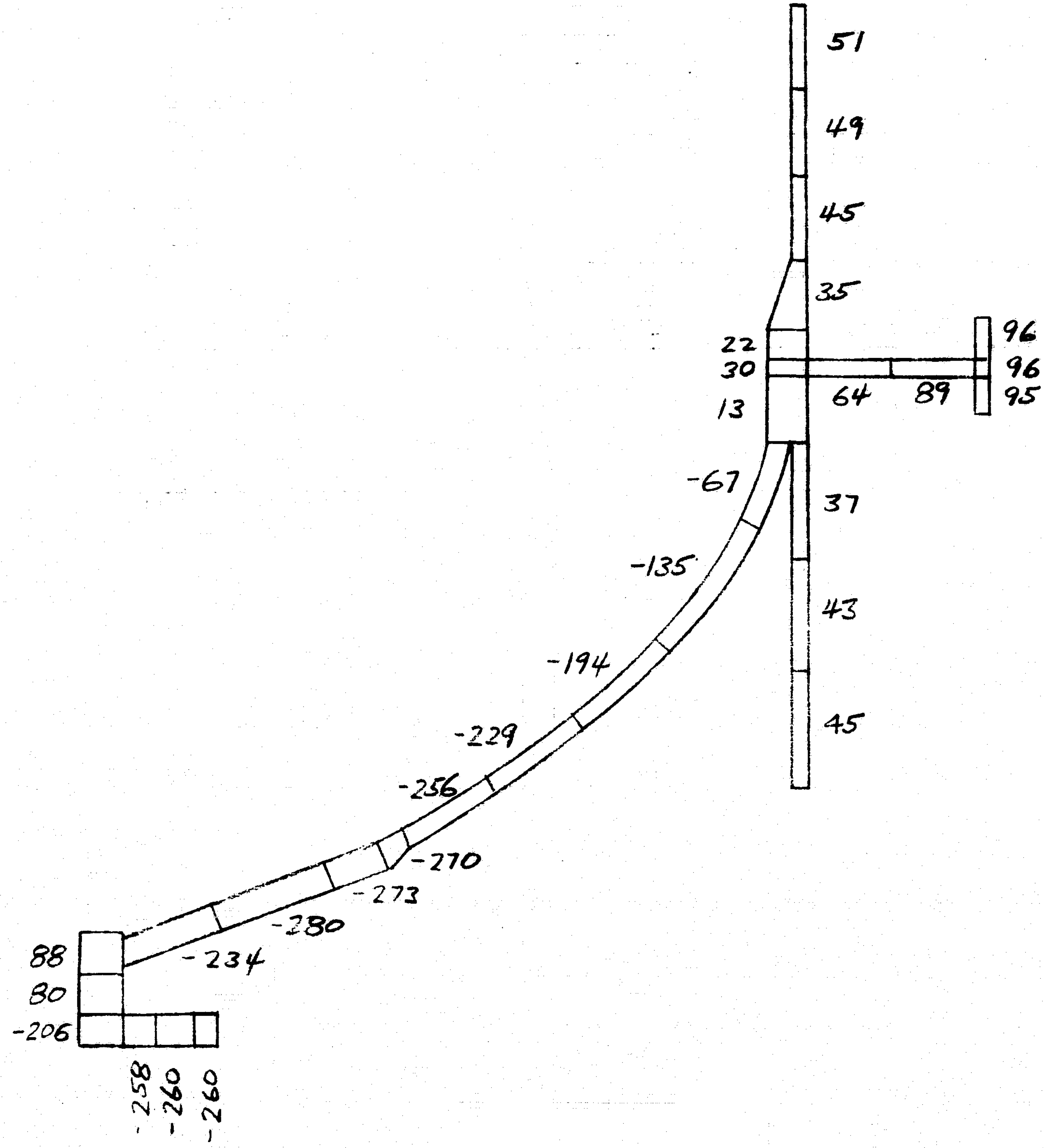


FIGURE - 9

(HEAT UP TIME OF 30 MINUTES)

STRESS INTENSITY

GATE VALVE CLOSED WITH TRANSIENT TEMPERATURE AND PRESSURE

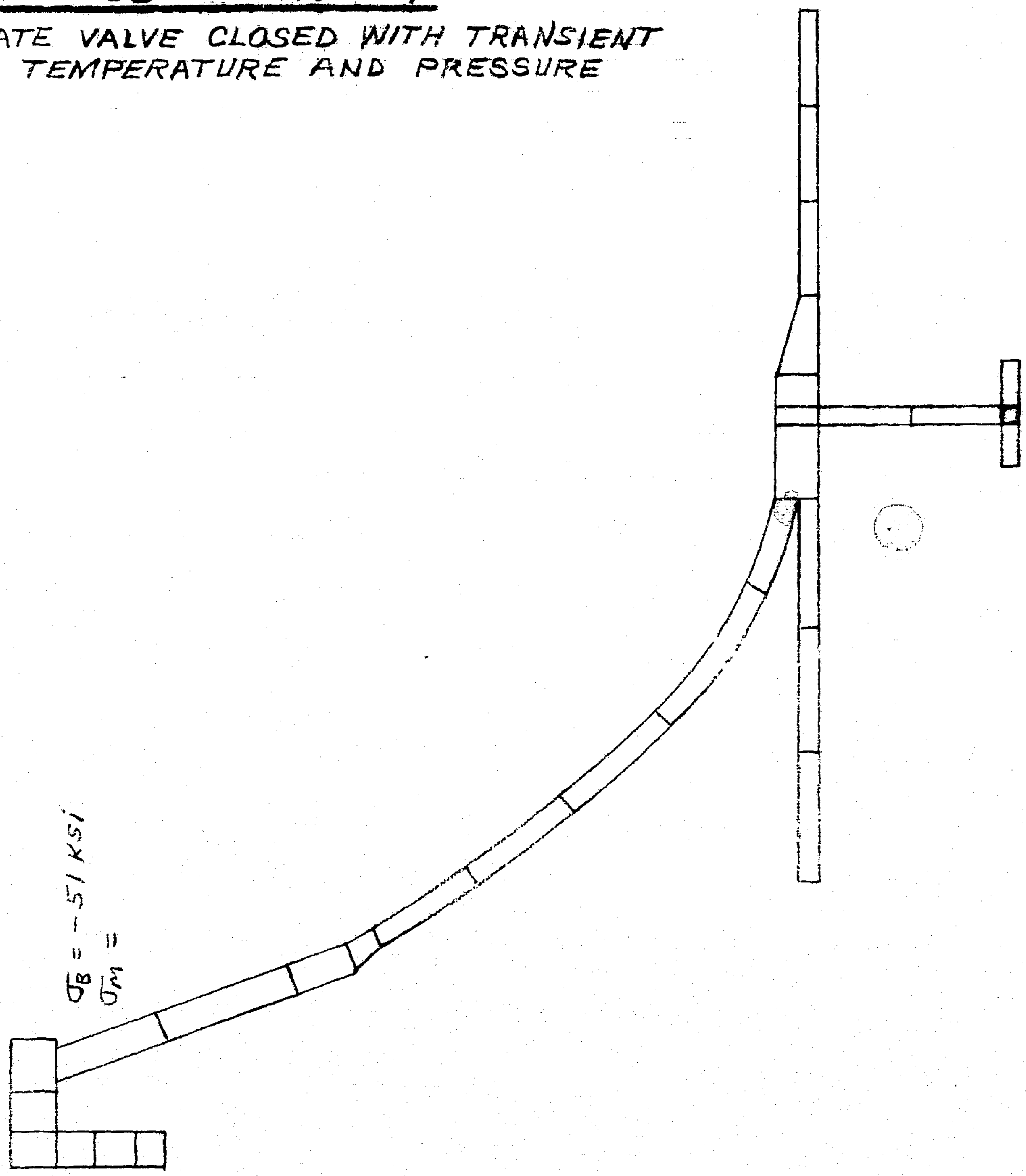


FIGURE - 10

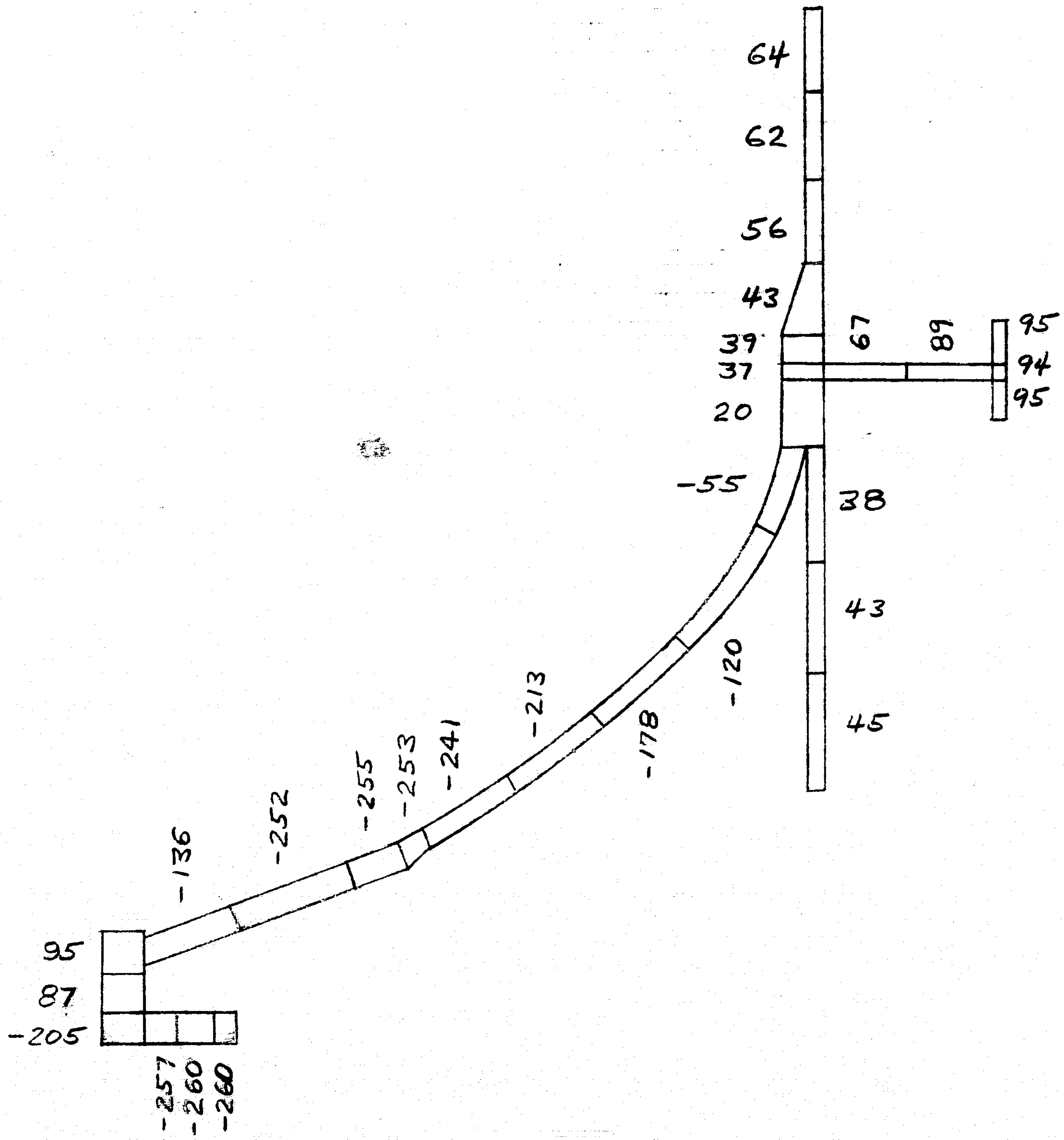


FIGURE - 11

(HEAT UP TIME OF 4 HOURS)

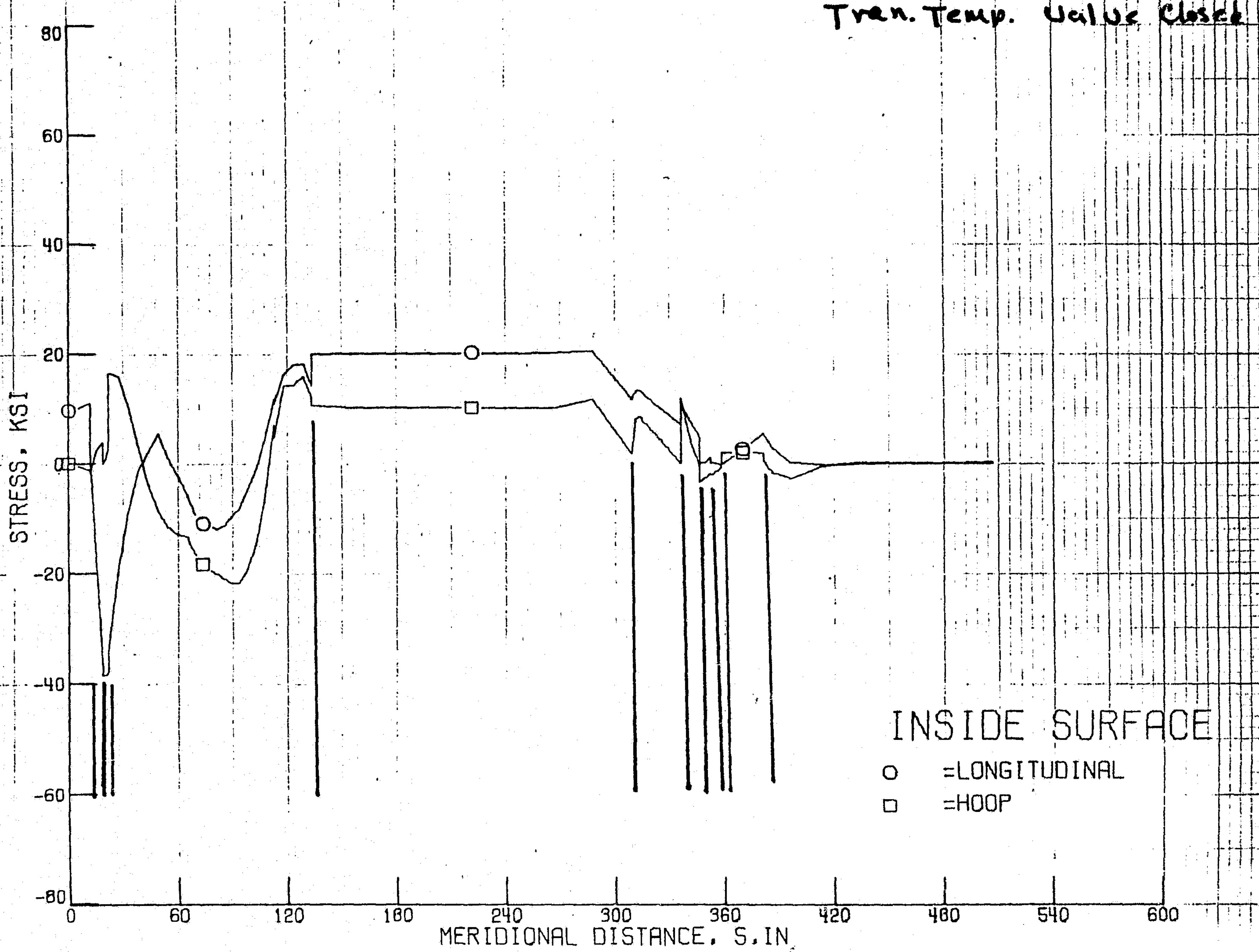

Tran. Temp. Valve Closed
STRESS, KSI
80
60
40
20
0
-20
-40
-60
-80
0
60
120
180
240
300
360
420
480
540
600
MERIDIONAL DISTANCE, S,IN
INSIDE SURFACE
○ =LONGITUDINAL
□ =HOOP

FIGURE 13

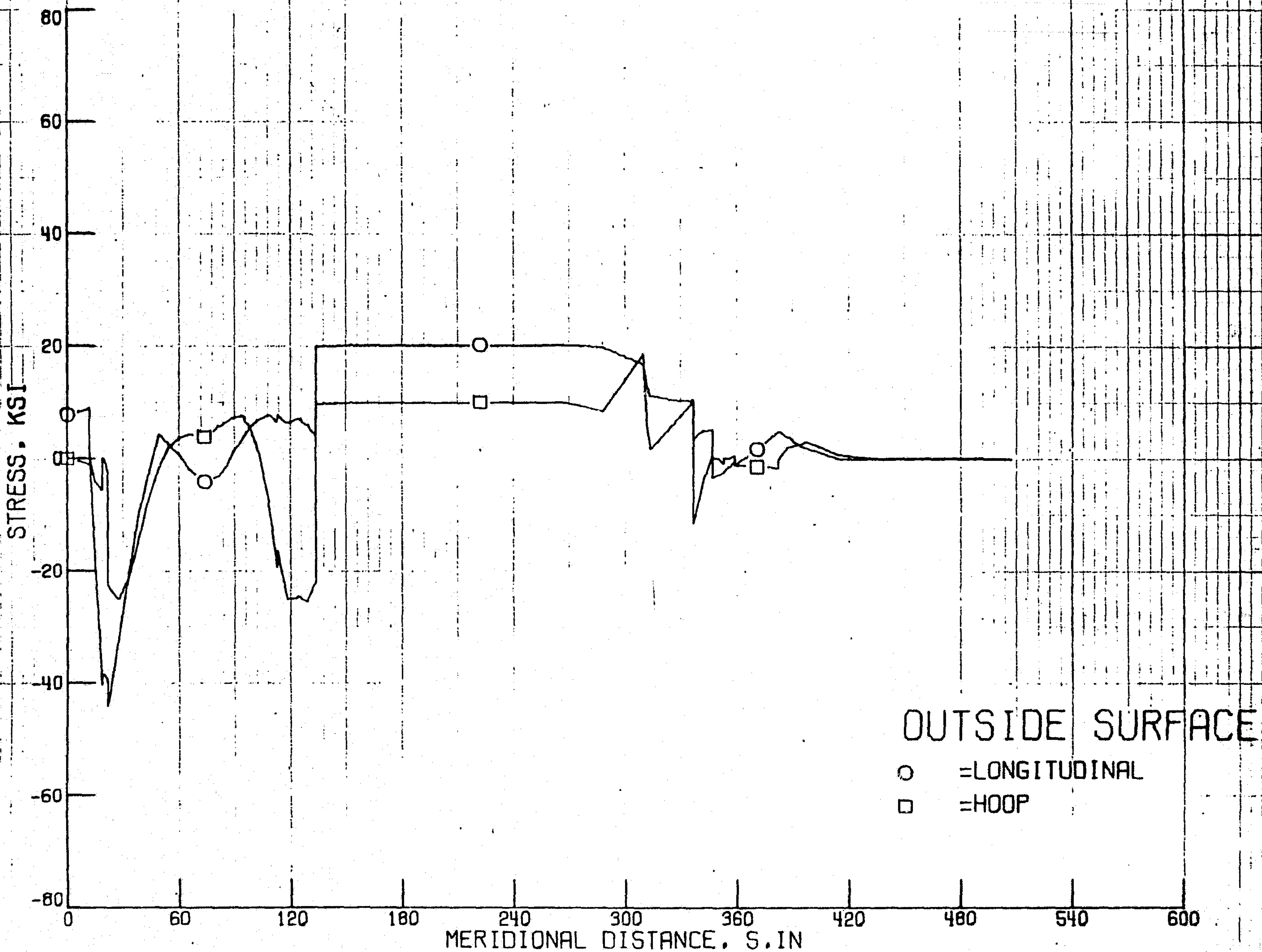
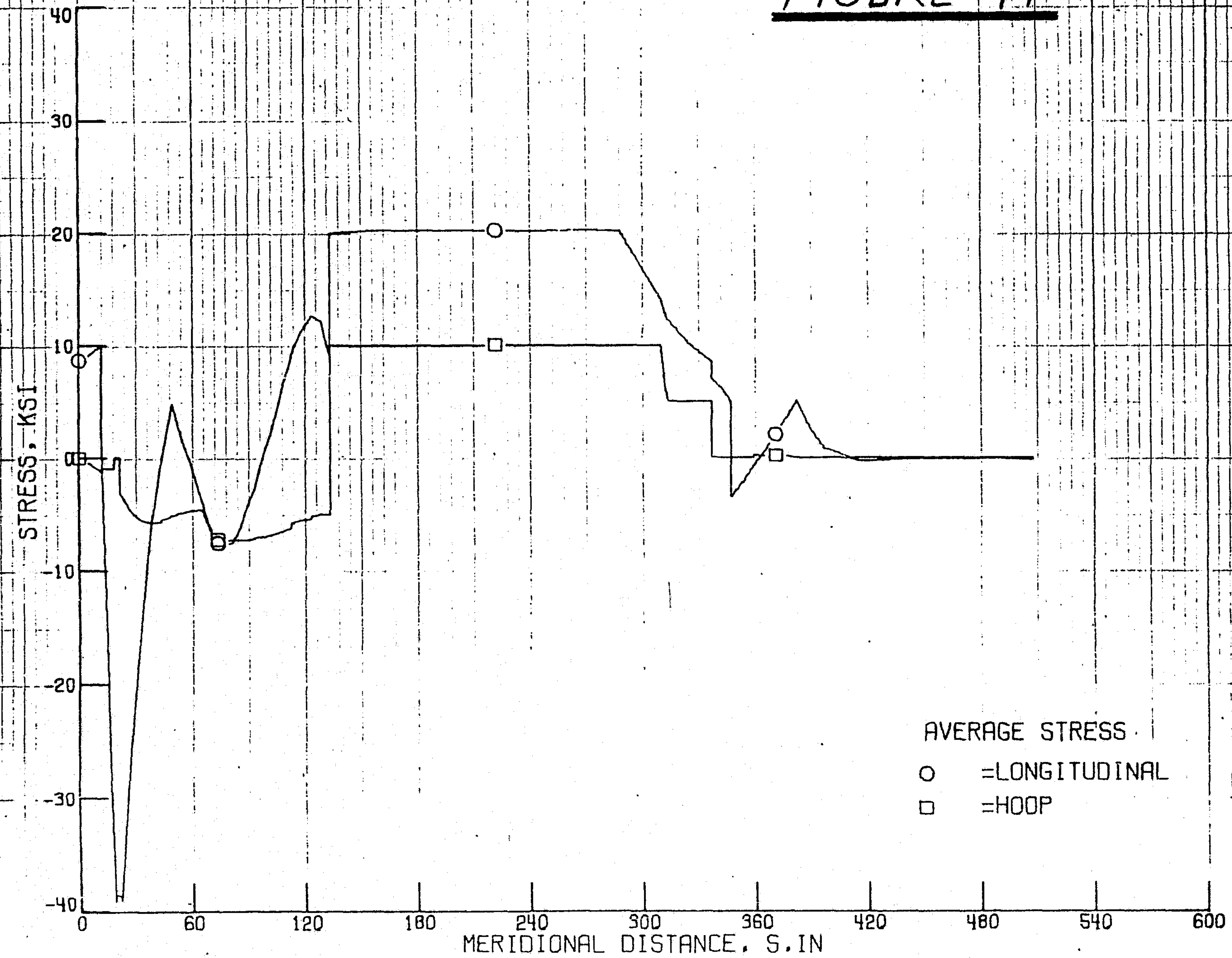

80
60
40
20
0
-20
-40
-60
-80
STRESS, KSI
0
60
120
180
240
300
360
420
480
540
600
MERIDIONAL DISTANCE, S.IN
OUTSIDE SURFACE
○ =LONGITUDINAL
□ =HOOP

FIGURE 14


40
30
20
10
0
-10
-20
-30
-40
STRESS, KSI
0
60
120
180
240
300
360
420
480
540
600
MERIDIONAL DISTANCE, S.IN
AVERAGE STRESS
○ =LONGITUDINAL
□ =HOOP

BY ______ DATE ______ SUBJECT ______ SHEET NO. 31 OF ______
CHKD. BY ______ DATE ______ JOB NO. ______

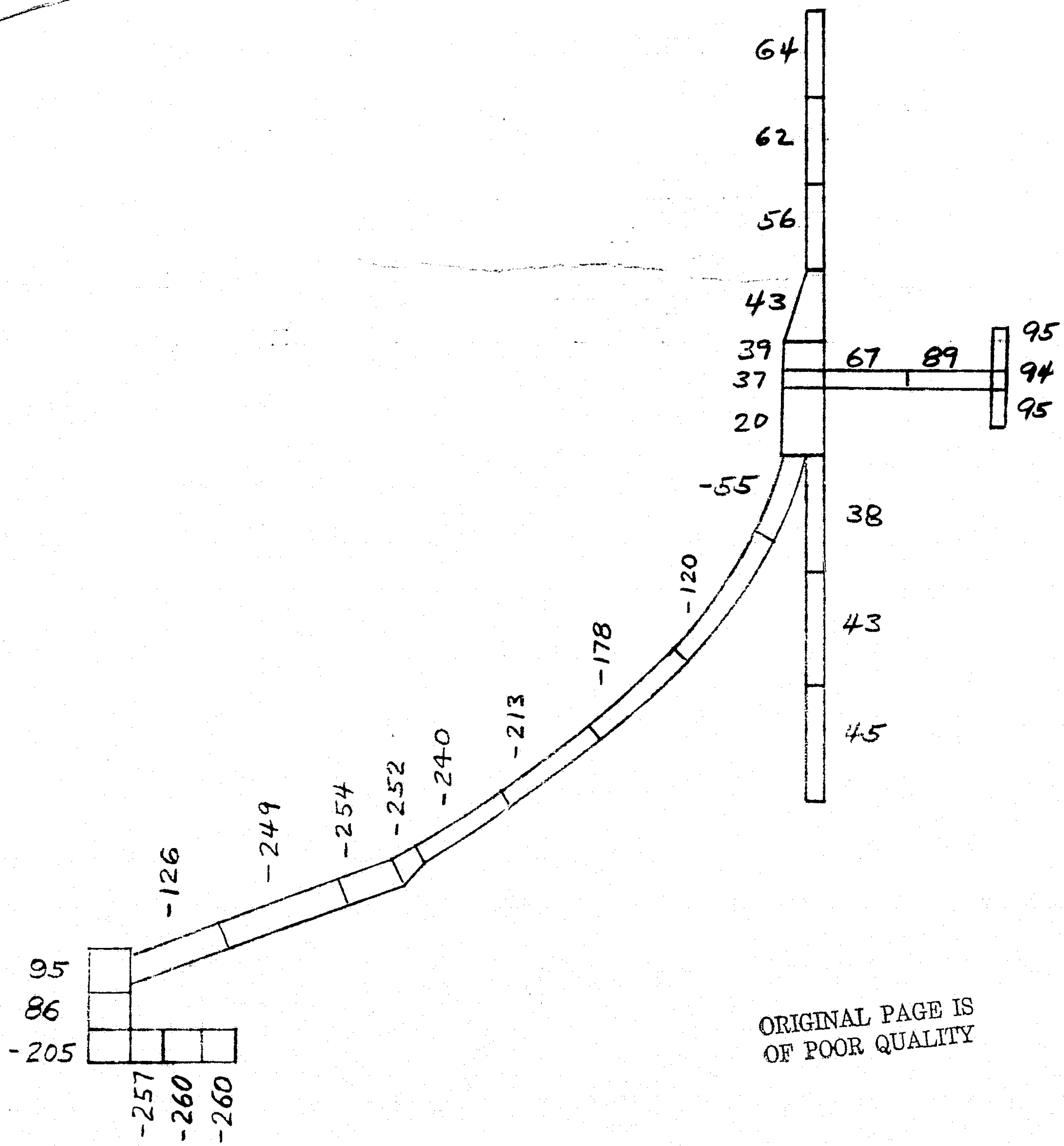


FIGURE 15

III – ACCIDENTAL EXPOSURE OF SHELL TO LN2 OR GN2

A thermal stress analysis of the pressure shell between corner rings S6 257 has been conducted for the local loss of insulation or LN_2 puddle. The thermal analysis indicates that the local loss of insulation will drive the bare shell temp. to within 3° of LN2 temp.; therefore, the LN2 puddle could not impose any more severe gradient than this, and it was not considered any further. The resulting thermal stresses for local loss of insulation peaked out (60,000 psi) for a 12° arc of bare shell. These stresses were superimposed to existing stresses at typical structural ring and elliptical ring to determine reduction in fatigue life for these areas.

Na = number of operating cycles with bare shell

L = Life (years)

Na	LIFE	
	TYP STRUCT RING	ELLIPTICAL RING WELD
0	31	15
1	31	15
10	29	15
50	25	14
100	21	12

Therefore the local loss of insulation or LN2 puddle would affect the fatigue life of sections of the tunnel differently. The important point is that this type of accident needs to be detected before a large number of cycles are accumulated.

Detail supporting calculations follow.

ACCIDENTAL EXPOSURE OF SHELL TO LN2 or GN2

Two types of accidents can occur which would expose the shell to LN2 or GN2

1. Loss of insulation
   this would expose shell to gaseous N_2

2. LN2 Puddle

BY ________ DATE 3/6 SUBJECT ________ SHEET NO. 3 OF ____
CHKD. BY ______ DATE ______ JOB NO. ________

I LOSS OF INSULATION

The worse place to loose insulation is the region where insulation is the flow liner, and the flow has a high velocity. This occurs in the short leg between corner rings 56 & 57.

A FILM COEF

Gas Film Coef.

The flow area changes in the short leg. The entrance has a 16' DIA AND flow m[illegible] an annulus is formed by the upstream nacelle. Therefore, will calculate an average coef.

Annulus: $D_o = 20 ft$ D_i $10 ft$

$$A = \frac{\pi}{4}(20^2 - 10^2) = 235.62 ft$$

$$\text{Average } A = \frac{1}{2}\left[235.62 + \frac{\pi 16^2}{4}\right] = 218 \ ft^2$$

$$RE = \frac{\dot{m} D}{\mu A}$$

Assume @ Test section $M = 1$

$P_o = 1$ ATM (given coldest temp)

$T_o = -320°F$

$$\text{Test section area} - \left(2.5 m \times 3.2808 \ ft/m\right)^2 = 67.27 \ ft^2$$

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$$\frac{A}{A^*} = \frac{218}{67.27} = 3.25 \Rightarrow M = .18 \quad P/P_0 = .9776 \quad \frac{T}{T_0} = .994$$

$$M = 1 \quad \frac{P}{P_0} = .528 \quad \frac{T}{T_0} = .8333$$

$$P_{TS} = 1\,atm(.528) = .528\ ATM$$

$$T_{TS} = (140)\,.8333 = 116.66\,°R$$

Short leg Areas:-

$$P_{SL} = .9776\ ATM \quad T_{SL} = 139\,°R$$

$$\mu = 2.16 \times 10^{-8}\ \frac{slugs}{ft\text{-}sec\sqrt{°R}} \left[\frac{139^{3/2}}{139 + 184}\right] \frac{32.17\ lbm}{slug} = 3.524 \times 10^{-6}$$

$$\mu = 3.524 \times 10^{-6}\ \frac{lbm}{ft\text{-}sec}$$

$$\dot{m} = 45{,}000\ \frac{lbm}{sec}$$

For circle $\frac{\pi D^2}{4} = A$ or $D = \sqrt{\frac{4A}{\pi}}$

16.66 ft

$$RE = \frac{\left(45{,}000\ \frac{lbm}{sec}\right)\sqrt{\frac{4\ 218\ ft^2}{\pi}}}{3.524 \times 10^{-6}\ \frac{lbm}{sec\text{-}ft}\ 218\ ft^2} = 9.76 \times 10^8 \Leftarrow \text{Turbulent pipe flow}$$

using pipe flow equations based on bulk fluid temps for $\Delta T \le 100°F$

$$N_{NU} = .023\,(N_{RE})^{.8}(N_{PR})^{.4}$$

$P_r = .739 \quad K = .01045 \frac{Btu}{hr\,ft\,°F} \Leftarrow$ estimates

$$h_g = \frac{(.023)(9.76\times10^{8})^{.8}(.739)^{.4}\left[.01045 \frac{Btu}{hr\,ft\,°F}\right]}{16.66\,ft} = 198.7 \frac{Btu}{ft^2\,hr\,°F}$$

Apply short length correction factor to midpoint of short leg:-

$$h_g = 198.7 \left(\frac{16.6}{42}\right)^{1/18} = 202$$

$$\therefore h_g = 200 \frac{Btu}{ft^2\,hr\,°F}$$

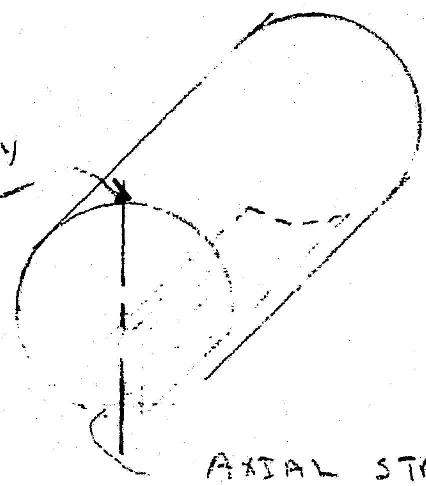
Outside Coef:-

$$h_o = .18\,(\Delta T)^{1/3} \frac{Btu}{ft^2\,hr\,°F} \qquad T_o = 100°F$$

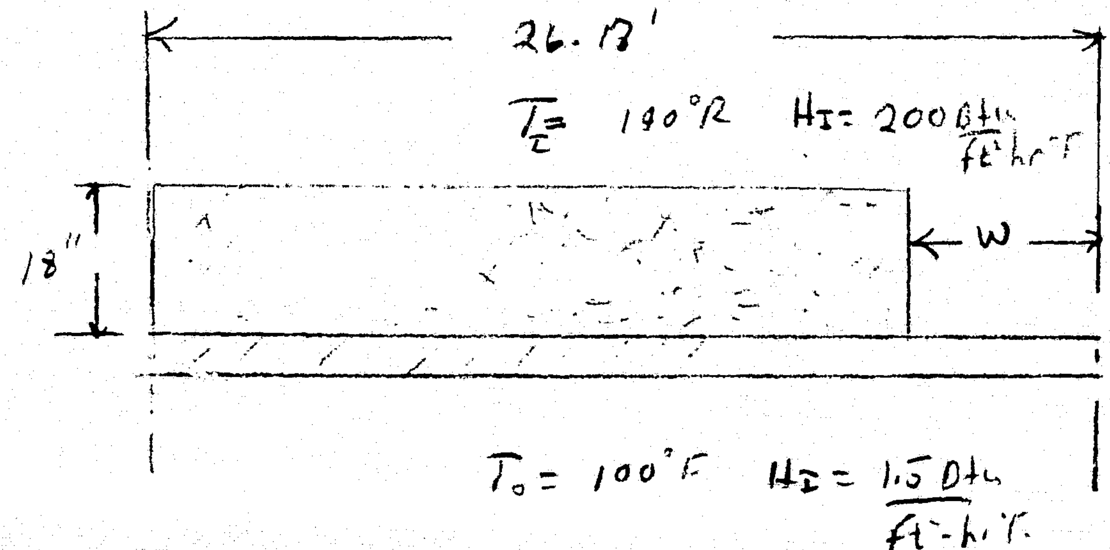
use $h_o = 1.5 \frac{Btu}{ft^2\,hr\,°F}$ as 1st estimate

B. THERMAL MODEL

The short leg will be used as the typical section to model. It will be assumed that a section of insulation will be removed for the entire length of leg.



Symmetry will be taken advantage of, and the shell will be unwrapped to form a linear model.



BY________ DATE________ SUBJECT________________________ SHEET NO. 39 OF______
CHKD. BY______ DATE________ ________________________ JOB NO.________

C. COMPUTER INPUT

The width of the insulation loss will be varied.

The insulation will be treated as an effective film coef. for modeling purposes and the shell will be divided into 30 blocks (maximum that program will handle

LEN = 26.17/30 = .87 ft or 10.47 in

WID = .167 in

VOL = 17.01 for 1" thk

Effective Film Coef inside:-

For a one Dimensional heat balance on insulated plate:

$$Q = \frac{T_o - T_s}{\frac{1}{h_I A_I} + \frac{t_I}{K_I A_c}}$$

For effective film coef:-

$$Q = h_{eff} A_{eff} (T_s - T_I)$$

$$h_{eff} A_{eff} = \frac{1}{\frac{1}{h_I A_I} + \frac{t_I}{K_I A_c}}$$

Neglecting curvature of shell:-

$A_I = A_c = A$

$$h_{eff} = \frac{1}{\frac{1}{h_I} + \frac{t_I}{K_I} + }$$

For insulated shell:-

$$h_{eff} = \frac{1}{\frac{1}{1.1389 \frac{Btu}{in^2 hr °F}} + \frac{18 in \times 144 in^2/ft^2}{1.47 \frac{Btu\text{-}in}{ft^2 hr °F}}} = 5.669 \times 10^{-4} \frac{Btu}{in^2 hr °F}$$

For uninsulated shell:-

$$h_{eff} = h_g = 1.389 \frac{Btu}{in^2 hr °F}$$

From previous work on bulkheads; the eff. film coeff. & Temps for blocks with different convective boundary conditions:-

$$h_{eff} = \frac{h_i A_i + h_o A_o}{A_i + A_o}$$

For $A_i = A_o = A$

$$h_{eff} = \frac{(h_i + h_o) A}{2A} = \frac{h_i + h_o}{2}$$

$$T_{eff} = \frac{h_i A_i T_i + h_o A_o T_o}{h_i A_i + h_o A_o}$$

$$T_{eff} = \frac{(h_i T_i + h_o T_o)}{h_i + h_o}$$

BY	DATE	SUBJECT	SHEET NO.	OF
CHKD. BY	DATE		JOB NO.	

For the insulated blocks:-

$$h_{eff} = \left[\frac{5.669\times10^{-4} + \frac{1.5}{144}}{2}\right] = .00549 \ \frac{Btu}{in^2\,hr\,°F}$$

$$T_{eff} = \left[\frac{(5.669\times10^{-4})(140) + (.01042)560}{2(.00549)}\right] = 539°$$

For the uninsulated blocks:-

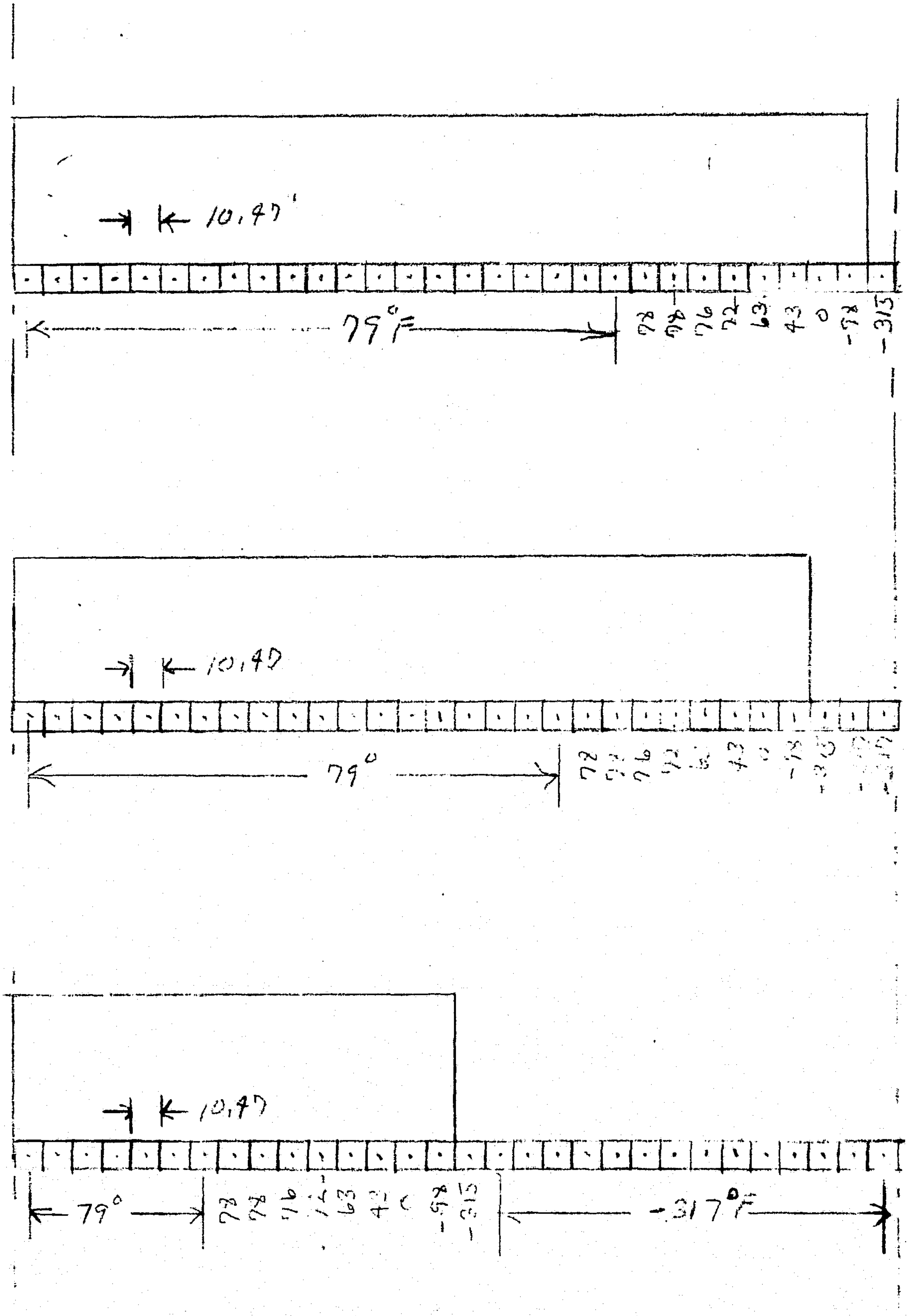
$$h_{eff} = \frac{1.389 + .01042}{2} = .7 \ \frac{Btu}{in^2\,hr\,°F}$$

$$T_{eff} = \left[\frac{(1.389)(140) + (.01042)560}{2(.7)}\right] = 143\ °R$$

$A_{COND} = (1)(.67) = .67\ in^2$

$CROAREA = 2A = (10.47)(1) = 10.47\ in^2$

BY________ DATE________ SUBJECT____________________ SHEET NO. 97 OF______
CHKD. BY________ DATE________ JOB NO.________



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CHKD. BY ______ DATE ______ JOB NO. ______

II LN$_2$ PUDDLING

Liquid Nitrogen puddling is a more complex problem than insulation loss. However, the resulting temperature distribution can be no worse than insulation loss because the bare shell temp. with no insulation is within 3° of the LN$_2$ temp. Therefore the results from the "insulation loss" case will bracket both of these accident problems.

III THERMAL STRESS IN SHELL

A CLOSED FORM SOLUTION

A closed form solution will be used to estimate the thermal stresses in the short leg region of the shell. This region will be modeled as a right circular cyl. with constant temp. thru the thk and circumferential temp. variation. This type of temp. dist. will cause thermal stresses in both the hoop and axial directions. However, due to the flexibility of a thin shell in the hoop direction (as compared to axial direction) the hoop stresses will be small compared to those in the axial direction. Therefore, only those stresses in the axial direction will be considered.

From ref 1, Axial stress (σ_x):-

$$\sigma_x = -\alpha E T(\phi) + \frac{E}{2\pi}\int_0^{2\pi} \alpha T(\phi)\, d\phi + \frac{E \sin\phi}{\pi}\int_0^{2\pi} \alpha T(\phi) \sin\phi\, d\phi$$

$$+ \frac{E \cos\phi}{\pi}\int_0^{2\pi} \alpha T(\phi) \cos\phi\, d\phi$$

The above equation is for NO constraint.
The second term is dropped for axial constraint and the last two are dropped for bending constraint. These equations were programmed for 3 types of boundary conditions:

1. Completely constrained
2. Bending restraint only
3. NO Restraint

```
PROGRAM LN2STRS (INPUT, OUTPUT)
DIMENSION [illegible], TEMP(100), SUM(10), WK(10)
COMMON R,H
EXTERNAL FX1,
READ *, E, ALPHA, PI, NTEMP
READ *, TEMP

100 CONTINUE
   READ*, A,B,H,N


   CALL SIMP(A,B,FX,H,N,SUM,WK,IERR)
    IF (IERR.NE.0.) GO TO 500
   DO 10 I=5,NTEMP,5
10   PHI= I*H/R
   SIGX1= - ALPHA*E * TEMP(I) + E/PI *(SUM(1)/R

     + SIN(PHI)* SUM(2) + COS(PHI) SUM(3)
     THETA= 180.* PHI /PI
   PRINT *, THETA, SIGX
10  CONTINUE
```

A= 0
B= 2πR= [illegible] = 628.2
H= 10.47
N=3
R= 99.96
E= 29X10^6
ALPHA= 5.5X10^-6
PI= 3.14159
NTEMP= 61

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R= 99.96

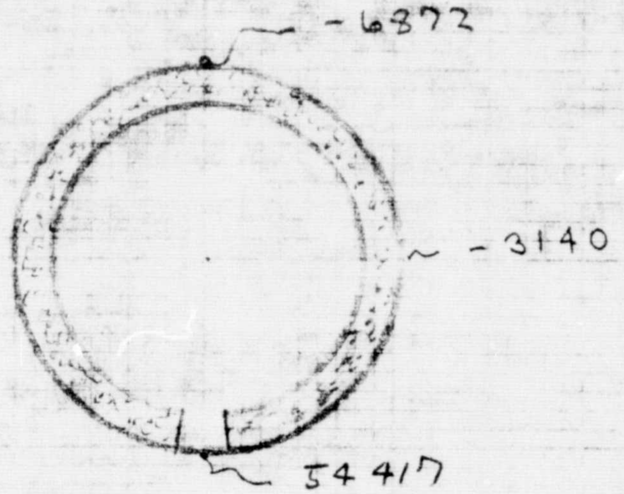
STRESSES

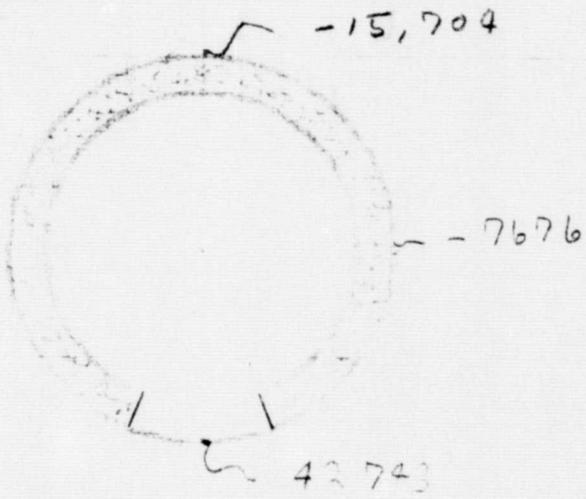
```
                     LF*
 PROGRAM LN2STRS(INPUT,OUTPUT)
 DIMENSION SUM(10),WK(10)
 COMMON R,H,TEMP(61)
 EXTERNAL FX
 READ*,A,B,H,N,E,ALPHA,PI,NTEMP,R
 READ*,TEMP
 CALL SIMP(A,B,FX,H,N,SUM,WK,IERR)
 PRINT*,SUM(1),SUM(2),SUM(3)
 IF(IERR.EQ.0) 20,30
0 PRINT*,IERR
0 DO 10 I=1,NTEMP,1
 II=I-1
 PHI=II*H/R
 SIGN=E*ALPHA/PI*(-PI*TEMP(I)+SIN(PHI)*SUM(2)+[illegible]
.+COS(PHI)*SUM(3))
 THETA=180.*PHI/PI
 PRINT*,THETA,TEMP(I),SIGN
0 CONTINUE
 STOP
 END
 SUBROUTINE FX(X,WK)
 DIMENSION SUM(10),WK(10)
 COMMON R,H,TEMP(61)
 PHI=X/R
 N=X/H+1.5
 WK(1)=TEMP(N)/R
 WK(2)=SIN(PHI)*TEMP(N)/R
 WK(3)=COS(PHI)*TEMP(N)/R
 RETURN
 END
D OF FILE-
```

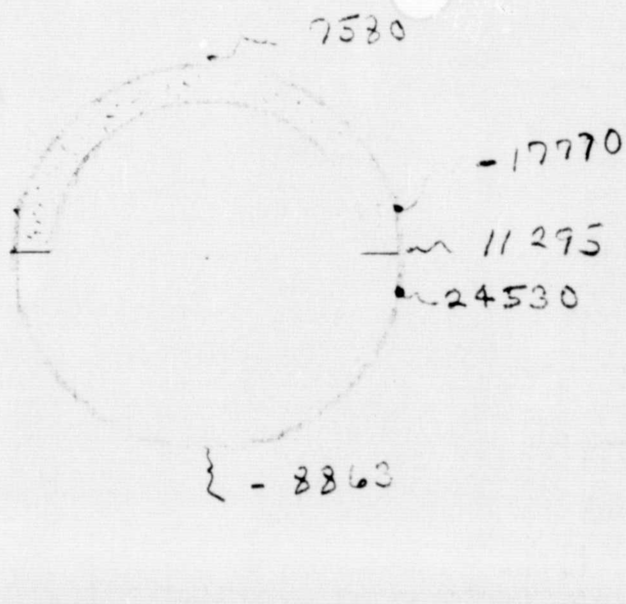
BY ________ DATE ________ SUBJECT PEAK THERMAL STRESSES IN SHELL SHEET NO. 47 OF ________
CHKD. BY ________ DATE ________ JOB NO. ________

NO RESTRAINT

NOTE:
VALUES
TAKEN FROM
FOLLOWING
PAGES







Program LN25TR3

No constraint

1 block

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18.00382107458 22. 846.2225565801
24.00509476611 53. -3918.012613223
30.00636845764 68. -6073.002973062
36.00764214917 74. -6737.85182775+
42.00891584069 77. -6872.76139886
48.01018953222 78. -6640.997725621
54.01146322375 78. -6206.349036014
60.01273691528 79. -3892.580490344
66.0140106068 79. -5385.870462597
72.01528429833 79. -4851.297181925
78.01655798986 79. -4294.69610747
84.01783168139 79. -3722.176053784
90.01910537292 79. -3140.01232077
96.02037906444 79. -2554.585911173
102.021652756 79. -1972.31358937
108.0229264475 79. -1399.577549315
114.024200139 79. -842.6554553232
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126.0267475221 79. 199.5697964872
132.0280212136 79. 673.4492925214
138.0292949051 79. 1108.732727847
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156.0331159797 79. 2138.309925847
162.0343896712 79. 2376.768500466
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174.0369370543 79. 2680.081365072
180.0382107458 79. 2741.6110917
186.0394844374 79. 2741.935969912
192.0407581289 79. 2681.052438769
198.0420318204 79. 2559.627832831
204.043305511 79. 2378.993867080
210.044579280 79. 2141.12805153
216.045052095 79. 1848.630934896
222.047126506 79. 1504.734781936
228.048400279 79. 1113.131932186
234.049673969 79. 678.2731832315
240.0509476611 79. 204.775499581
246.0522213526 79. -302.121136771
252.053495044 79. -836.880863331
258.054768735 79. -1393.502331753
264.056042427 79. -1966.18346637
270.057316113 79. -2548.388070034
276.058589103 79. -3133.314696287
282.059863501 79. -3716.046576256
288.061137191 79. -4288.701961119
294.062410849 79. -4845.504066752
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312.0662319594 79. -6201.520090133
318.067505651 79. -6636.611438808
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[illegible]
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354.0751478001 -49. 12050.64539184
360.0764214917 -207. 37190.50239413
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[illegible]
 [illegible]
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Bending const only

No constr

3 slack',

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66.0140106068 78. -12931.82680069
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78.01655798986 79. -10460.18910144
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90.01910537292 79. -7675.705873808
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306.0649582679 78. -14132.84594247
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318.067505651 74. -15686.99480203
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78.01655798986 -317. 20608.36434366
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90.01910537292 -207. 11294.61080184
96.02037906444 -49. -9733.500044333
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216.045852895 79. 1201.573228869
222.0471265865 79. -1251.669636437
228.0484002781 78. -3884.086159704
234.0496739696 78. -6986.465676375
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246.0522213526 74. -13339.21741535
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276.0585898103 -207. 11293.74049895
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306.0649582679 -317. 9215.149652042
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348.0738741086 -317. -7567.293521429
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ILLEGAL CONTROL CARD.
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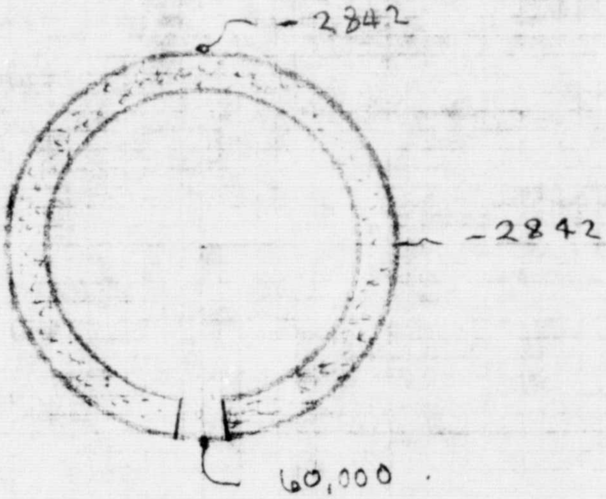
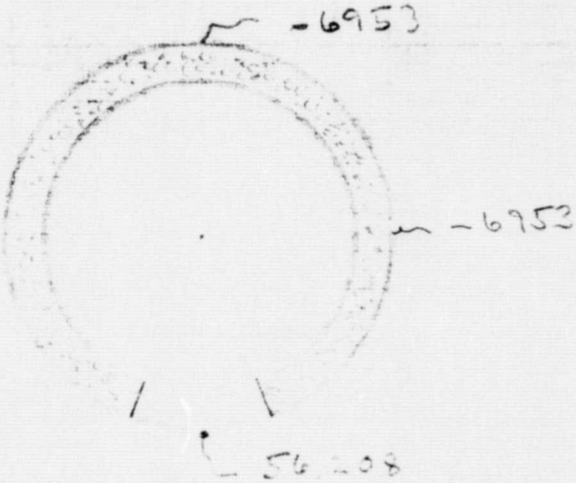
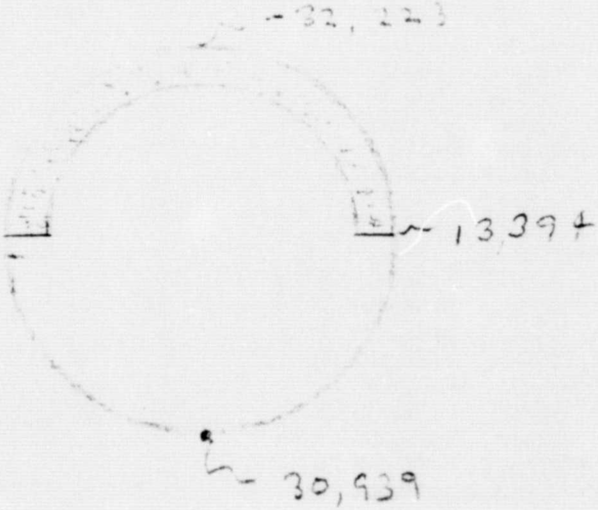
No constr is slack

CONSTRAINED IN BENDING ONLY







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336.0713267256 68. -6070.188520901
342.0726004171 53. -3915.79189514
348.0738741086 22. 847.8251996648
354.0751478001 -49. 12050.64533184
360.0764214917 -207. 37190.50239413
 ILLEGAL CONTROL CARD.
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156.0331159797 79. -2842.345626553
162.0343896712 79. -2842.345626553
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174.0369370543 79. -2842.345626553
180.0382107458 79. -2842.345626553
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192.0407581289 79. -2842.345626553
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204.0433055119 79. -2842.345626553
210.0445792035 79. -2842.345626553
216.045852895 79. -2842.345626553
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234.0496739696 79. -2842.345626553
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246.0522213526 79. -2842.345626553
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276.0585898103 79. -2842.345626553
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318.067505651 78. -2682.845626553
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 ILLEGAL CONTROL CARD.
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Bending const o.d,
1 Block

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78.01655798986 79. -6953.001392588
84.01783168139 79. -6953.001392588
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96.02037906444 79. -6953.001392588
 102.021652756 79. -6953.001392588
108.0229264475 79. -6953.001392588
114.024200139 79. -6953.001392588
120.0254738306 79. -6953.001392588
126.0267475221 79. -6953.001392588
132.0280212136 79. -6953.001392588
138.0292949051 79. -6953.001392588
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276.0585898103 79. -6953.001392588
282.0598635018 79. -6953.001392588
288.0611371933 79. -6953.001392588
294.0624108849 79. -6953.001392588
300.0636845764 78. -6793.501392588
306.0649582679 78. -6793.501392588
312.0662319594 77. -6634.001392588
318.067505651 74. -6155.501392588
324.0687793425 68. -5198.501392588
330.070053034 53. -2806.001392588
336.0713267256 22. 2138.498607412
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354.0751478001 -316. 56049.49860741
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ILLEGAL CONTROL CARD.
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Bending constraint

3 blocks

Bending ...

15 Slots

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114.024200139 68. -30468.66465287
120.0254738306 74. -31425.66465287
126.0267475221 77. -31904.16465287
132.0280212136 78. -32063.66465287
138.0292949051 78. -32063.16465287
144.0305685967 79. -32223.16465287
150.0318422882 79. -32223.16465287
156.0331159797 79. -32223.16465287
162.0343896712 79. -32223.16465287
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192.0407581289 79. -32223.16465287
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204.0433055119 79. -32223.16465287
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216.0458528895 79. -32223.16465287
222.0471265965 79. -32223.16465287
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234.0496739696 78. -32063.66465287
240.0509476611 77. -31904.16465287
246.0522213526 74. -31425.66465287
252.0534950442 68. -30468.66465287
258.0547687357 53. -28076.16465287
264.0560424272 22. -23131.66465287
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282.0598635018 -316. 30779.33534713
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342.0726004171 -317. 30938.83534713
348.0738741086 -317. 30938.83534713
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ILLEGAL CONTROL CARD.
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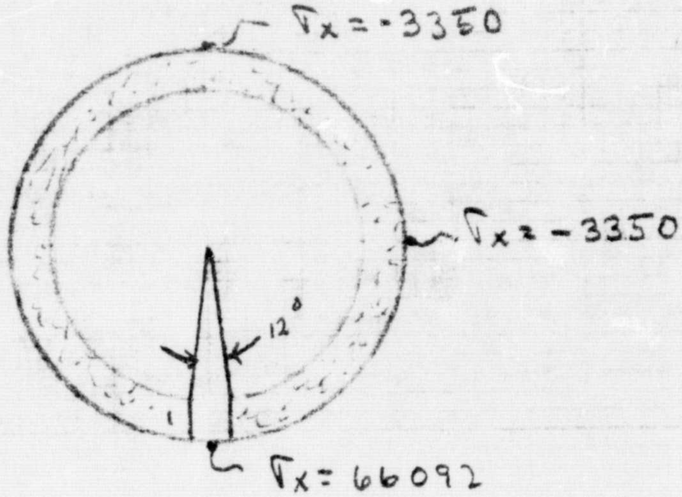
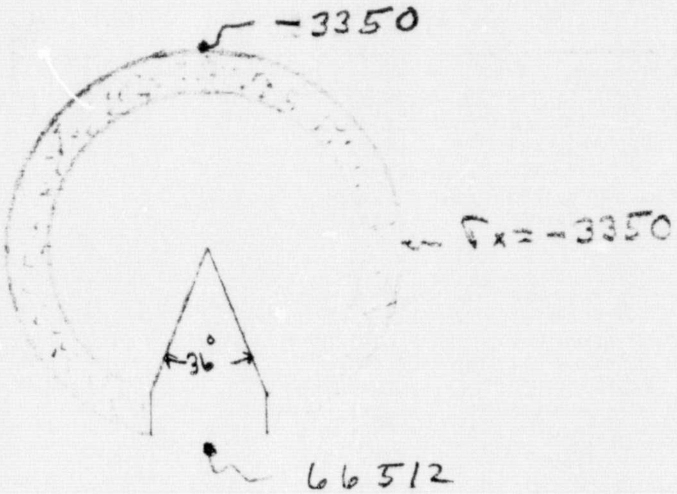
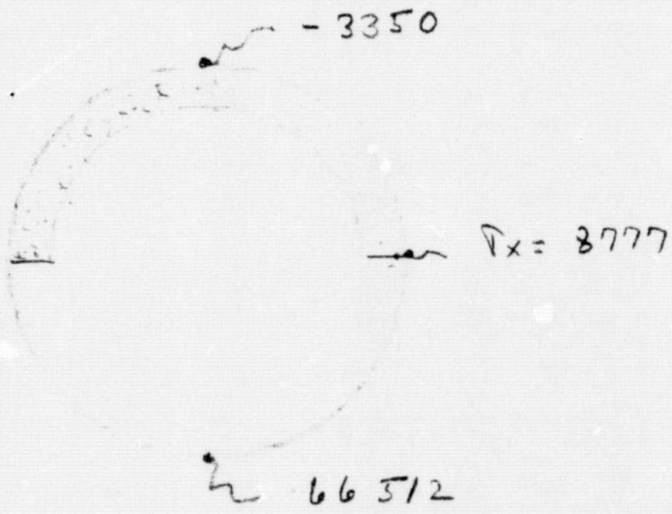
COMPLETELY RESTRAINED CYL.







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$CALL(PLN2(IN=LN2STRS,DATA=LN2DAT)
/RFL,45000
RFL,45000.
-PLN2
$CALL(PLN2.
/GET,PLN2
/LIST,F=PLN2
GET,IN.
GET,DATA.
RF,OFF.
TN(I=IN,L=X)
TTACH(FTNMLIB/UN=LIBRARY)
DSET(LID=FTNMLIB)
GO(DATA)

/-PLN2(IN=LN2STRS,DATA=LN2DAT)
 0. -315. 50242.5
 6.001273691528 -207. 33016.5
12.00254738306 -49. 7815.5
18.00382107458 22. -3509.
24.00509476611 53. -8453.5
30.00636845764 68. -10846.
36.00764214917 74. -11803.
42.00891584069 77. -12281.5
48.01018953222 78. -12441.
54.01146322375 78. -12441.
60.01273691528 79. -12600.5
66.0140106068 79. -12600.5
72.01528429833 79. -12600.5
78.01655798986 79. -12600.5
84.01783168139 79. -12600.5
90.01910537292 79. -12600.5
96.02037906444 79. -12600.5
102.021652756 79. -12600.5
108.0229264475 79. -12600.5
114.024200139 79. -12600.5
120.0254738306 79. -12600.5
126.0267475221 79. -12600.5
132.0280212136 79. -12600.5
138.0292949051 79. -12600.5
144.0305685967 79. -12600.5
150.0318422882 79. -12600.5
156.0331159797 79. -12600.5
162.0343896712 79. -12600.5
168.0356633628 79. -12600.5
174.0369370543 79. -12600.5
180.0382107458 79. -12600.5
186.0394844374 79. -12600.5
192.0407581289 79. -12600.5
198.0420318204 79. -12600.5
204.0433055119 79. -12600.5
210.0445792035 79. -12600.5
216.045852895 79. -12600.5
222.0471265865 79. -12600.5
228.0484002781 79. -12600.5
234.0496739696 79. -12600.5
240.0509476611 79. -12600.5
246.0522213526 79. -12600.5
252.0534950442 79. -12600.5
258.0547687357 79. -12600.5
264.0560424272 79. -12600.5
270.0573161187 79. -12600.5
276.0585898103 79. -12600.5
282.0598635018 79. -12600.5
288.0611371933 79. -12600.5
294.0624108849 79. -12600.5
300.0636845764 79. -12600.5
306.0649582679 79. -12600.5
312.0662319594 78. -12441.
318.067505651 78. -12441.
324.0687793425 77. -12281.5
330.070053034 74. -11803.
336.0713267256 68. -10846.
342.0726004171 53. -8453.5
348.0738741086 22. -3509.
354.0751478001 -49. 7815.5
360.0764214917 -207. 33016.5
 ILLEGAL CONTROL CARD.
/
```

completely constrained

(see note on next page)

```
 0. -317. 50561.5
 6.001273691528 -317. 50561.5
12.00254738306 -316. 50402.
18.00382107458 -207. 33016.5
24.00509476611 -49. 7815.5
30.00636845764 22. -3509.
36.00764214917 53. -8453.5
42.00891584069 68. -10846.
48.01018953222 74. -11803.
54.01146322375 77. -12281.5
60.01273691528 78. -12441.
66.0140106068 78. -12441.
72.01528429833 79. -12600.5
78.01655798986 79. -12600.5
84.01783168139 79. -12600.5
90.01910537292 79. -12600.5
96.02037906444 79. -12600.5
102.021652756 79. -12600.5
108.0229264475 79. -12600.5
114.024200139 79. -12600.5
120.0254738306 79. -12600.5
126.0267475221 79. -12600.5
132.0280212136 79. -12600.5
138.0292949051 79. -12600.5
144.0305685967 79. -12600.5
150.0318422882 79. -12600.5
156.0331159797 79. -12600.5
162.0343896712 79. -12600.5
168.0356633628 79. -12600.5
174.0369370543 79. -12600.5
180.0382107458 79. -12600.5
186.0394844374 79. -12600.5
192.0407581289 79. -12600.5
198.0420318204 79. -12600.5
204.0433055119 79. -12600.5
210.0445792035 79. -12600.5
216.045852895 79. -12600.5
222.0471265865 79. -12600.5
228.0484002781 79. -12600.5
234.0496739696 79. -12600.5
240.0509476611 79. -12600.5
246.0522213526 79. -12600.5
252.0534950442 79. -12600.5
258.0547687357 79. -12600.5
264.0560424272 79. -12600.5
270.0573161187 79. -12600.5
276.0585898103 79. -12600.5
282.0598635018 79.  12600.5
288.0611371933 79. -12600.5
294.0624108849 79. -12600.5
300.0636845764 78. -12441.
306.0649582679 78. -12441.
312.0662319594 77. -12281.5
318.067505651 74. -11803.
324.0687793425 68. -10846.
330.070053034 53. -8453.5
336.0713267256 22. -3509.
342.0726004171 -49. 7815.5
348.0738741086 -207. 33016.5
354.0751478001 -316. 50402.
360.0764214917 -317. 50561.5
 ILLEGAL CONTROL CARD.
/new
```

completely constrained

Note I put in final Temps as if initial Temp was 0°

∴ stresses should be modified by

$$\frac{100-T}{T} \times \sigma$$

$\theta = 0$ $\left[\frac{100-(-317)}{317}\right] 50562 = 66512$

$\theta = 180$ $\left(\frac{100-79}{79}\right)(-12600.5) = -$

15 Blocks

completely constrained

same as above

See N...

```
0. -317. 50561.5
6.001273691528 -317. 50561.5
12.00254733306 -317. 50561.5
18.00382107458 -317. 50561.5
24.00509476611 -317. 50561.5
30.00636845764 -317. 50561.5
36.00764214917 -317. 50561.5
42.00891584069 -317. 50561.5
48.01018953222 -317. 50561.5
54.01146322375 -317. 50561.5
60.01273691528 -317. 50561.5
66.0140106068 -317. 50561.5
72.01528429833 -317. 50561.5
78.01655798986 -317. 50561.5
84.01783168139 -316. 50402.
90.01910537292 -207. 33016.5
96.02037906444 -49. 7815.5
102.021652756 22. -3509.
108.022926475 53. -8453.5
114.024200139 68. -10846.
120.0254738306 74. -11803.
126.0267475221 77. -12281.5
132.0280212136 78. -12441.
138.0292949051 78. -12441.
144.0305685967 79. -12600.5
150.0318422882 79. -12600.5
156.0331159797 79. -12600.5
162.0343896712 79. -12600.5
168.0356633628 79. -12600.5
174.0369370543 79. -12600.5
180.0382107458 79. -12600.5
186.0394844374 79. -12600.5
192.0407581289 79. -12600.5
198.0420318204 79. -12600.5
204.0433055119 79. -12600.5
210.0445792035 79. -12600.5
216.045852895 79. -12600.5
222.0471265865 79. -12600.5
228.048[illegible]02781 78. -12441.
234.04[illegible]739696 78. -12441.
240.0509476611 77. -12281.5
246.0522213526 74. -11803.
252.0534950442 68. -10846.
258.0547687357 53. -8453.5
264.0560424272 22. -3509.
270.0573161187 -49. 7815.5
276.0585898103 -207. 33016.5
282.0598635018 -316. 50402.
288.061137193[illegible] -317. 50561.5
294.0624108849 -317. 50561.5
300.0636845764 -317. 50561.5
306.0649582679 -317. 50561.5
312.0662319594 -317. 50561.5
318.067505651 -317. 50561.5
324.0687793425 -317. 50561.5
330.070053034 -317. 50561.5
336.0713267256 -317. 50561.5
342.0726004171 -317. 50561.5
348.0738741086 -317. 50561.5
354.0751478001 -317. 50561.5
360.0764214917 -317. 50561.5
ILLEGAL CONTROL CARD.
```

BY ______ DATE ______ SUBJECT ______ SHEET NO. 59 OF ______
CHKD. BY ______ DATE ______ JOB NO. ______

The peak stresses are tensile stresses and they are proportional to the amount of end constraint on the cylinder. For the completely constrained cyl. the amount of exposed surface does not affect the peak stress, whereas for the other two cases the more exposed shell the lower the thermal stress. The boundary conditions that approximate the short leg best are the bending constraint only. This part of the tunnel is flexible in the axial direction. Therefore the peak tensile stress occurs with only a small exposed area and will have a maximum value of 60,000 psi. The compressive stress increases with increasing exposed area, For half of the shell exposed this stress is − 32,223 psi, need to check this for buckling. From ref 1.

$$\sigma_x)_{cr} = .606\,\gamma\,\frac{Et}{R} \qquad \gamma \equiv \text{Knock down factor}$$

$$\frac{R}{t} = \frac{96.66\,in}{.167\,in} = 182 \qquad \frac{L}{R} = \frac{25'}{8.33'} = 3.0$$

$$\gamma = .28$$

$$\sigma_x)_{cr} = (.606)(.28)(29\times10^6)\,.167/96.66 = 34,108\,psi$$

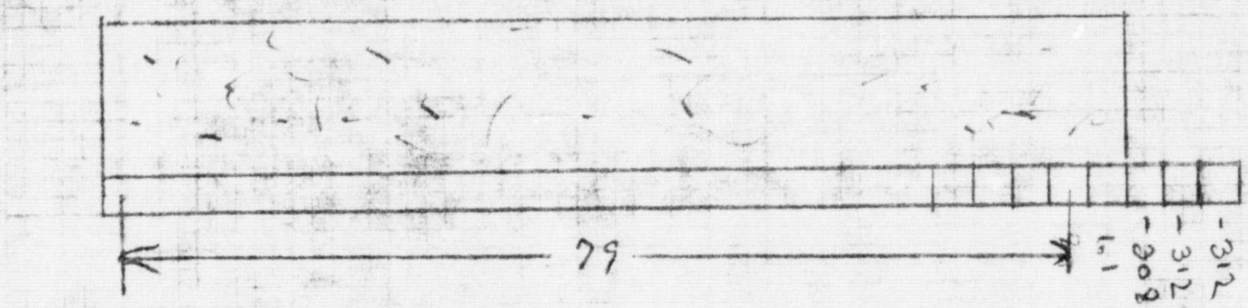
∴ for even half the shell exposed to LN_2 or GN_2 the compressive stress is less than critical.

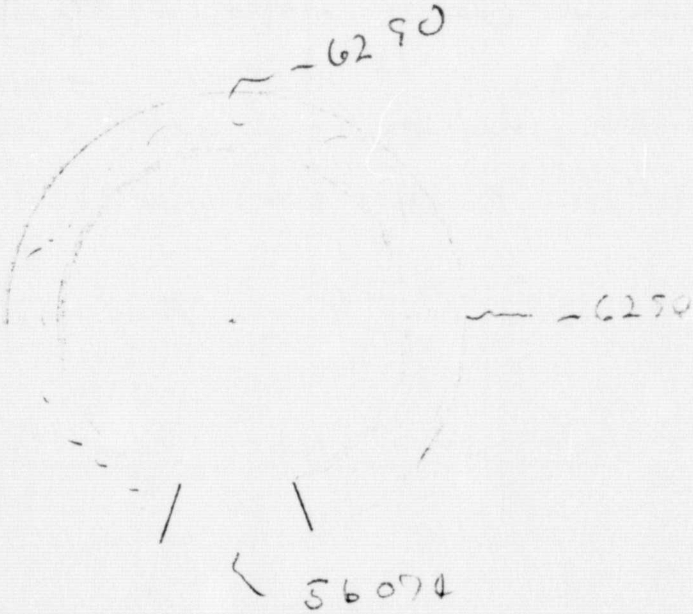
ORIGINAL PAGE IS
OF POOR QUALITY

BY ______ DATE ______ SUBJECT ______ SHEET NO. ______ OF ______
CHKD. BY ______ DATE ______ JOB NO. ______

TRANSIENT STRESSES FOR 3 BLOCKS

18" INSUL.





∴ steady state is the worse thermal stress

6" INSULATION

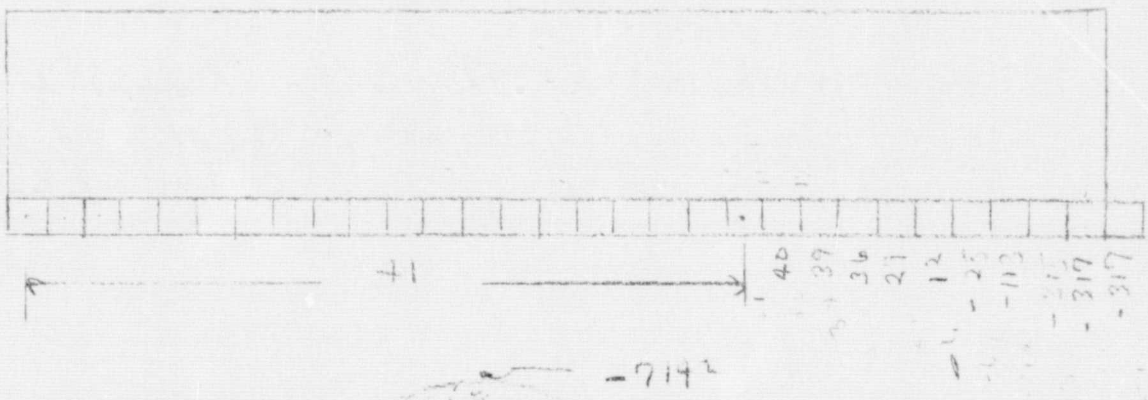
check to see if 6" insulation yields higher thermal stresses than 18".

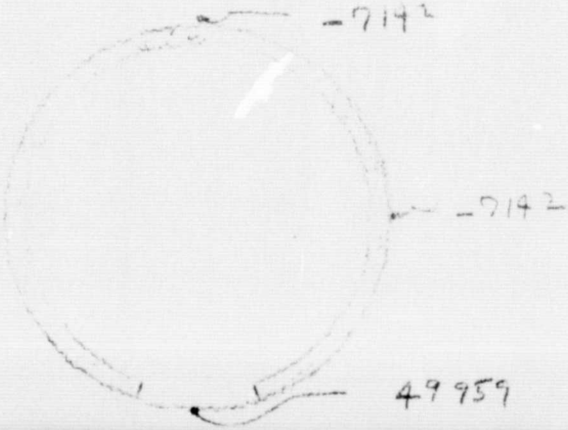
For insul shell $h_{eff} = \dfrac{1}{\dfrac{1}{1.389} + \dfrac{6 \times 144}{1.97}} = 1.7 \times 10^{-3}$

∴ For Insul Blks:—

$h_{eff} = [1.7 \times 10^{-3} + 1.5/144]/2 = 6.059 \times 10^{-3}$

$T_{eff} = \dfrac{1.7 \times 10^{-3}(140) + .01042)(530)}{2(.00545)} = 501\ °R$





∴ 18" INSULATION IS WORSE CASE

FINITE ELEMENT MODEL.

The closed form solution is not valid near the ends and also assumes that hoop stresses are small compared to axial stresses. A right circular cyl., 25' long, was modeled to check these two points plus allow for complex accident simulation and complex structural geometry (reinforcing rings). A completely constrained model was run with half the cyl. exposed to GN2 flow. The results in the center of the cyl. (away from ends) agreed excellently; However much higher axial (factor of 2) stresses and hoop stresses existed near the ends. Also, a restrained in bending only model was run. The stresses in the middle did not agree with closed form (they were lower) and stresses at fixed ends were much higher. Therefore, end conditions are significant and the finite element model should be used to predict fatigue life.

RESULTS OF SPAR FINITE ELEMENT

THE 1 BLOCK CASE WAS RUN IN SPAR COMPUTER RUN NO. "EDQ"

THE MAXIMUM BENDING STRESS AT JOINT 496 (CORNER LOCATION) IS 99,640 psi

THE MEMBRANE STRESS AT THIS LOCATION IS 54,860 psi

THE 3 BLOCK CASE IN SHOWN IN RUN "DFZ". THE 15 BLOCK CASE WAS RUN IN SPAR COMPUTER RUN NO. "ECK."

MAXIMUM BENDING STRESS AT JOINT 496 IS 127,110 psi

MEMBRANE STRESS IS 65,940 psi

THE MODEL AND RESULTS ARE SHOWN IN THE FOLLOWING PAGES. THE MAX. STRESSES OCCUR AT THE FIXED BOUNDARY CONDITIONS.

1 BLOCK @ -315°F
COMPUTER RUN CBZ

1/1/1

DISPLAY= SX /1000 , NODE= 4 , SURFACE= 0

0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16

0 23
SCALE

SPEC 4.1 TOP HALF OF CYLINDER
THERMO LOADS

PRECEDING PAGE BLANK NOT FILMED

1/1/1

DISPLAY= SX /1000 , NODE= 4, SURFACE= 1

0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22

SPEC 4.1 TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

1/1/1

DISPLAY= SX /1000 , NODE= 4 , SURFACE= 2

0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11

SPEC 4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 2.
SCALE

1 BLK
4 OF 17

DISPLAY= SX /1000 , NODE= 4 , SURFACE= 0

0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	1	-2	-15
0	0	0	0	0	0	0	0	0	0	0	1	-1	-12
0	0	0	0	0	0	0	0	0	0	0	0	0	-8
0	0	0	0	0	0	0	0	0	0	0	-1	2	3
0	0	0	0	0	0	0	0	0	0	0	-3	3	26
0	0	0	0	0	0	0	0	0	0	0	-4	3	54

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 SCALE

1/1/1

DISPLAY= SX /1000 , NODE= 4 , SURFACE= 1

0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
1	1	1	1	1	0	0	0	0	0	0	2	1	-22
1	1	1	1	1	1	1	1	1	0	0	2	2	-21
1	1	1	1	1	1	1	1	1	1	0	2	2	-20
0	0	0	0	1	1	1	1	1	1	1	2	3	-18
0	0	0	0	0	0	0	1	1	1	1	1	5	-13
0	0	0	0	0	0	0	0	0	0	1	-2	8	0
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-6	6	23
-1	-1	-1	-1	-1	-1	-1	-1	-2	-2	-1	-8	-4	65

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0
SCALE

)

1/1/1

DISPLAY= SX /1000 , NODE= 4, SURFACE= 2

0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
-1	-1	-1	-1	-1	-1	0	0	0	0	0	2	-6	-10
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	1	-5	-10
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	1	-5	-9
0	0	0	0	-1	-1	-1	-1	-1	-1	-1	0	-6	-7
0	0	0	0	0	0	0	-1	-1	-1	-2	0	-6	-2
0	0	0	0	0	0	0	0	0	0	-1	0	-4	6
1	1	1	1	1	1	1	1	1	1	0	1	1	24
1	1	1	1	1	1	1	1	2	2	2	1	10	43

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

1/1/1

DISPLAY= SY /1000 , NODE= 4, SURFACE= 0

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13

SPEC 4.1

TOP HALF OF CYLINDER
THERMO LOADS

0
SCALE

1/1/1

DISPLAY= SY /1000 , NODE= 4 , SURFACE= 1

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32

PEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

1/1/1

DISPLAY= SY /1000 , NODE= 4, SURFACE= 2

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6

SPEC 4.1 TOP HALF OF CYLINDER
THERMO LOADS

0 SCALE

1/1/1 .

DISPLAY= SY /1000 , NODE= 4, SURFACE= 0

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-12	-12	-12	-12	-12	-12	-12	-12	-12
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12
-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-10
-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-8	-7
-3	-3	-3	-3	-3	-3	-3	-4	-4	-4	-4	-4	-4	-2
8	8	8	8	8	8	8	8	8	8	8	8	7	10
33	33	33	33	33	33	33	33	33	33	33	33	33	34

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 ——— 2?
SCALE

1 BLK

)

1/1/1

DISPLAY= SY /1000 , NODE= 4, SURFACE= 1

.3	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-13	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-13	-3	-32
-12	-12	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-13	-12	-3	-31
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-13	-12	-3	-31
-12	-12	-12	-12	-12	-12	-12	-12	-12	-11	-12	-11	-3	-29
-11	-11	-11	-11	-11	-11	-11	-11	-11	-10	-11	-10	-2	-26
-9	-9	-9	-9	-9	-9	-9	-9	-9	-8	-8	-8	-2	-17
-4	-4	-4	-4	-4	-4	-4	-4	-4	-4	-3	-5	-2	-1
8	8	8	8	8	8	8	8	8	7	9	5	-1	30
33	33	33	33	33	33	33	33	33	33	35	31	11	75

0 SCALE

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

1 BLK

12 OF 17

1/1/1

DISPLAY= SY /1000 , NODE= 4, SURFACE= 2

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-22	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-22	6
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-11	-12	-21	6
-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-19	5
-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-15	3
-3	-3	-3	-3	-3	-3	-3	-3	-3	-4	-4	-3	-6	-3
8	8	8	8	8	8	8	8	8	8	7	10	16	-9
34	34	34	34	34	34	34	34	34	34	32	36	55	-7

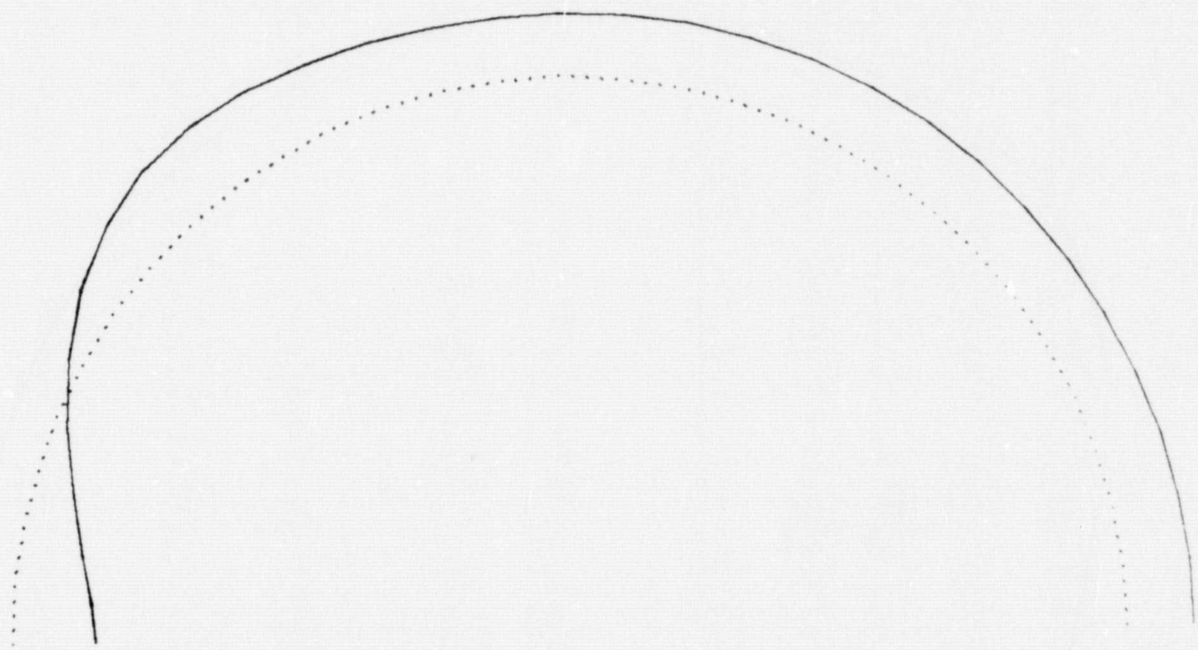
SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 2
SCALE

1 BLK

13 OF 17

1/1/1



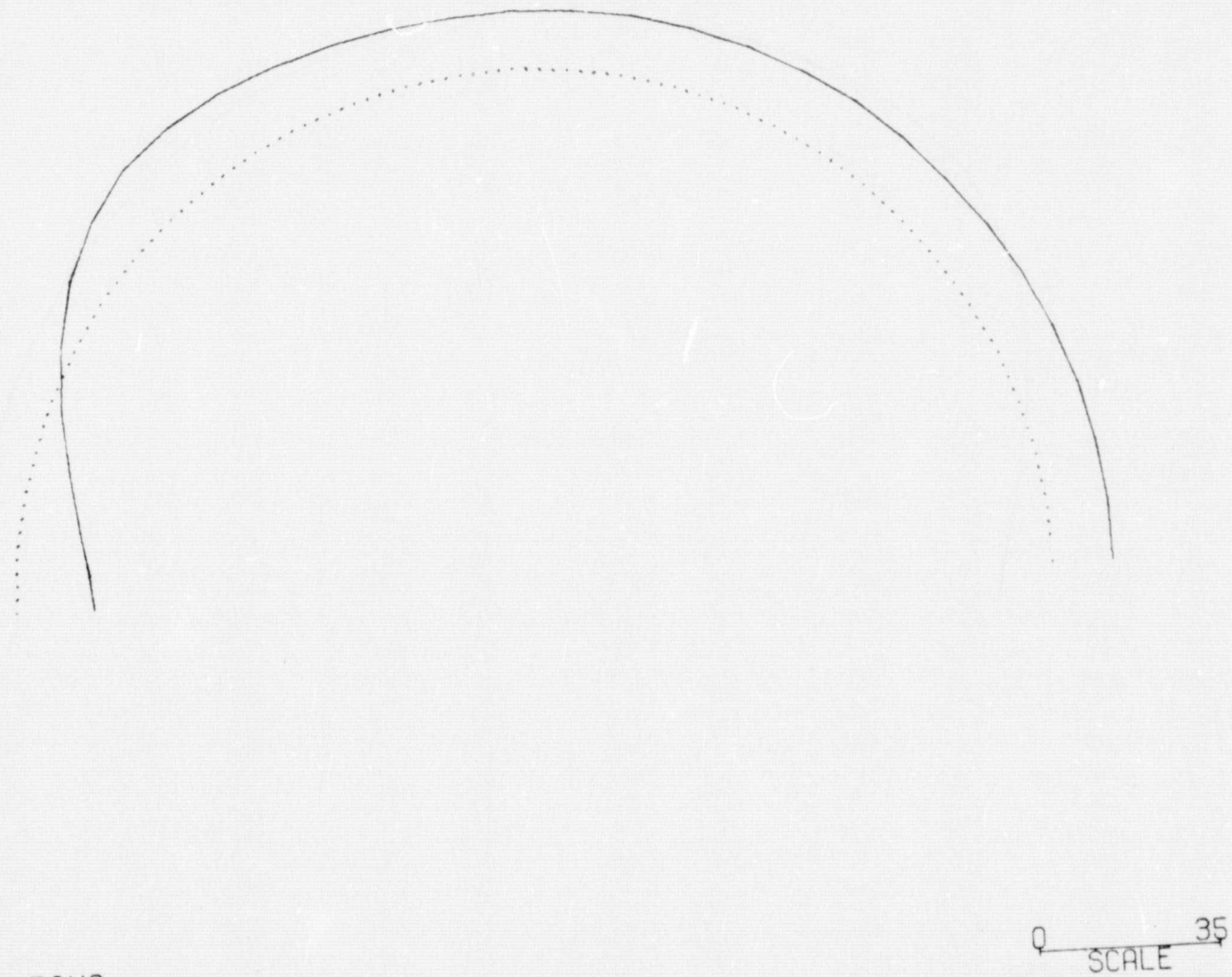
SPEC
2.1

RING

0 SCALE

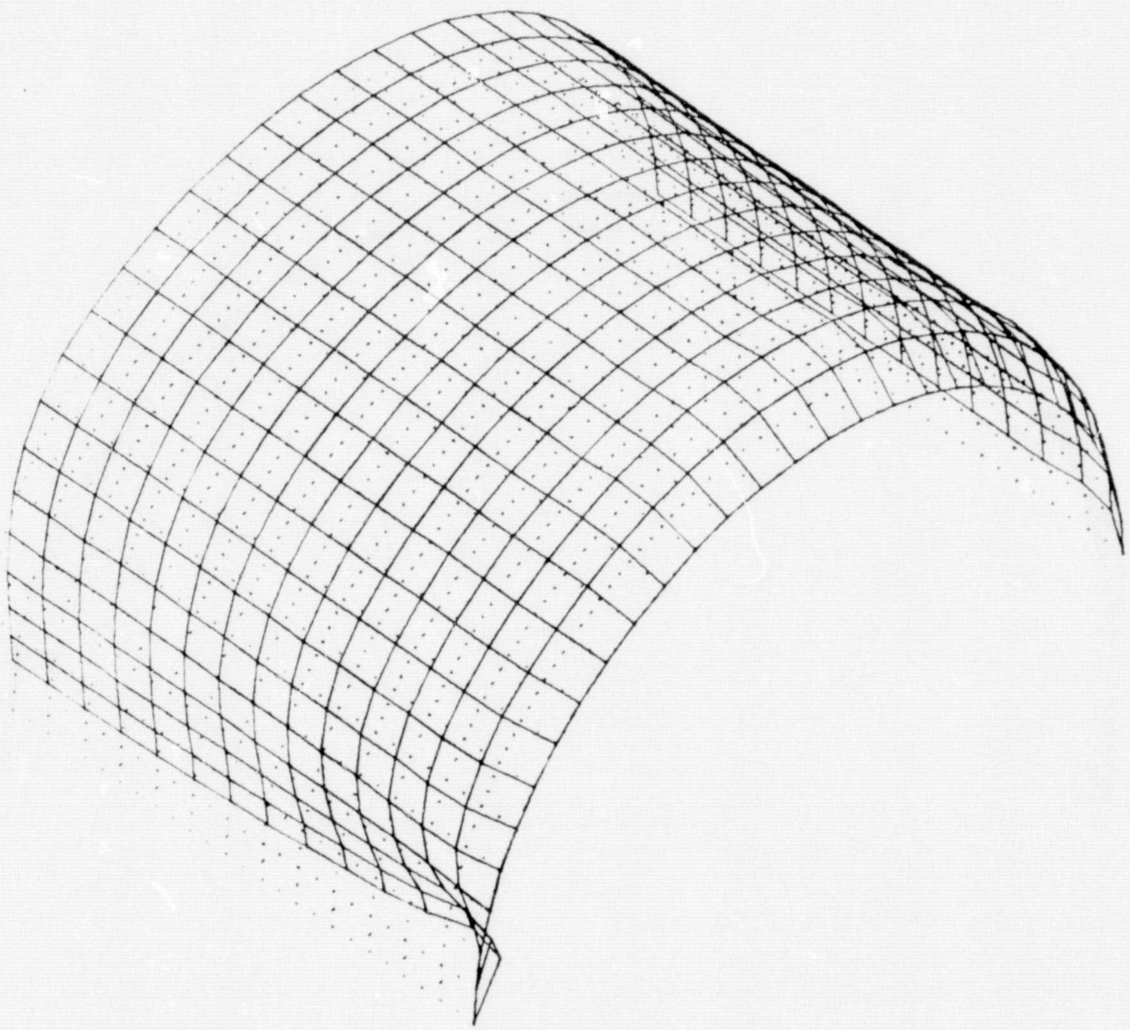
1 BLK
14 OF 17

1/1/1



EC RING
1

1/1/1



PEC ALL
.1

0 42
SCALE

1 BLK
16 OF 17

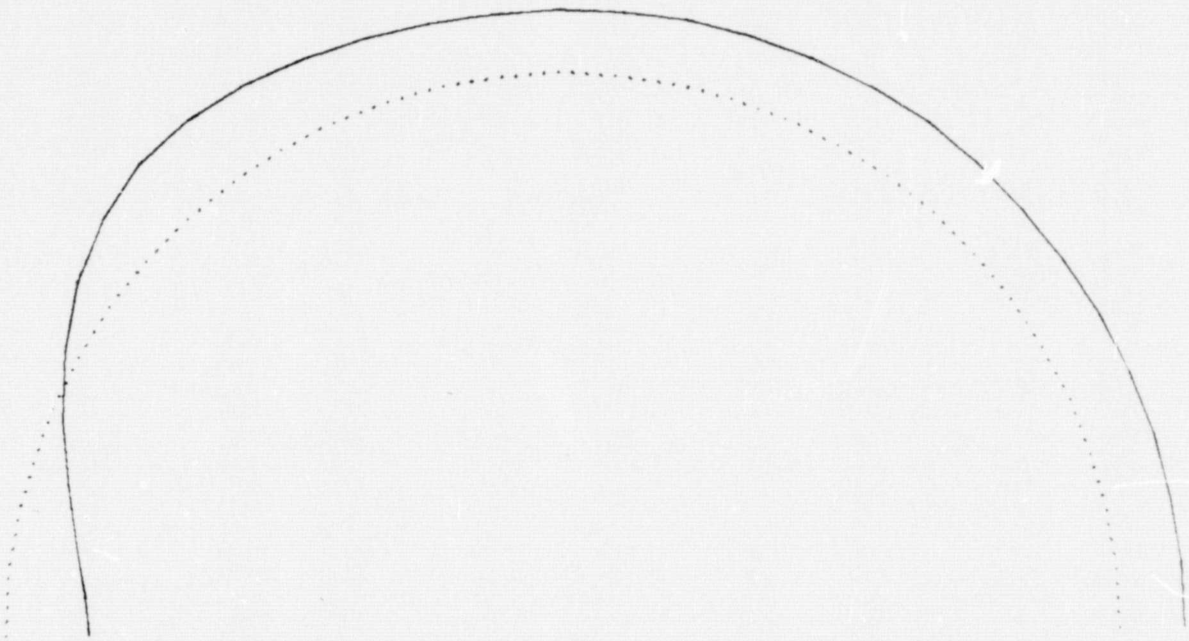
1/1/1

0 35
SCALE

SPEC 7.1

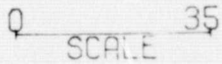
ALL

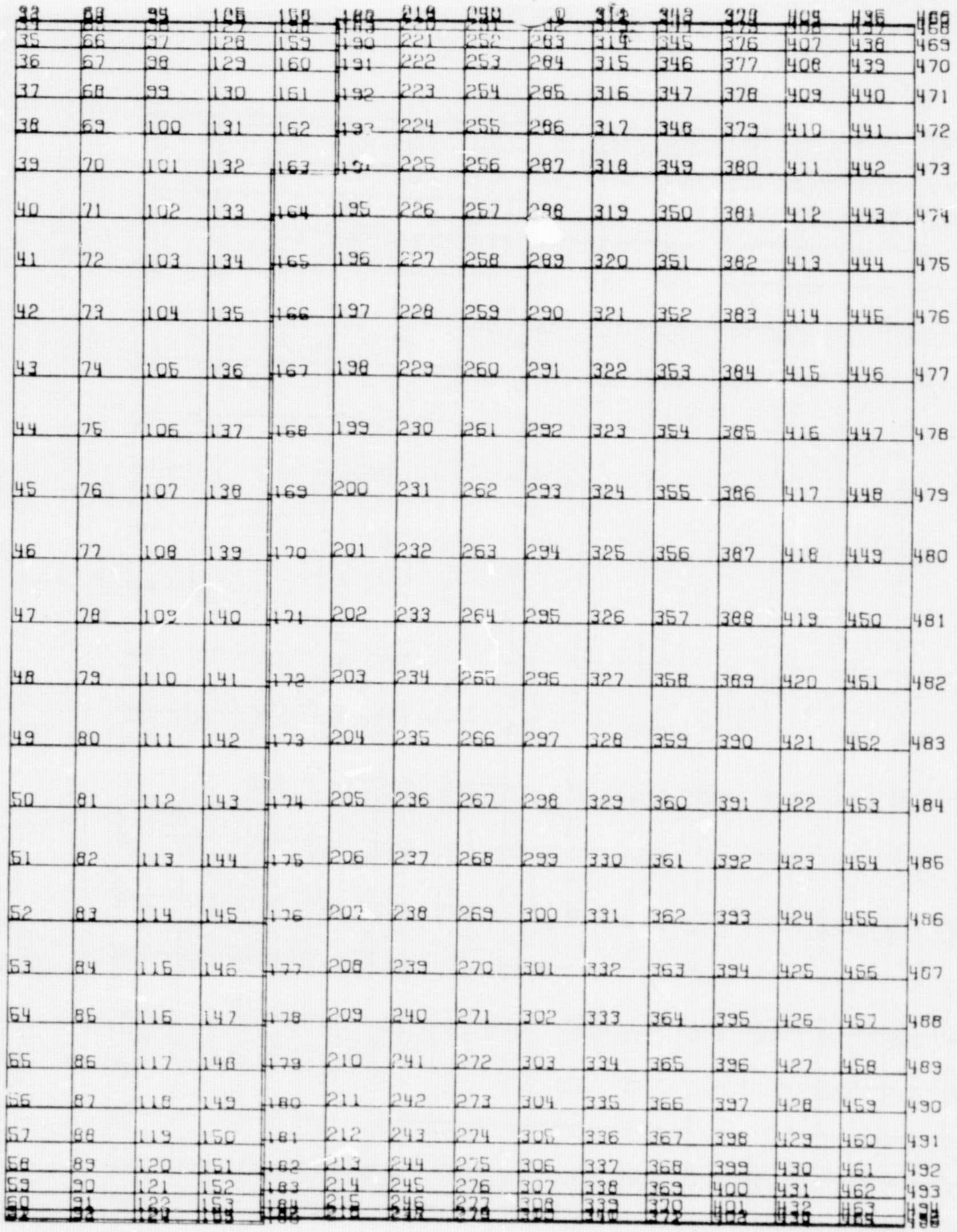
1/1.1



SPEC
2.1

RING





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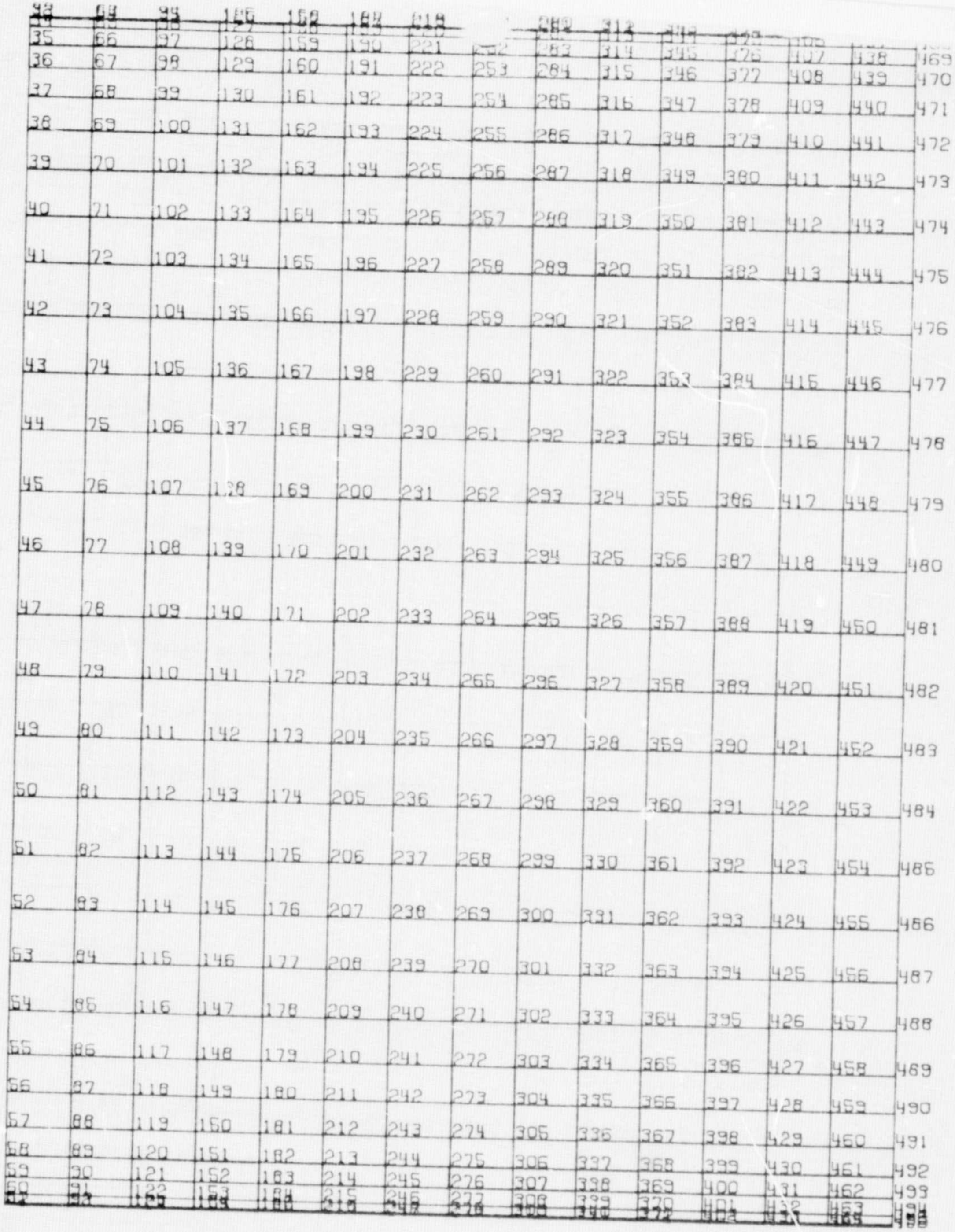
3 BLK CASE
RUN "DF2"
1 OF 21

SPEC 1.1 SHELL AND RING ....ALL....

0 30
SCALE

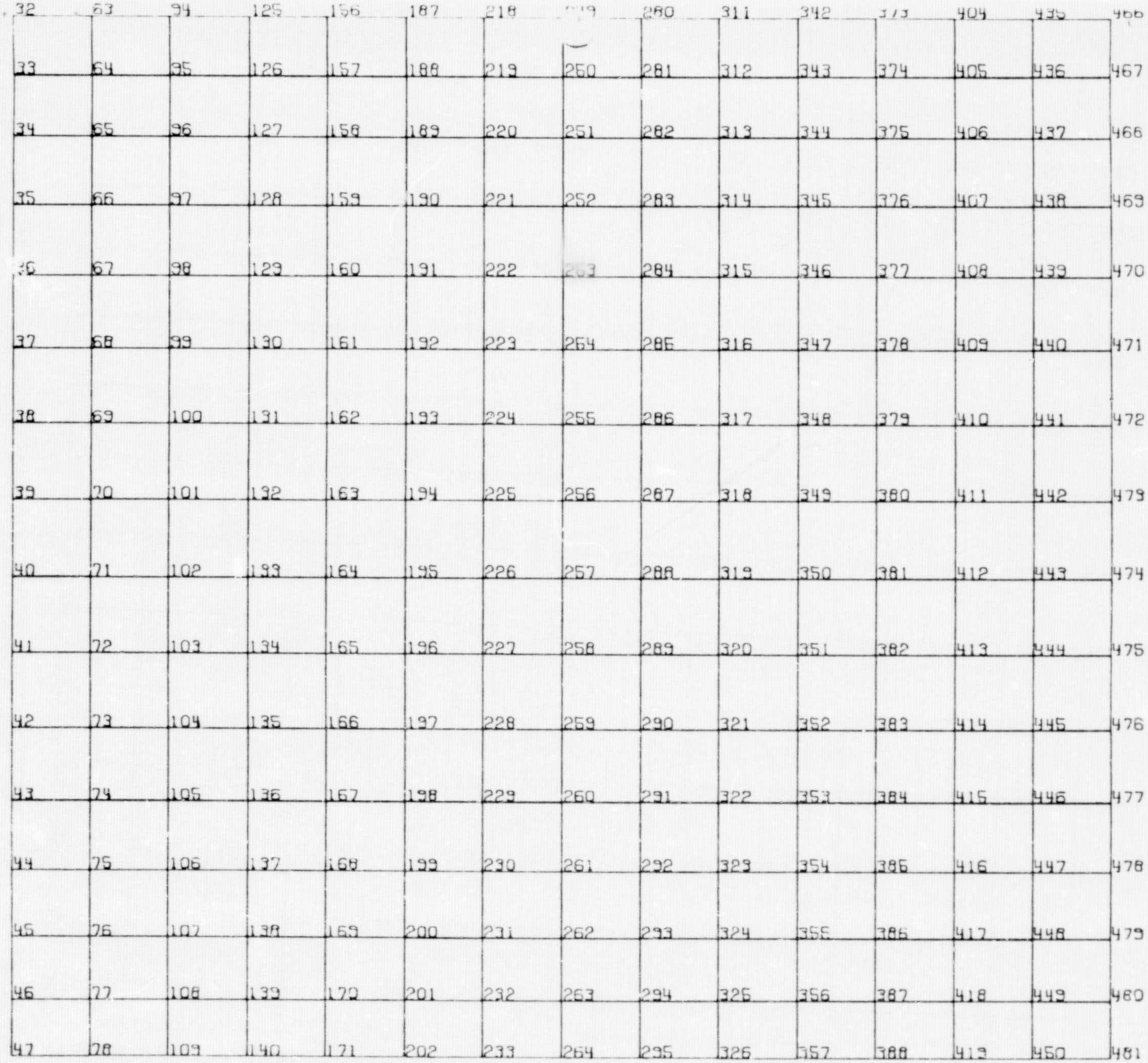


SPEC
3.1

SHELL

0 30
SCALE

3 BLK
2 OF 21



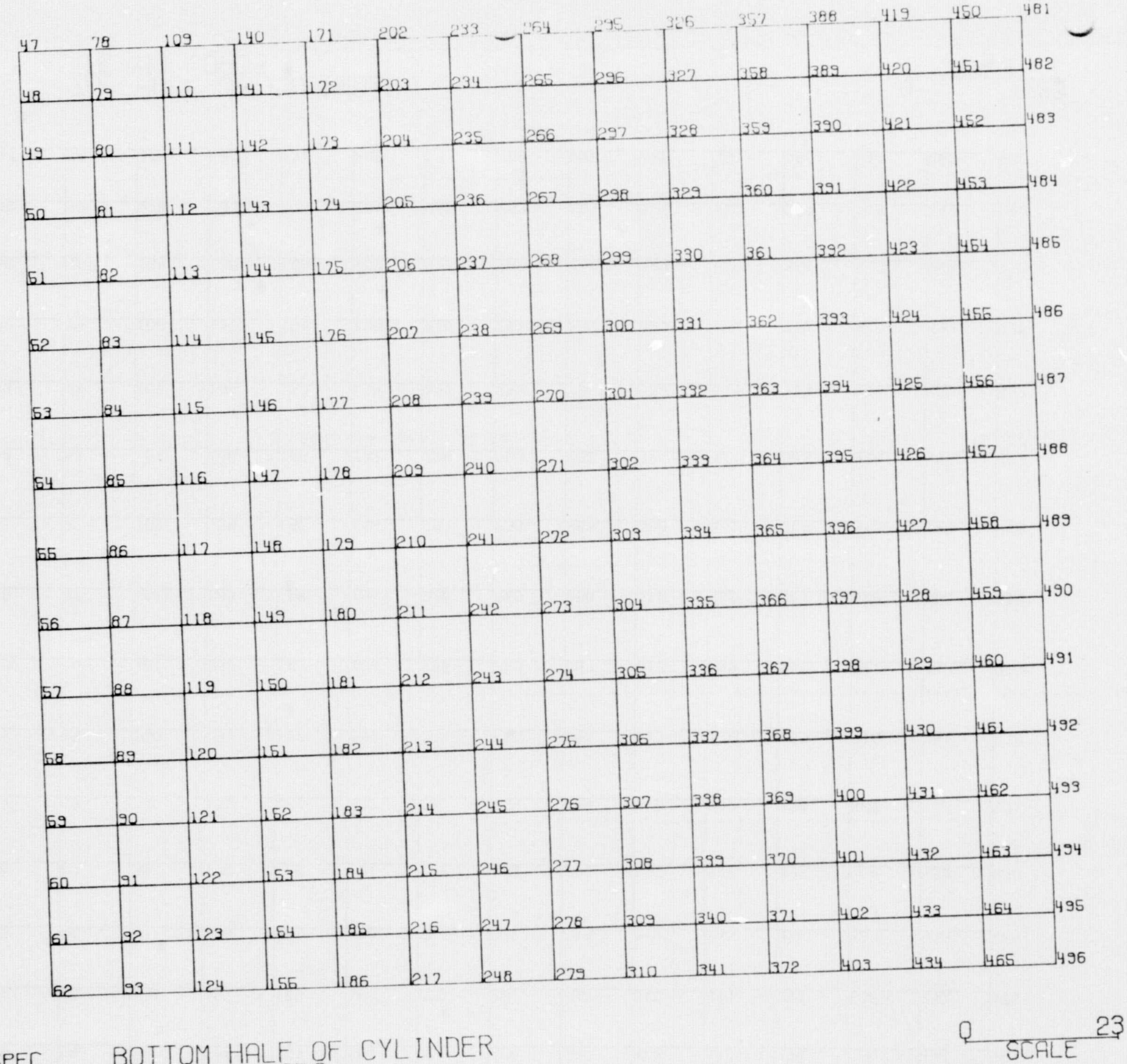
SPEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

3 BLK
3 OF 21

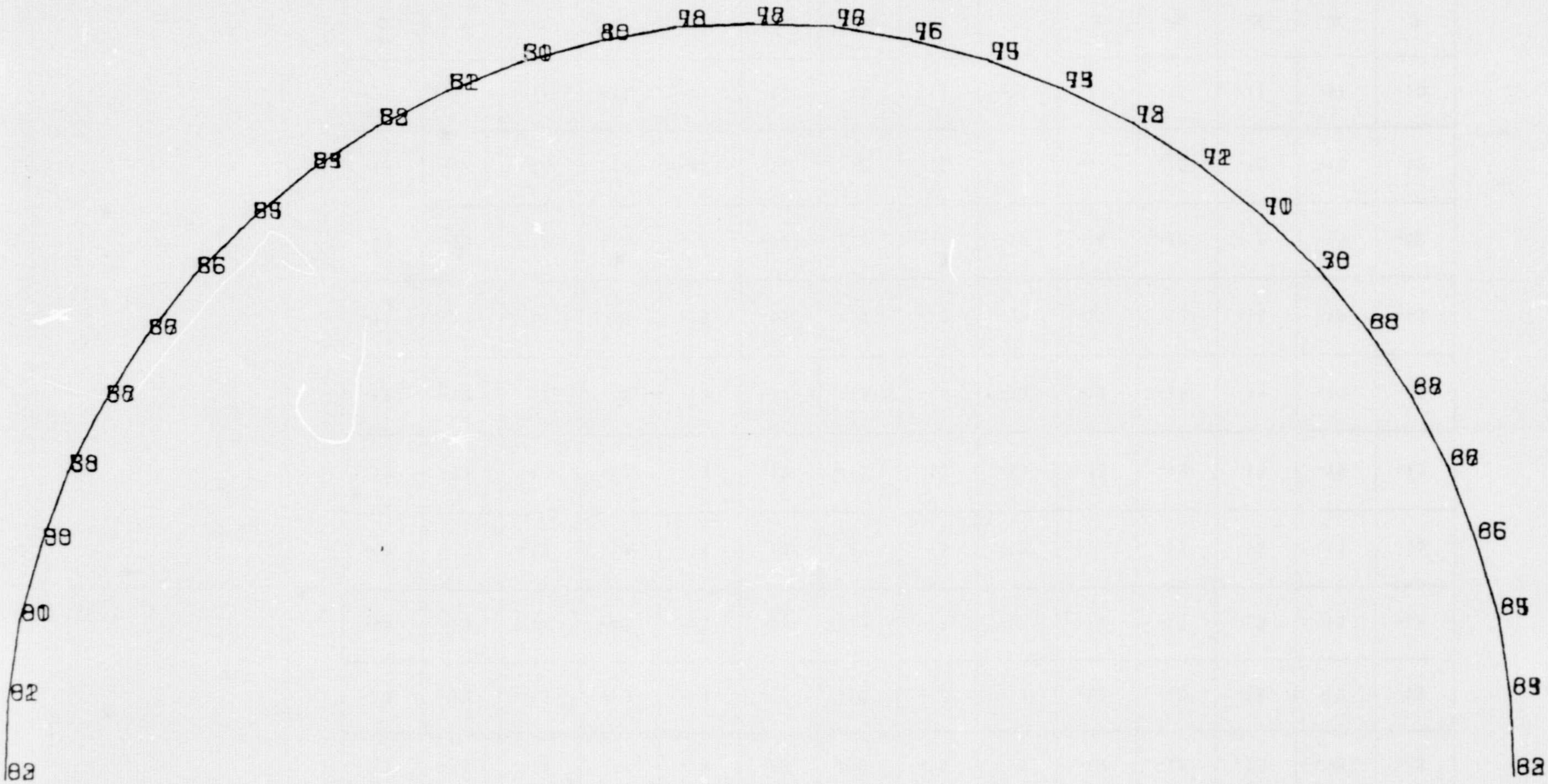
C-2



SPEC
5.1

BOTTOM HALF OF CYLINDER
THERMO LOADS

3 BLK
4 OF 21



3 BLK
5 OF 21

DISPLAY= SY /1000 . NODE= 4 . SURFACE= 0

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-13	-12
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12
-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-10
-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-8	-7
-3	-3	-3	-3	-3	-3	-4	-4	-4	-4	-4	-4	-4	-2
8	8	8	8	8	8	8	8	8	8	8	8	8	10

SPEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 SCALE 23

3 BLK
6 OF 21

1/1/1

DISPLAY= SX /1000 . NODE= 4. SURFACE= 1

0	0	0	0	0	0	0	0	-1	0	0	-8	1	64
0	0	0	-1	-1	-1	-1	-1	-1	-1	0	-10	-4	83
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-10	-5	87
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-9	-6	88
-1	-1	-1	-1	-1	-1	-1	-1	-1	0	1	-8	-4	89
-1	-1	-1	-1	-1	-1	-1	0	-1	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	1	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89

SPEC BOTTOM HALF OF CYLINDER·

0 23
SCALE

3 BLK
7 OF 21

DISPLAY= SX /1000 . NODE= 4. SURFACE= 2

0	0	0	0	0	0	0	0	0	1	0	-1	8	45
0	0	1	1	1	1	1	1	1	2	1	-2	16	49
1	1	1	1	1	1	1	1	1	2	1	-4	20	45
1	1	1	1	1	1	1	1	1	2	0	-5	20	43
1	1	1	1	1	1	1	1	1	1	0	-6	20	43
1	1	1	1	1	1	1	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43
0	0	0	0	0	0	0	0	0	1	-1	-6	20	43

SPEC 5.1

BOTTOM HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

3 BLK
8 OF 21

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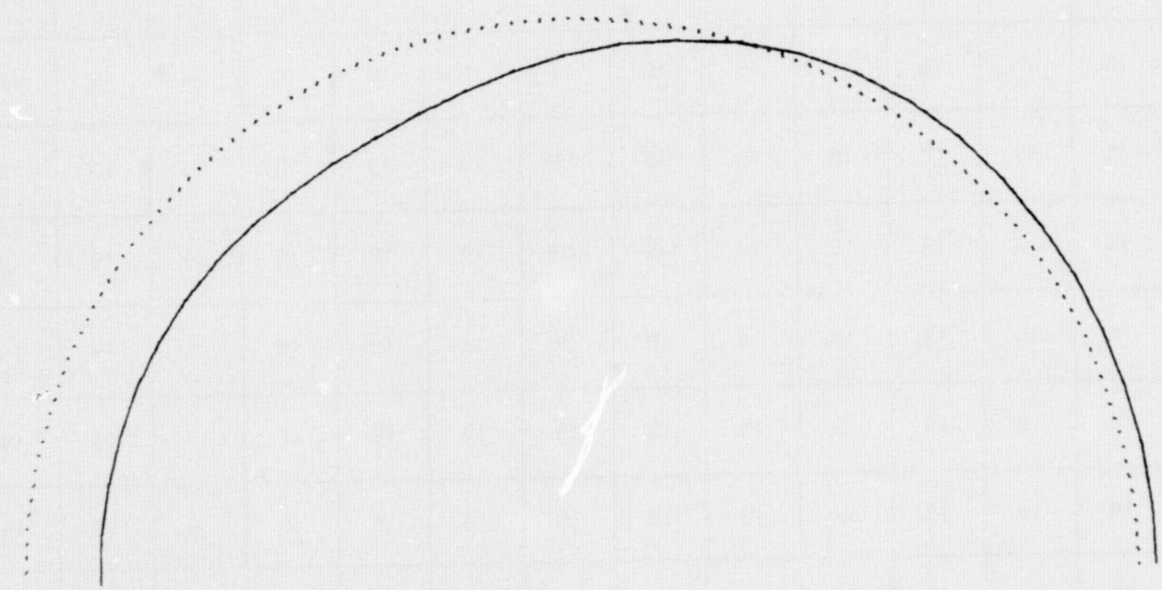

0 35
SCALE

SPEC 7.1

3 BLK
9 of 21

1/1/1

DISPLAY= SY /1000 . NODE= 4. SURFACE= 0

33	33	33	33	33	33	33	33	33	33	33	33	34	35
61	61	61	61	61	61	61	61	61	61	61	61	62	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61
61	61	61	61	61	61	61	61	61	61	61	61	61	61

SPEC 5.1 BOTTOM HALF OF CYLINDER THERMO LOADS

0 23 SCALE

1 BLK
1D OF 21

DISPLAY= SY /1000 . NODE= 4. SURFACE= 1

33	33	33	33	33	33	33	33	33	33	35	31	12	78
61	61	61	61	61	61	61	61	61	61	64	49	18	114
51	51	51	51	51	51	51	51	61	51	64	49	13	123
61	61	61	61	61	61	61	61	61	61	64	49	11	127
61	61	61	61	61	61	61	61	61	61	66	60	11	128
61	61	61	61	61	61	61	61	61	61	65	60	12	128
61	61	61	61	61	61	61	61	61	61	65	60	12	127
61	61	61	61	61	61	61	61	61	61	66	60	12	127
61	61	61	61	61	61	61	61	61	61	66	60	12	127
61	61	61	61	61	61	61	61	61	61	66	60	12	127
61	61	51	51	61	61	61	61	61	61	65	60	12	127
61	61	51	61	51	61	51	61	61	61	65	60	12	127
61	51	61	61	61	61	61	61	61	61	66	60	12	127
51	51	51	51	51	61	61	51	51	51	65	60	12	127
61	61	61	61	61	61	61	61	61	61	66	60	12	127

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SPEC 5.1 BOTTOM HALF OF CYLINDER THERMO LOADS

0 23 SCALE

3 BLK

11 OF 21

DISPLAY= SY /1000 . NODE= 4. SURFACE= 2

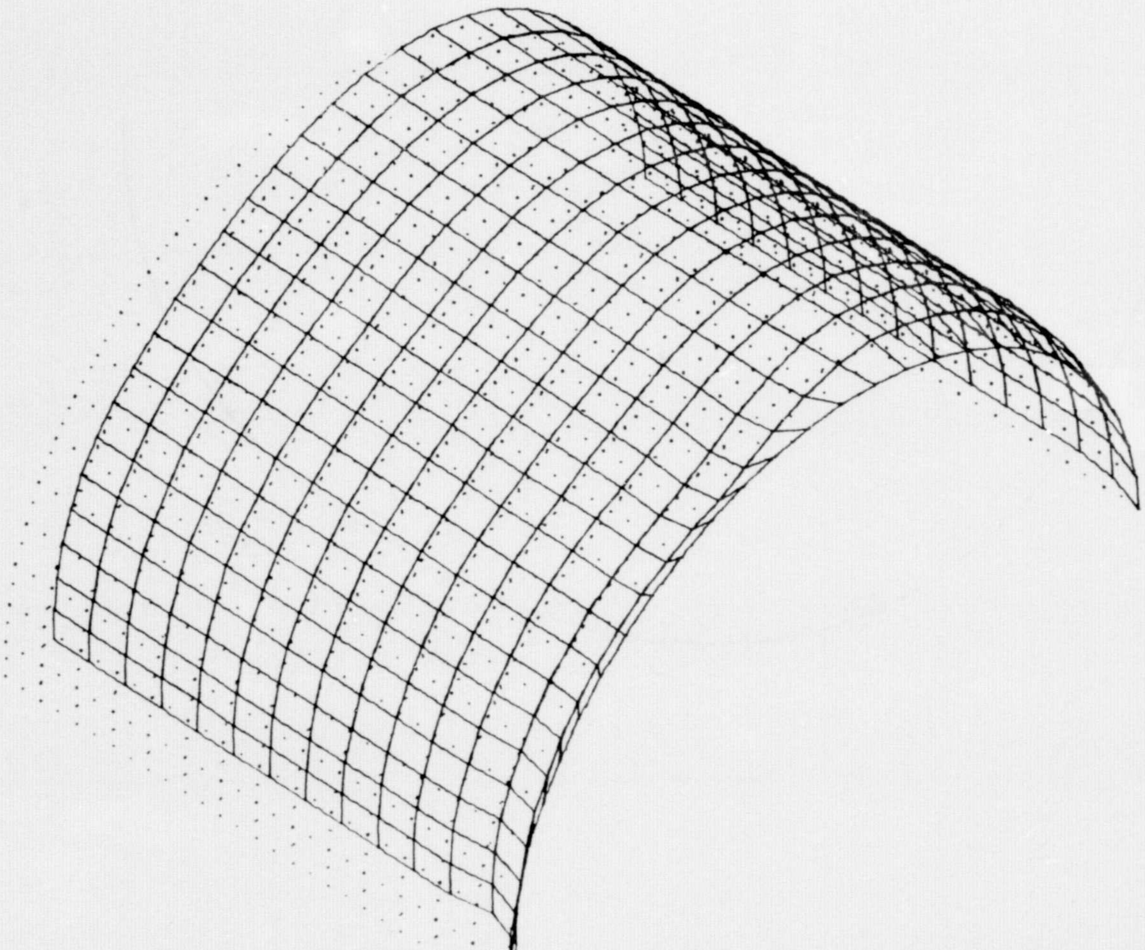
33	33	33	33	33	34	34	34	34	34	31	35	66	-8
61	61	61	61	61	61	61	61	62	62	49	63	86	-11
62	62	62	62	62	52	62	62	62	62	48	53	90	-22
62	62	62	62	62	62	62	62	62	62	48	63	91	-25
62	62	62	62	62	62	62	61	62	62	48	53	91	-26
62	62	62	61	61	61	61	61	62	62	48	53	91	-26
61	61	61	61	61	61	61	61	62	62	48	63	91	-26
61	61	61	61	61	61	61	61	62	62	48	63	91	-24
61	61	61	61	61	61	61	61	62	62	48	63	91	-26
61	61	61	61	61	61	61	61	62	62	48	63	91	-26
61	61	61	61	61	61	61	61	62	62	48	63	91	-25
61	61	61	61	61	61	61	61	62	62	48	63	91	-26
61	61	61	61	61	61	61	61	62	62	48	63	91	-26
51	61	51	51	61	51	61	61	62	62	48	53	91	-25
61	61	61	61	61	61	61	61	62	62	48	63	91	-26

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

3 BLK
12 OF 21

1/1/1



3 BLK
13 OF 21

0 42

SPEC 2.1 RING

0 35
SCALE

3 BLK
14 OF 21

DISPLAY= SX /1000 , NODE= 4 , SURFACE= 2 1/1/1

0	0	0	0	0	0	0	0	0	0	0	2	-6	-11
0	0	0	0	0	0	0	0	0	0	0	2	-6	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-6	-11
0	0	0	0	0	0	0	0	0	0	0	2	-6	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-6	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
-1	-1	0	0	0	0	0	0	0	0	0	2	-5	-10
-1	-1	-1	-1	-1	-1	-1	-1	0	-1	0	1	-6	-10
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	1	-5	-9
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-6	-7
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	-2	0	-6	-2
0	0	0	0	0	-1	-1	-1	-1	-1	-2	-1	-4	6
0	0	0	0	0	0	0	0	0	0	-1	-1	0	24

SPEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

3 BLK
15 OF 21

DISPLAY= SY /1000 , NODE= 4. SURFACE= 1

1/1/1

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-13	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-13	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-12	-12	-12	-12	-12	-12	-12	-13	-12	-12	-14	-12	-3	-32
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-13	-12	-3	-31
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-11	-3	-29
-11	-11	-11	-11	-11	-11	-11	-11	-11	-10	-11	-10	-2	-26
-9	-9	-9	-9	-8	-8	-8	-8	-8	-8	-8	-8	-2	18
-3	-3	-3	-3	-3	-3	-3	-3	-3	-3	-3	-4	-2	-1
8	8	8	8	8	8	8	8	8	8	9	6	0	30

SPEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

3 BLK
16 OF 21

1/1/1

DISPLAY= SX /1000 . NODE= 47 SURFACE= 1

0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
1	0	0	0	0	0	0	0	0	0	0	2	1	-22
1	1	1	1	1	1	1	1	1	0	0	2	1	-21
1	1	1	1	1	1	1	1	1	1	0	2	2	-20
1	1	1	1	1	1	1	1	1	1	1	2	3	-18
1	1	1	1	1	1	1	1	1	1	2	1	6	-13
0	0	0	0	0	0	1	1	1	1	2	-2	8	0
0	0	0	0	0	0	0	0	0	1	1	-6	7	29

SPEC 4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 SCALE 23

3 BLK

17 of 21

1/1/1

DISPLAY= SX /1000 . NODE= 4 . SURFACE= 0

0	0	0	0	0	0	0	0	0	0	0	-5	4	64
0	0	0	0	0	0	0	0	0	1	0	-6	6	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66
0	0	0	0	0	0	0	0	0	1	0	-7	8	66

SPEC 5.1

BOTTOM HALF OF CYLINDER
THERMO LOADS

0 ____ 23
SCALE

3 BLK
18 OF 21

DISPLAY= SX /1000 . NODE= 9. SURFACE= 1

0	0	0	0	0	0	0	0	-1	0	0	-8	1	64
0	0	0	-1	-1	-1	-1	-1	-1	-1	0	-10	-4	83
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-10	-5	87
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-9	-5	88
-1	-1	-1	-1	-1	-1	-1	-1	-1	0	1	-8	-4	89
-1	-1	-1	-1	-1	-1	-1	0	-1	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	1	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89
0	0	0	0	0	0	0	0	0	0	1	-7	-4	89

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

3 BLK
19 OF 21

DISPLAY= SX /1000 . NODE= 4. SURFACE= 0

0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	1	-2	-16
0	0	0	0	0	0	0	0	0	0	0	1	-1	-12
0	0	0	0	0	0	0	0	0	0	0	0	0	-8
0	0	0	0	0	0	0	0	0	0	0	-1	2	3
0	0	0	0	0	0	0	0	0	0	0	-3	4	27

0 ——— 23
SCALE

SPEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

3 BLK
20 of 21

DISPLAY= SY /1000 , NODE= 4 SURFACE= 2

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-22	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-22	6
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-11	-12	-21	6
-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-19	6
-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-16	3
-4	-4	-4	-4	-4	-4	-4	-4	-4	-4	-4	-3	-5	-2
8	8	8	8	8	8	8	8	8	8	7	9	16	-9

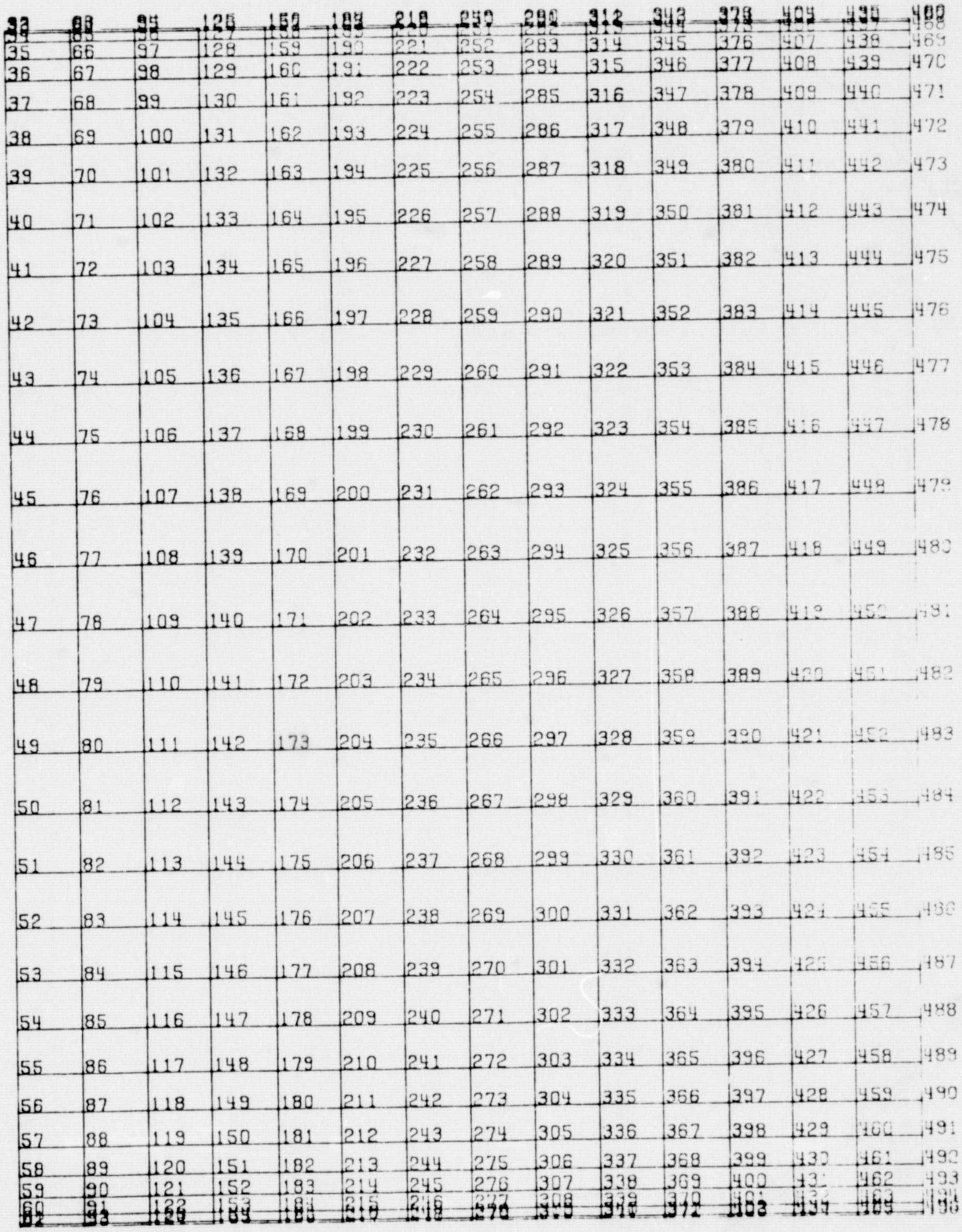
SPEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

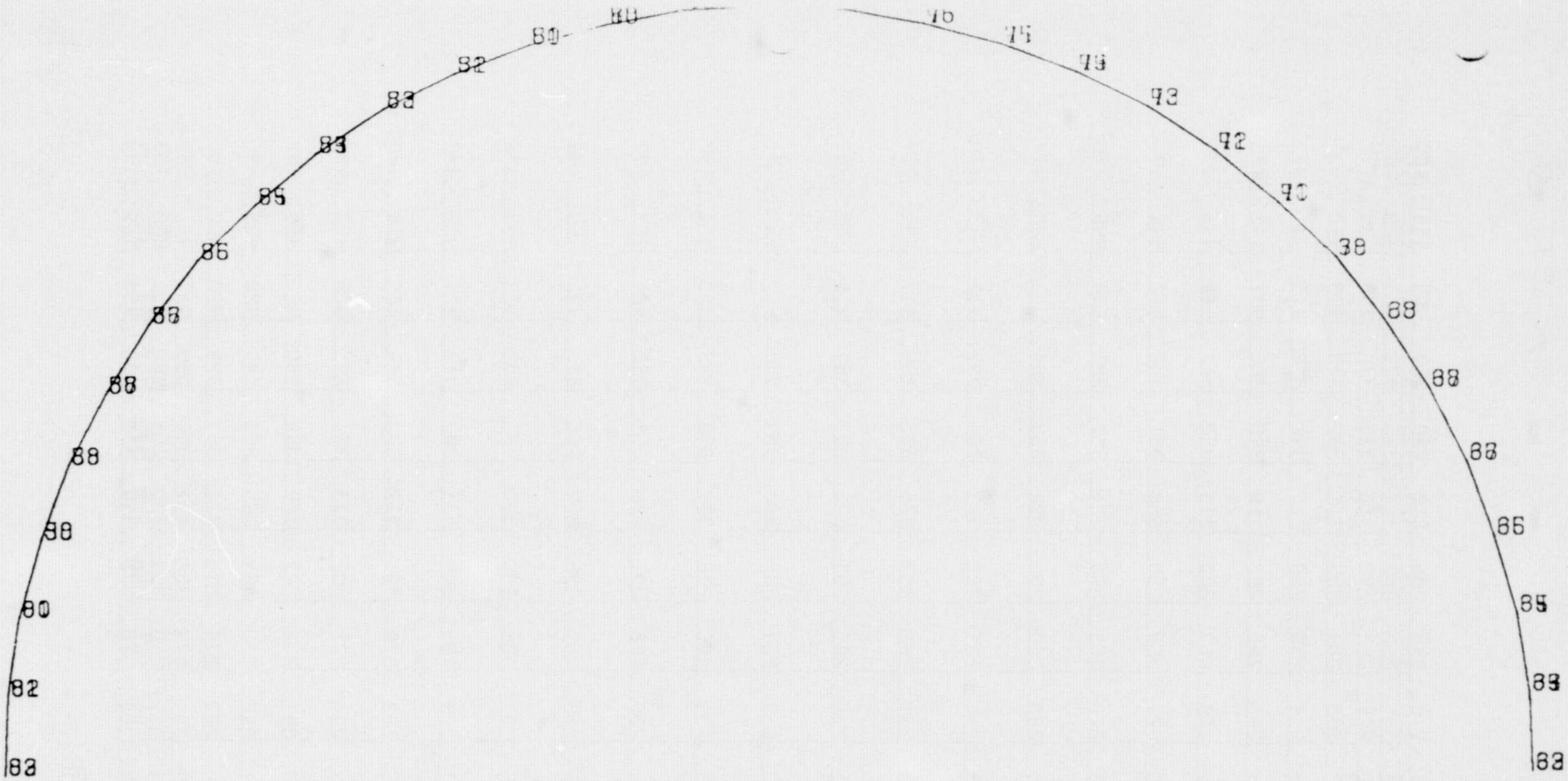
3 BLK
21 OF 21

15 BLK
- RUN "ECK"
1 OF 18



SHELL AND RING ....ALL....

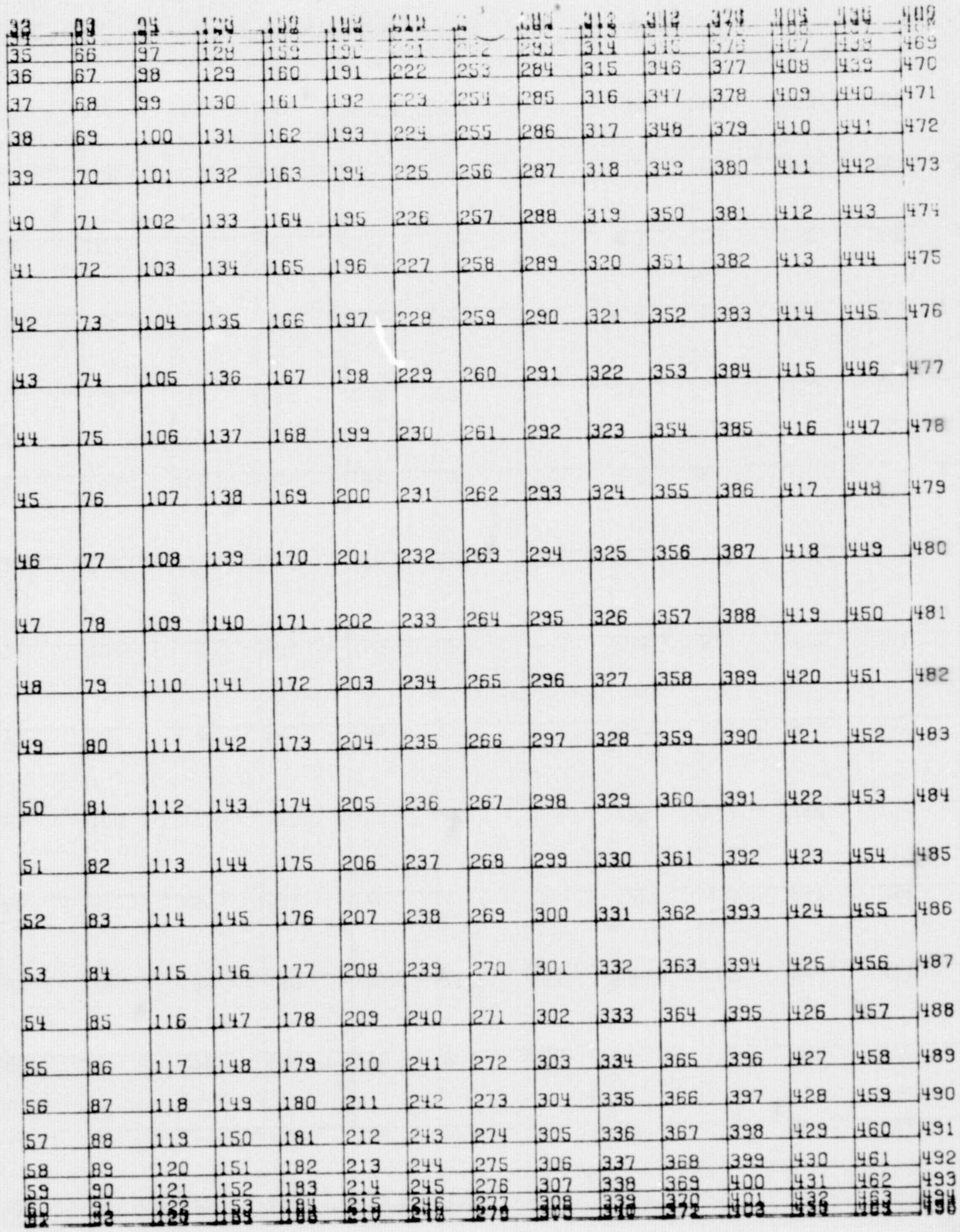
SCALE



SPEC RING
2.1

0 30
SCALE

15 BLK
2 of 18

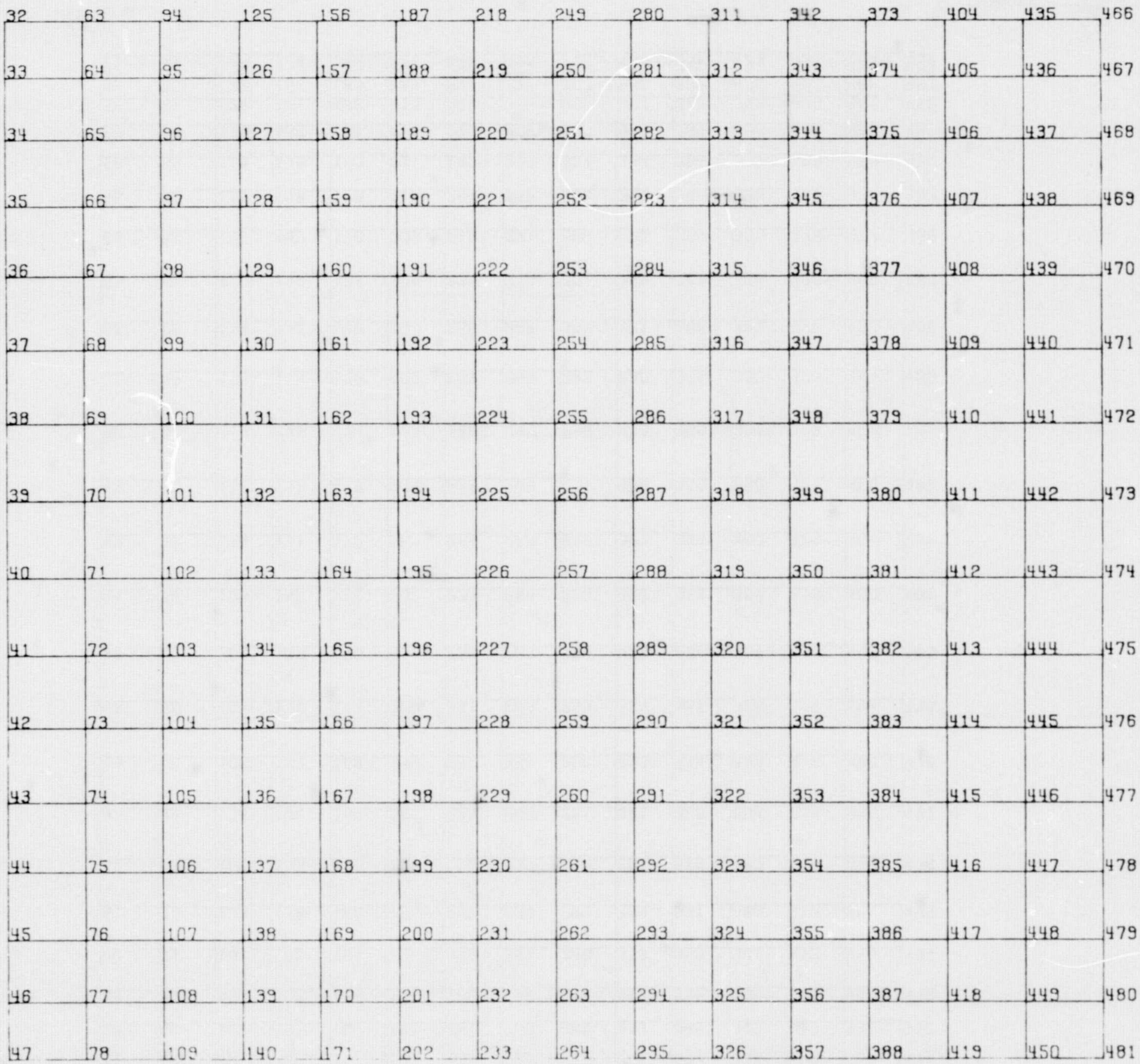


0 30
SCALE

SPEC
3.1

SHELL

15 BLK
3 OF 18



SPEC 4.1

TOP HALF OF CYLINDER
THERMO LOADS

Q .23
SCALE

15" BLK
4 OF 18

15 BLK
5 OF 13

47	78	109	140	171	202	233	264	295	326	357	388	419	450	481
48	79	110	141	172	203	234	265	296	327	358	389	420	451	482
49	80	111	142	173	204	235	266	297	328	359	390	421	452	483
50	81	112	143	174	205	236	267	298	329	360	391	422	453	484
51	82	113	144	175	206	237	268	299	330	361	392	423	454	485
52	83	114	145	176	207	238	269	300	331	362	393	424	455	486
53	84	115	146	177	208	239	270	301	332	363	394	425	456	487
54	85	116	147	178	209	240	271	302	333	364	395	426	457	488
55	86	117	148	179	210	241	272	303	334	365	396	427	458	489
56	87	118	149	180	211	242	273	304	335	366	397	428	459	490
57	88	119	150	181	212	243	274	305	336	367	398	429	460	491
58	89	120	151	182	213	244	275	306	337	368	399	430	461	492
59	90	121	152	183	214	245	276	307	338	369	400	431	462	493
60	91	122	153	184	215	246	277	308	339	370	401	432	463	494
61	92	123	154	185	216	247	278	309	340	371	402	433	464	495
62	93	124	155	186	217	248	279	310	341	372	403	434	465	496

DISPLAY= SY /1000 , NODE 4, SURFACE C

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-13	-13
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12
-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-10
-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-8	-8	-8	-7
-4	-4	-4	-4	-4	-4	-4	-4	-3	-3	-3	-4	-4	-2
8	8	8	8	8	8	8	8	8	8	8	8	8	11
33	33	33	33	33	33	33	33	33	33	33	33	34	35
51	51	51	51	51	51	51	51	51	51	51	51	52	51
52	52	51	51	51	51	51	51	51	51	51	51	51	51

3-BLK CASE
RUN DFE

6 OF 18

15 BLK

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 ——— 23
SCALE

DISPLAY= SY /1000 , NODE= 4 , SURFACE= 1

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-13	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-13	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-13	-3	-32
-12	-12	-12	-12	-12	-12	-12	-13	-12	-13	-14	-12	-3	-32
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-13	-12	-3	-31
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-11	-3	-29
-11	-11	-11	-11	-11	-11	-11	-11	-11	-10	-11	-10	-2	-26
-9	-9	-9	-9	-8	-8	-8	-8	-8	-8	-8	-8	-2	-18
-4	-3	-3	-3	-3	-3	-3	-3	-3	-3	-3	-4	-2	-1
8	8	8	8	8	8	8	8	8	8	9	6	0	30
33	33	33	33	33	33	33	33	33	33	35	31	12	78
51	51	51	51	51	51	51	51	50	50	54	49	18	114
51	51	51	51	51	51	51	51	50	50	54	49	13	124

0 23
SCALE

SPEC
5.1

BOTTOM HALF OF CYLINDER
THERMO LOADS

15 BLK
7 OF 18

1/1/1

DISPLAY= SY /1CCC , NODE= 4, SURFACE= 2

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-22	6
-12	-12	-12	-12	-12	-12	-12	-12	-12	-12	-11	-12	-21	6
-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-11	-19	5
-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-9	-15	3
-4	-4	-4	-4	-4	-4	-4	-4	-4	-4	-4	-3	-5	-2
8	8	8	8	8	8	8	8	8	8	7	10	16	-9
34	34	34	34	34	34	34	34	34	34	31	35	56	-8
51	51	51	51	51	51	51	51	52	52	49	53	85	-12
52	52	52	52	52	52	52	52	52	52	48	53	90	-22

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

15 RLK
8 of 18

DISPLAY= SY /1000 , NODE= - 1, SURFACE= 0

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-13

SPEC
4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

15 BLK
9 of 19

1/1/1

DISPLAY= SY /1000 , NODE= -1 , SURFACE= 1

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-14	-12	-3	-32

SPEC 4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 ——— 23
SCALE

15 BLK
10 OF 18

1/17

DISPLAY= SY /1000 , NODE= 4, SURFACE= 2

-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	[illegible]	6
-13	-13	-13	-13	-13	-13	-13	-13	-13	-13	-12	-13	-23	6

SPEC 4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

15 BLK
11 OF 18

1/1/1

DISPLAY= SX /1000 , NODE= 4 , SURFACE= 1

0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22

15 BLK
12 OF 18

SPEC 4.1

TOP HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

1/1/1

DISPLAY= SX /1000 , NODE= 4, SURFACE= 0

0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16
0	0	0	0	0	0	0	0	0	0	0	2	-2	-16

15 BLK
13 OF 18

SPEC TOP HALF OF CYLINDER

DISPLAY= SX /1000 , NODE= 4, SURFACE= 1

0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
0	0	0	0	0	0	0	0	0	0	0	2	1	-22
1	1	1	0	0	0	0	0	0	0	0	2	1	-22
1	1	1	1	1	1	1	1	1	0	0	2	1	-21
1	1	1	1	1	1	1	1	1	1	0	2	2	-20
1	1	1	1	1	1	1	1	1	1	1	2	3	-18
1	1	1	1	1	1	1	1	1	1	2	1	5	-13
0	0	0	0	0	1	1	1	1	1	2	-2	8	0
0	0	0	0	0	0	0	0	0	1	1	-5	7	29
-1	-1	-1	-1	-1	-1	-1	-1	-1	0	0	-8	1	64
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-10	-4	83
-1	-1	-1	-2	-2	-2	-2	-2	-2	-2	-1	-10	-5	87

SPEC 5.1 BOTTOM HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

15 BLK
14 OF #8

DISPLAY= SX /1000 , NODE= [illegible] , SURFACE= 2

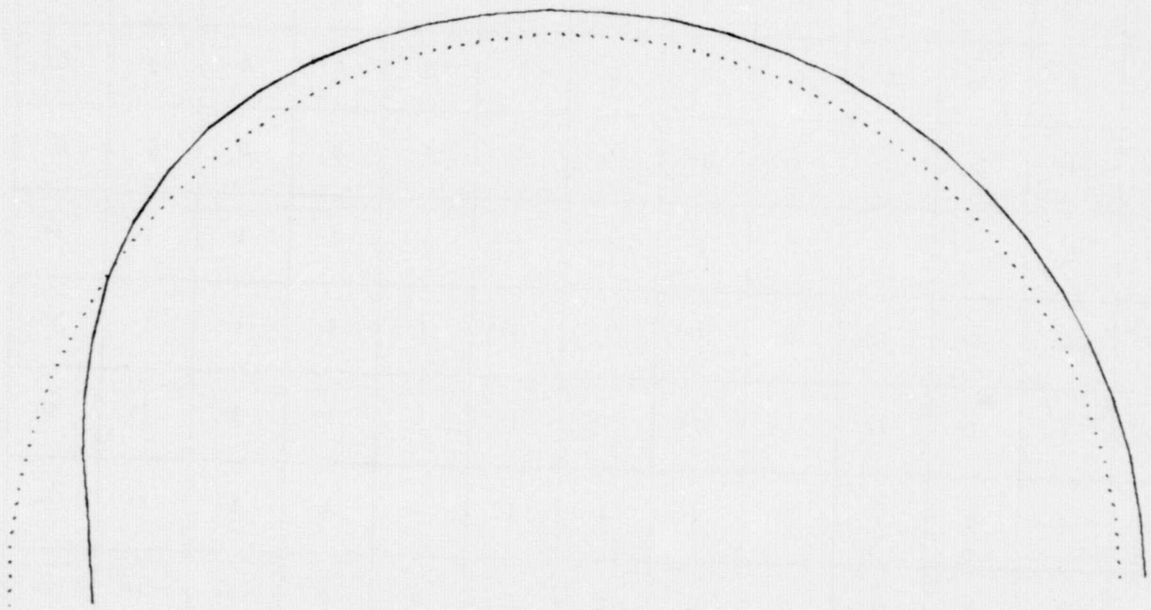
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
0	0	0	0	0	0	0	0	0	0	0	2	-5	-11
-1	-1	-1	0	0	0	0	0	0	0	0	2	-5	-10
-1	-1	-1	-1	-1	-1	-1	-1	0	-1	0	1	-5	-10
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	1	-5	-9
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	0	-6	-7
-1	-1	-1	-1	-1	-1	-1	-1	-1	-1	-2	0	-6	-2
0	0	0	0	-1	-1	-1	-1	-1	-1	-2	-1	-4	6
0	0	0	0	0	0	0	0	0	0	-1	-1	0	25
1	1	1	1	1	1	1	1	1	1	0	-1	8	45
1	1	1	1	1	1	1	1	1	2	1	-2	16	49
1	1	2	2	2	2	2	2	2	3	1	-3	20	44

SPEC 5.1

BOTTOM HALF OF CYLINDER
THERMO LOADS

0 23
SCALE

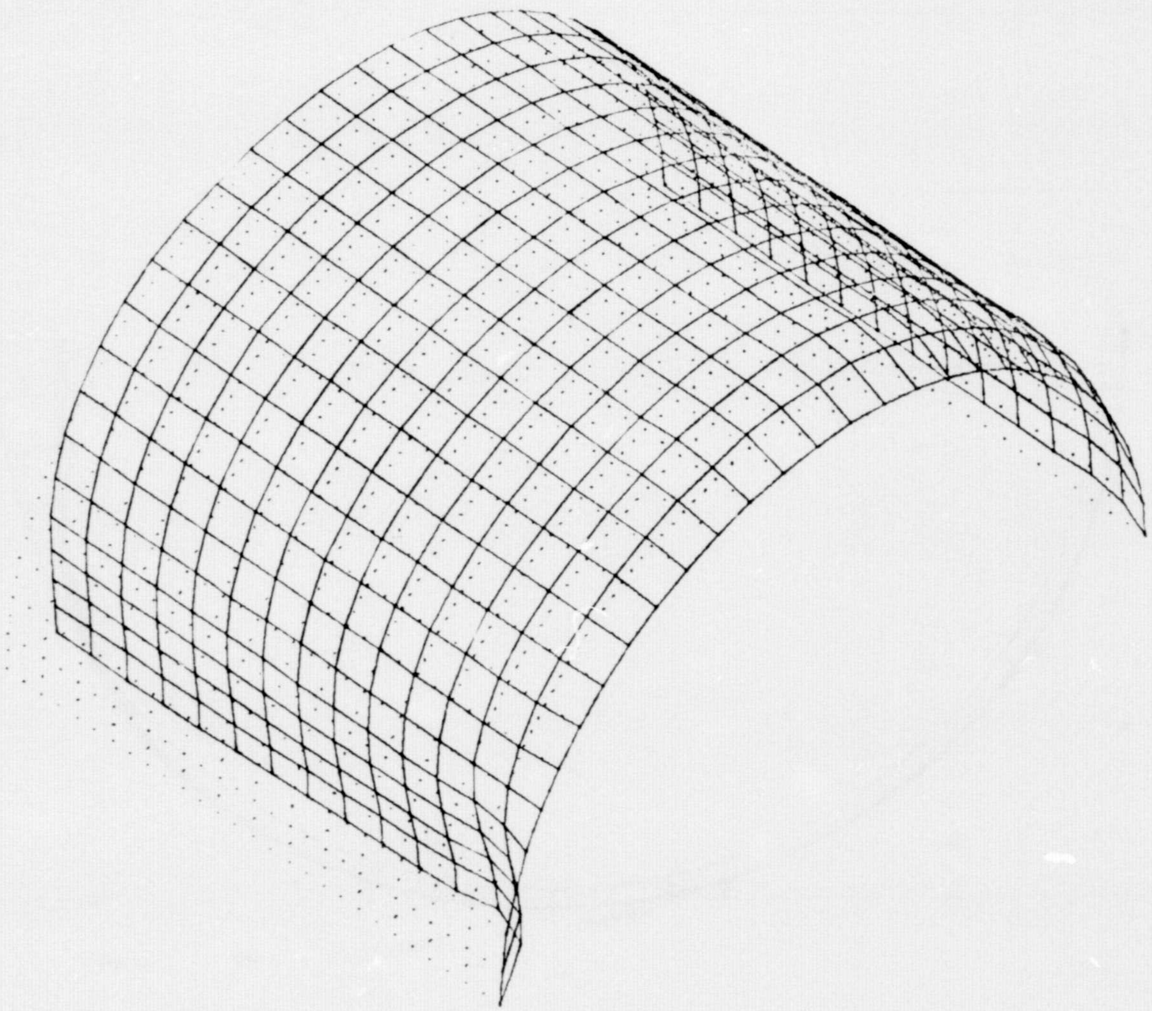
15 BLK
15 of 18



0 35
SCALE

SPEC RING
2.1

15 BLK
16 OF 15



SPEC
6.1

ALL

15 BLK
17 of 18

SPEC ALL
7.1

0 35
SCALE

15 BLK
18 OF 18

BY	DATE	SUBJECT	SHEET NO.	OF
CHKD. BY	DATE		JOB NO.	

FATIGUE DAMAGE FROM LN2 OR GN2 AT DIFF LOCATIONS IN TUNNEL

1. TYPICAL STRUCT RING

Stress Values

	Pressure	Transient Thermal	LN2 ACCIDENT THERMAL STRESSES SMALL accid.	LARGE accid.
σ_H	17	6.5	—	—
σ_L	25	-16.0	60, & -2.8	31. & -32.

operating cycle – NORMAL

	cd+P	P	Htup+P	End	
σ_H	23.5	17	10.5	0	$= \Delta S = 41$
σ_L	9	25	41	0	

operating cycle with accident during S.S.
small accident

σ_H	23.5	17	10.5	0	$\Rightarrow \Delta S = 85$
σ_L	68.0	85.	41 *	0	

operating cycle with accident begining Trans cd
large accident

σ_H	23.5	17	10.5	0	$\Delta S = 46.5$
σ_L	-23	-7	9	0	

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∴ small accident yields higher stresses

* Accident stresses do not add to ht up cycle because

BY________ DATE________ SUBJECT________________ SHEET NO.______ OF______
CHKD. BY______ DATE________ JOB NO.____________

$$SA = \frac{1}{2}(85)(3) = 127.5 \Rightarrow N = 300 \text{ cycles}$$

from ASME CODE

This is stress level during accident and the fatigue damage from this accident must be added to the fatigue damage from normal operation to determine how it affects shell life.

Life of vessel for normal operation (L) = 31 years

Damage factor for normal operation $\equiv \sum_{i=1}^{n} \frac{n_i}{N_i}$

$$20\left(\frac{1}{\sum \frac{n_i}{N_i}}\right) = L \quad \text{or} \quad \sum \frac{n_i}{N_i} = \frac{20}{L}$$

$$\therefore \sum \frac{n_i}{N_i} = \frac{20}{31} = .645$$

$N_a \equiv$ # of accidents

Total fatigue damage ≤ 1 in 20 years

$$\sum \frac{n_i}{N_i} + \frac{N_a}{N} \leq 1$$

$$\text{or } L = \frac{20}{\sum \frac{n_i}{N_i} + \frac{N_a}{N}}$$

N_a	L years
1	31
10	29
50	25
100	21

$\Delta L = .0973\, N_a$

BY ______ DATE ______ SUBJECT ______ SHEET NO. ______ OF ______
CHKD. BY ______ DATE ______ JOB NO. ______

2. ELLIPTICAL RING-WELD

Stress Values

		Press	therm	LN2 Accident Thermal Stress small	large
σ_H	I	22.22	6.5	-16	
	O	12.57	22.0		
σ_L	I	20.63	-16.0	60. & -2.8	31. & -32.0
	O	-11.22	16.0		

worst stresses will occur during small accident on Inside

	cd+P	P	Ht+P	End
σ_H	28.77	22.22	15.72	0
σ_L	64.63	80.63	36.63	0

$\Delta S = 80.63$

$S_a = \frac{1}{2}(80.63)(3) = 121 \Rightarrow N \doteq 300$

For normal operation L = 15 years

Na	L
1	15
10	15
50	14
100	12

$\Rightarrow$ from linear regression Anal.
$\Delta L = .03\ Na$

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REF: (1) NASA-CR-115301
ENGINEERING HANDBOOK
FOR THERMOSTRUCTURAL
ANALYSIS OF PLATES & SHELLS.

TSN 5.3.0-1

THERMAL BUCKLING OF ISOTROPIC CIRCULAR CYLINDRICAL SHELLS; EITHER EDGE CLAMPED OR SIMPLY SUPPORTED

NOTATION

A	= Area of cross section taken normal to the axis of revolution, in^2.
E	= Young's modulus, psi.
I_y, I_z	= Area moments of inertia taken about the y and z axes, respectively, in^4.
L	= Overall length of the cylinder, in.
M_x	= Running bending moment about middle surface of shell wall (see Figure 2), $\frac{in\text{-}lb}{in}$.
$\overline{M}_y$, $\overline{M}_z$	= Overall bending moments about the y and z axes, respectively (see Figure 2), in-lb.
$(\overline{M}_y)_A$, $(\overline{M}_z)_A$	= Artificial values for $\overline{M}_y$ and $\overline{M}_z$, respectively [see Equations (7)], in-lb.
$(\overline{M}_y)_B$, $(\overline{M}_z)_A$	= Artificial values for $\overline{M}_y$ and $\overline{M}_z$, respectively [see Equations (9)], in-lb.
$\overline{P}$	= Axial force (see Figure 1), lb.
$\overline{P}_A$	= Artificial value for $\overline{P}$ [see Equations (6)], lb.
$\overline{P}_B$	= Artificial value for $\overline{P}$ [see Equations (9)], lb.
R	= Radius of cylinder middle surface, in.
T	= Temperature change from that of an initial unstressed state or reference temperature (positive for a temperature rise), °F.
t	= Thickness of shell wall, in.
w	= Radial deflection of shell wall, in.
x, y, z	= Rectangular Cartesian coordinates (see Figure 1), in.
α	= Coefficient of linear thermal expansion, $\frac{in}{(in)(°F)}$.

NOTATION

γ = Knock-down factor (see Figure 3), dimensionless.

ν = Poisson's ratio, dimensionless.

σ_A = Artificial axial stress defined by Equation (5), psi.

$\left(\sigma_{\overline{M}_y}\right)_B, \left(\sigma_{\overline{M}_z}\right)_B$ = Axial stresses due to the artificial bending moments $\left(\overline{M}_y\right)_B$ and $\left(\overline{M}_z\right)_B$, respectively, psi.

$\left(\sigma_{\overline{P}}\right)_B$ = Axial stress due to the artificial force $\overline{P}_B$, psi.

σ_x = Axial stress, psi.

$\left(\sigma_x\right)_{Max}$ = Peak value for σ_x, psi.

$\left(\sigma_x\right)_{cr}$ = Critical axial stress for buckling of the cylinder, psi.

ϕ = Angular coordinate (see Figure 1), radians.

Note: All stresses are positive in tension.

CONFIGURATION

The design curves and equations provided here apply only to thin-walled, right circular cylinders which satisfy the relationship

$$L/R \geq \frac{3.2}{\left(\frac{R}{t}\right)^{1/2}} \qquad (1)$$

and are made of isotropic material. It is assumed that the shell wall is free of holes, obeys Hooke's law, and that it is of constant thickness. Figure 1 depicts the isotropic cylindrical shell configuration. Figure 2 shows the sign convention for forces, moments, and pressures.

BOUNDARY CONDITIONS

The following types of boundary conditions are covered:

a. Simply supported edge; that is,

$$w = M_x = 0 \quad \text{at } x = 0 \text{ and/or } x = L \qquad (2)$$

b. Clamped edge; that is,

$$w = \frac{\partial w}{\partial x} = 0 \quad \text{at } x = 0 \text{ and/or } x = L \qquad (3)$$

It is not required that the conditions at the two ends be the same. In every case, it is assumed that the cylinder (including any end rings) is not subjected to external axial constraints at any location around the boundaries at x = 0 and x = L.

TEMPERATURE DISTRIBUTION

The supposition is made that no thermal gradients exist through the wall thickness and in the axial direction. However, arbitrary circumferential variations may be present. The permissible distributions can therefore be expressed in the form

$$T = T(\phi) \qquad (4)$$

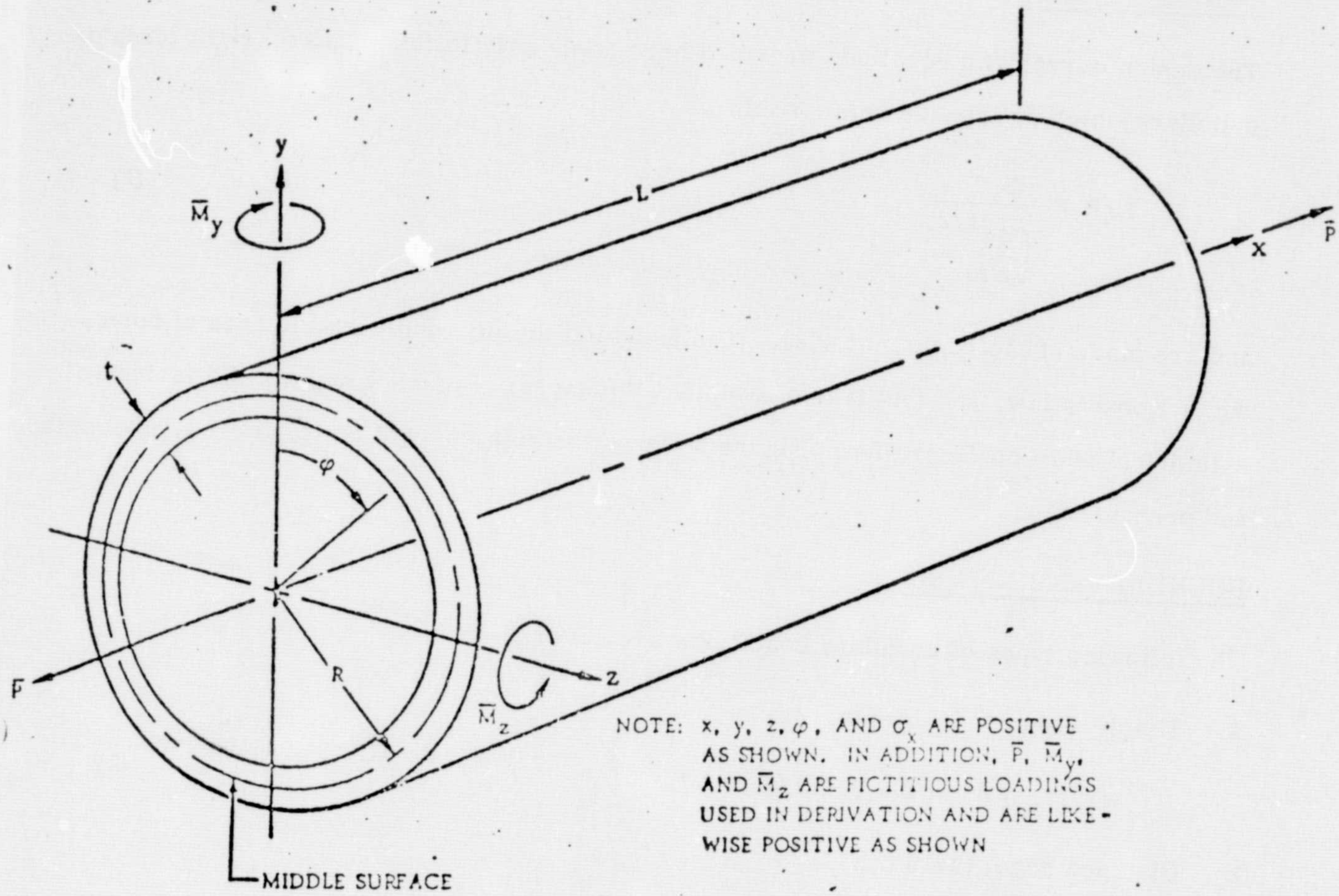

Figure 1. Isotropic Cylindrical Shell Configuration

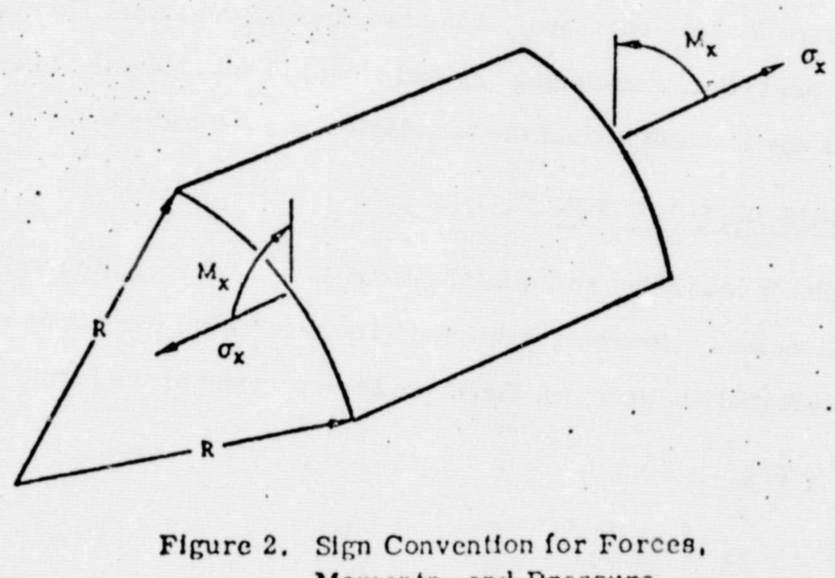

Figure 2. Sign Convention for Forces, Moments, and Pressure

Hoop membrane compression may develop in regions adjacent to the two ends due to external radial constraint. However, the buckling mode associated with this condition is not considered. Because of this and the lack of external axial constraints, the special case of a uniform temperature is of no interest here.

DESIGN CURVES AND EQUATIONS

It is assumed that Young's modulus and Poisson's ratio are unaffected by temperature changes. Hence, in using the contents of this TSN, the user must select effective values for each of these properties by applying engineering judgement. It will sometimes be desirable to employ different effective moduli in each of the following operations:

a. Computation of the stresses σ_x present in the cylinder.

b. Computation of the critical buckling stress $(\sigma_x)_{cr}$.

On the other hand, the results are presented in a form which enables the user to fully account for temperature-dependence of the thermal-expansion coefficient α.

The appropriate formulation for σ_x can be obtained by first imposing a fictitious stress distribution σ_A around the boundaries at $x = 0$ and $x = L$ such that all axial thermal deformations are entirely suppressed. It follows that

$$\sigma_A = -\alpha \bar{E} T(\phi) \tag{5}$$

These stresses may be integrated around the circumference and through the wall thickness to arrive at the force

$$\bar{P}_A = -EtR \int_0^{2\pi} \alpha T(\phi)\, d\phi \tag{6}$$

and the moments

$$(\bar{M}_y)_A = -ER^2 t \int_0^{2\pi} \alpha T(\phi) \sin\phi\, d\phi \tag{7}$$

$$\left(\overline{M}_z\right)_A = -ER^2 t \int_0^{2\pi} \alpha T(\phi) \cos\phi \, d\phi \qquad (7) \text{ (Contd)}$$

Since it is assumed that the shell is free of external axial constraints, the conditions

$$\overline{P} = \overline{M}_y = \overline{M}_z = 0 \qquad (8)$$

must be satisfied at $x=0$ and $x=L$. To restore the shell to such a state, it is necessary to superimpose a force $\overline{P}_B$ equal and opposite to $\overline{P}_A$ as well as moments $\left(\overline{M}_y\right)_B$ and $\left(\overline{M}_z\right)_B$ which are equal and opposite to $\left(\overline{M}_y\right)_A$ and $\left(\overline{M}_z\right)_A$, respectively. Hence,

$$\overline{P}_B = -\overline{P}_A$$

$$\left(\overline{M}_y\right)_B = -\left(\overline{M}_y\right)_A \qquad (9)$$

$$\left(\overline{M}_z\right)_B = -\left(\overline{M}_z\right)_A$$

The stress corresponding to $\overline{P}_B$ is easily found to be

$$\left(\sigma_{\overline{P}}\right)_B = \frac{\overline{P}_B}{A} = \frac{\overline{P}_B}{2\pi Rt} = \frac{E}{2\pi} \int_0^{2\pi} \alpha T(\phi) \, d\phi \qquad (10)$$

The stresses due to $\left(\overline{M}_y\right)_B$ are

$$\left(\sigma_{\overline{M}_y}\right)_B = \frac{\left(\overline{M}_y\right)_B z}{I_y} = \frac{\left(\overline{M}_y\right)_B z}{\pi R^3 t} = \frac{E \sin\phi}{\pi} \int_0^{2\pi} \alpha T(\phi) \sin\phi \, d\phi \qquad (11)$$

And those due to $\left(\overline{M}_z\right)_B$ are

$$\left(\sigma_{\overline{M}_z}\right)_B = \frac{\left(\overline{M}_z\right)_B y}{I_z} = \frac{\left(\overline{M}_z\right)_B y}{\pi R^3 t} = \frac{E \cos\phi}{\pi} \int_0^{2\pi} \alpha T(\phi) \cos\phi \, d\phi \qquad (12)$$

The procedure being used constitutes an application of Saint-Venant's principle. Hence, the stresses from Equations (10) through (12) will be accurate representations only at sufficient distances from the ends $x=0$ and $x=L$. If end rings are present,

the greater their resistance to out-of-plane bending, the shorter will be this distance. Subject to these conditions, the actual longitudinal thermal stresses at various points in the shell may be computed from the relationship

$$\sigma_x = \sigma_A + \left(\sigma_{\bar{P}}\right)_B + \left(\sigma_{\bar{M}_y}\right)_B + \left(\sigma_{\bar{M}_z}\right)_B \tag{13}$$

or

$$\sigma_x = -\alpha E T(\phi) + \frac{E}{2\pi}\int_0^{2\pi} \alpha T(\phi)\,d\phi + \frac{E\sin\phi}{\pi}\int_0^{2\pi} \alpha T(\phi)\sin\phi\,d\phi$$

$$+ \frac{E\cos\phi}{\pi}\int_0^{2\pi} \alpha T(\phi)\cos\phi\,d\phi \tag{14}$$

Complex distributions may be encountered which make it difficult to perform the required integrations. In such instances, use can be made of numerical techniques whereby the integral signs are replaced by summation symbols.

To investigate the stability of a particular shell, the maximum longitudinal stress $\left(\sigma_x\right)_{Max}$ must be compared against the critical value which can be obtained from the formula

$$\left(\sigma_x\right)_{cr} = \bar{\gamma}\,\frac{Et}{R\sqrt{3(1-\nu^2)}} \tag{15}$$

For the design to be satisfactory, it is required that

$$\left(\sigma_x\right)_{Max} < \left(\sigma_x\right)_{cr} \tag{16}$$

The quantity γ appearing above is a so-called knock-down factor which mainly accounts for the detrimental effects from initial imperfections. Note that Equation (15) is identical to that used for uniformly compressed circular, cylindrical shells. Its application to the present problem is justified on the basis of small-deflection studies reported in References 1 and 2. From the results given in these references, it can be concluded that, regardless of the nature of the circumferential stress distribution, classical

theoretical instability is reached when the peak axial compressive stress satisfies the expression

$$\left(\sigma_x\right)_{Max} \approx \frac{Et}{R\sqrt{3(1-\nu^2)}} \tag{17}$$

In view of this, the values used here for γ were determined from the 99% probability (confidence = 0.95) data for uniformly compressed cylinders as reported in Reference 3. The resulting γ values are plotted in Figure 2 for $\frac{L}{R}$ ratios of 0.25, 1.0, and 4.0.

SUMMARY OF EQUATIONS AND CURVES

$$\sigma_x = -\alpha E T(\phi) + \frac{E}{2\pi}\int_0^{2\pi} \alpha T(\phi)\, d\phi + \frac{E \sin\phi}{\pi}\int_0^{2\pi} \alpha T(\phi) \sin\phi\, d\phi$$

$$+ \frac{E \cos\phi}{\pi}\int_0^{2\pi} \alpha T(\phi) \cos\phi\, d\phi \tag{18}$$

$$\left(\sigma_x\right)_{cr} = \gamma \frac{Et}{R\sqrt{3(1-\nu^2)}} \tag{19}$$

When $\nu = 0.3$ this gives

$$\left(\sigma_x\right)_{cr} = 0.606\, \gamma \frac{Et}{R} \tag{20}$$

The knock-down factor γ is obtained from Figure 3.

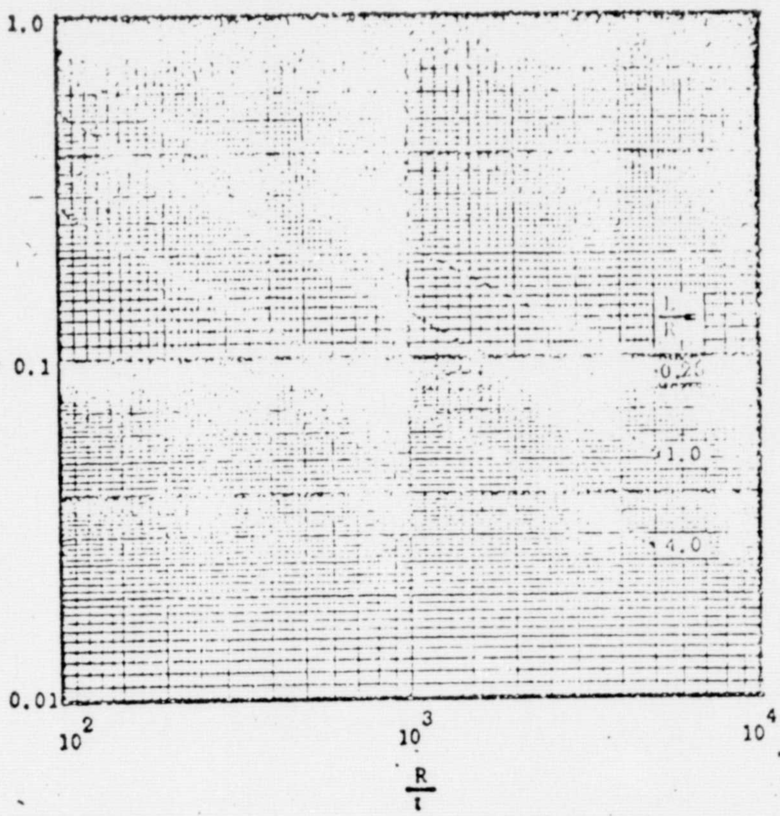

Figure 3. Knock-down Factor

7/12/76

Estimated Thermal Stresses in Deep "T"

There will be some 19" high "T" rings in the LN_2 injection area of tunnel. Need to factor these into fatigue analysis.

Temperature Distribution

Both the insulation thickness and the "T" ring depth will be increased to 15". Therefore the resistance of the composite insulation will be increased approximately by a factor of 4. The deep "T" rings are located in a higher speed leg of the tunnel, therefore the film coef. will be higher. However, this will be a very small part of the total resistance and can be neglected. Therefore the overall heat loss will be reduced by a factor of 4, and it would be reasonable to assume that the temp. grad. in the deep "T" ($T_{flange} - T_{shell}$) will be the same as the small "T".

Heat loss thru "T".

$$Q_{DT} = \frac{KA}{t}\left(T_{flange} - T_{shell}\right)$$

$$Q_{DT} = Q_{ST}/4 \qquad t_{DT} = 4\,t_{ST}$$

$$\left(T_f - T_s\right)_{DT} = \frac{Q_{ST}}{4}\,\frac{4t_{ST}}{KA} = \left(T_f - T_s\right)_{ST} = 10\,F^{\circ}$$

Thermal Stress

Use the results for the completely restrained shell:-

For $\Delta T = 10°$

	σ_L	σ_H
Inside	-3000 *	500
outside	3000	2000

the shell geometry in the LN_2 region is similar to that for which curves were generated and will be good enough for estimate

$$* \sigma_L = \alpha E \Delta T = (10 \times 10^{-6})(30 \times 10^{6})(10°F)$$

$$\sigma_L = 3000 \text{ psi}$$